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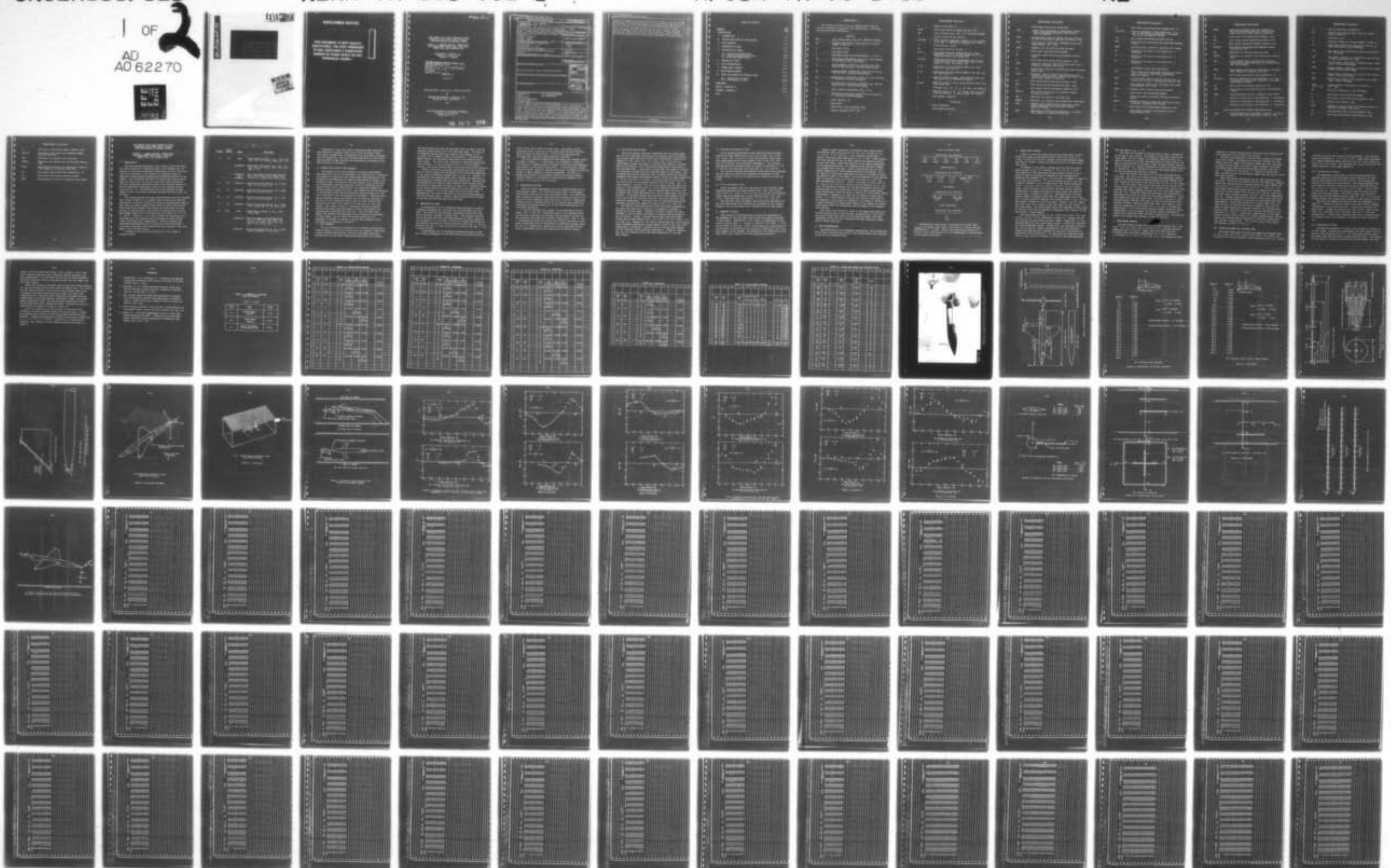

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DATA REPORT FOR A TEST PROGRAM TO STUDY TRANSONIC FLOW FIELDS ABOUT AIRCRAFT WITH APPLICATION TO EXTERNAL STORES

VOLUME I. - SUMMARY REPORT, TUNNEL-EMPTY AND MACH-NUMBER SURVEY DATA, FORCE AND MOMENT DATA, AND PRESSURE DATA

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A test program was conducted to obtain measurements of flow velocities and static pressures in the vicinity of wing-body-store model (representative of a fighter-type aircraft) as well as surface pressures, forces, and moments on the model. Flow velocities and static pressures were also measured near the tunnel walls to provide outer flow field information. This report presents the data obtained during the test program conducted in the 4T and 16T Wind Tunnels at Arnold Engineering Development Center. The Flow-field data were obtained at Mach numbers 0.925, 0.975, and 1.025 and constitute the major part of the data. (cont)

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Volume I is a summary report which gives detailed information on the test program and presents uncertainties associated with the various types of data taken in the 4T Wind Tunnel. The volume also presents tunnel-empty and Mach-number surveys, as well as tabulated force and moment and pressure data for the Mach number range 0.80 to 1.15 and angles of attack -2°, -5°, 0°, 2°, and 5°. Volumes II, III, and IV present the tabulated flowfield data for the 4-percent thick wing model at Mach numbers 0.925, 0.975 and 1.025, respectively. Volume V presents the tabulated flow-field data for the 6-percent thick wing model, and Volume VI presents data obtained for the 4-percent thick wing model in the 16T Wind Tunnel.

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NOMENCLATURE

This section provides a list of symbols which identify various aerodynamic parameters, axis designations, subscripts, and tabulated data nomenclature.

Symbols

AAL	local upwash angle, deg; $\tan^{-1}[(WL/VM)/(UL/VM)]$
A_W	planform area of both wings (does not include body), 0.4444 ft^2
b	wing span, 16 in.
c	local wing chord
$\bar{c}$	reference length for pitching moment, 5.3444 in.
C_A	axial-force coefficient, positive in the positive X direction, axial force/$q_\infty A_W$
C_ℓ	rolling-moment coefficient, positive left wing down as seen by pilot, rolling moment/$q_\infty A_W b$
C_m	pitching-moment coefficient, positive nose up as seen by pilot, pitching moment/$q_\infty A_W \bar{c}$
C_N	normal-force coefficient, positive in the positive Z direction, normal force/$q_\infty A_W$
C_n	yawing-moment coefficient, positive nose left as seen by pilot, yawing moment/$q_\infty A_W b$
C_P	local pressure coefficient, $(p_\ell - p)/q_\infty$
C_Y	side-force coefficient, positive in the positive Y direction, side force/$q_\infty A_W$
D	probe diameter, in.
M	Mach number
p	free-stream static pressure, psfa
q	dynamic pressure, $\frac{1}{2}\rho V^2$, psf

NOMENCLATURE (Continued)

r	radius of the body, in.
Re/ft	free-stream Reynolds number per foot, ft^{-1}
SWL	local sidewash angle, deg; $\tan^{-1}[(VL/VM)/(UL/VM)]$
t	airfoil thickness, see figure 3
UL,VL,WL	local velocity components positive in the positive X, Y, and Z directions, respectively, ft/sec
V	total velocity, ft/sec
VM	free-stream velocity, ft/sec
X,Y,Z	body-fixed Cartesian coordinates with origin coincident with the aircraft model nose at all angles of attack, see figure 6(a)
XT,YT,ZT	tunnel-fixed Cartesian coordinates with origin coincident with the aircraft model nose at zero angle of attack, see figure 6(b)
Y_1,Z_1	coordinates of wing trailing edge at the wing root, in.; see figure 10
Y_2,Z_2	coordinates of wing trailing edge at the wing tip, in.; see figure 10
α	angle of attack of model, angle between body axis and tunnel axis as defined in figure 7
α_{probe}	angle of attack of probe, angle between probe axis and tunnel axis
Δ	increment along XT, YT, or ZT axis, see Table II
θ	azimuthal angle in the Y-Z plane, deg; measured from the positive Y axis as shown in figure 6
ρ	mass density, $slugs/ft^3$

Subscripts

ℓ	local conditions
∞	free-stream conditions

NOMENCLATURE (Continued)

Tunnel-Empty Survey Data Tabulations

AATL	upwash angle referenced to tunnel-axis coordinates, calculated from probe measurements, deg; $\tan^{-1}(WT/VT)$
ALFA	aircraft-model angle of attack, positive nose up as seen by the pilot (nose down in tunnel), deg
CPL	local pressure coefficient calculated from probe measurements, (PL - P)/Q
DATE	calendar time at which data were recorded
M	wind tunnel free-stream Mach number
ML	local Mach number calculated from probe measurements
P	wind tunnel free-stream static pressure, psfa
PART	sequential indexing number for referencing data; a constant throughout each survey
PL	local static pressure calculated from probe measurements, psfa
POINT	sequential indexing number for referencing data obtained during one part; indexes each time a new set of data inputs is obtained
PT	wind tunnel free-stream total pressure, psfa
PTL	local total pressure measured by probe, psfa
Q	wind tunnel free-stream dynamic pressure, psf
REX10-6	wind tunnel free-stream unit Reynolds number, millions per foot
RUN	identifier for specific user test type
SURVEY	identifier for specific user grid-survey combination
SWTL	sidewash angle referenced to tunnel-axis coordinates, calculated from probe measurements, deg; $\tan^{-1}(VT/UT)$
TEST	alpha-numeric notation for referencing a specific test program in a specific test unit

NOMENCLATURE (Continued)

TT	wind tunnel free-stream total temperature, $^{\circ}F$
UT,VT,WT	velocity components in the tunnel-axis X, Y, and Z directions, respectively, calculated from probe measurements, ft/sec
VM	wind tunnel free-stream velocity, ft/sec
VML	local velocity calculated from probe measurements, ft/sec
WING	wing designation used for a specific part number
XT	location of the probe in the tunnel-axis X direction
YT	location of the probe in the tunnel-axis Y direction
ZT	location of the probe in the tunnel-axis Z direction

Mach-Number Survey Data Tabulations

AAL	local upwash angle referenced to body-axis coordinates, calculated from probe measurements, deg; $\tan^{-1}$(WL/UL)
ALFA	aircraft-model angle of attack, positive nose up as seen by the pilot (nose down in tunnel), deg
CPL	local-pressure coefficient calculated from probe measurements, (PL - P)/Q
DATE	calendar time at which data were recorded
M	wind tunnel free-stream Mach number
ML	local Mach number calculated from probe measurements
PART	sequential indexing number for referencing data; a constant throughout each survey
P	wind tunnel free-stream static pressure, psfa
PL	local static pressure calculated from probe measurements, psfa

POINT	sequential indexing number for referencing data obtained during one part; indexes each time a new set of data inputs is obtained
PT	wind tunnel free-stream total pressure, psfa
PTL	local total pressure measured by probe, psfa
Q	wind tunnel free-stream dynamic pressure, psf
REX10-6	wind tunnel free-stream unit Reynolds number, millions per foot
RUN	identifier for specific user test type
SURVEY	identifier for specific user grid-survey combination
SWL	local sidewash angle referenced to body-axis coordinates, calculated from probe measurements, deg; $\tan^{-1}$(VL/UL)
TEST	alpha-numeric notation for referencing a specific test program in a specific test unit
TT	wind tunnel free-stream total temperature, ^{o}F
UL,VL,WL	velocity components in the body-axis X, Y, and Z directions, respectively, calculated from probe measurements, ft/sec
VM	wind tunnel free-stream velocity, ft/sec
VML	local velocity calculated from probe measurements, ft/sec
WING	wing designation used for a specific part number
X	location of the probe in the body-axis X direction
Y	location of the probe in the body-axis Y direction
Z	location of the probe in the body-axis Z direction

Force and Moment and Pressure Data Tabulations

ALFA	aircraft-model angle of attack, positive nose up as seen by the pilot (nose down in tunnel), deg

NOMENCLATURE (Continued)

A_b	area of model base, .038785 ft^2
A_W	planform area of both wings (does not include body), 0.4444 ft^2
b	wing span, 16 in.
$\bar{c}$	reference length for pitching moment, 5.3444 in.
CA	axial-force coefficient measured by balance, in body coordinates, axial force/QA_W
CAB	base axial-force coefficient in body coordinates, $(P - \bar{P}_b)A_b/QA_W$
CAF	axial-force coefficient corrected for base effects, in body coordinates, CA - CAB
CLL	rolling-moment coefficient in unrolled body coordinates, rolling moment/QA_Wb
CLMF	pitching-moment coefficient in unrolled body coordinates, pitching moment/$QA_W\bar{c}$
CLN	yawing-moment coefficient in unrolled body coordinates, yawing moment/QA_Wb
CNF	normal-force coefficient in unrolled body coordinates, normal force/QA_W
CPS N (N = 1,25)	surface-pressure coefficient at orifice N, (PS - P)/Q
CY	side-force coefficient, side force/QA_W
M	wind tunnel free-stream Mach number
P	wind tunnel free-stream static pressure, psfa
$\bar{P}_b$	average base pressure, psfa
PART	sequential indexing number for referencing data; a constant throughout each sweep
PS	aircraft-model local surface pressure, psfa
PT	wind tunnel free-stream total pressure, psfa

Q	wind tunnel free-stream dynamic pressure, psf
REX10-6	wind tunnel free-stream unit Reynolds number, millions per foot
RUN	identifier for specific user test type
SURVEY	identifier for specific user grid-survey combination
TEST	alpha-numeric notation for referencing a specific test program in a specific test unit
TT	wind tunnel free-stream total temperature, ^{o}F
VM	wind tunnel free-stream velocity, ft/sec
WING	wing designation used for a specific part number

DATA REPORT FOR A TEST PROGRAM TO STUDY
TRANSONIC FLOW FIELDS ABOUT AIRCRAFT
WITH APPLICATION TO EXTERNAL STORES

VOLUME I. - SUMMARY REPORT, TUNNEL-EMPTY
AND MACH-NUMBER SURVEY DATA, FORCE AND
MOMENT DATA, AND PRESSURE DATA

1. INTRODUCTION

The test program described in this report, authorized under Air Force Contract No. F44620-75-C-0047, was conducted for the purpose of obtaining experimental measurements of flow velocities and static pressures in the vicinity of wing-body models in addition to body surface pressures and forces and moments on the models. Flow velocities and static pressures were also measured near the tunnel walls to provide outer flow field information. The test program was conducted in the 4T and 16T Wind Tunnels at Arnold Engineering Development Center at Mach numbers ranging from 0.80 to 1.15. The flow-field data were obtained at Mach numbers 0.925, 0.975, and 1.025 and constitute the major part of the data.

This report presents the data obtained during the test program. This volume, Volume I, is a summary report which gives detailed information on the test program and presents uncertainties associated with the various types of data taken in the 4T Wind Tunnel. This volume also presents tunnel-empty and Mach-number surveys, as well as tabulated force and moment and pressure data for the Mach number range 0.80 to 1.15 and angles of attack -2°, -5°, 0°, 2° and 5°. Volumes II, III, and IV present the tabulated flow-field data for the 4-percent thick wing model at Mach numbers 0.925, 0.975 and 1.025, respectively. Volume V presents the tabulated flow-field data for the 6-percent thick wing model, and Volume VI presents data obtained for the 4-percent thick wing model in the 16T Wind Tunnel.

A summary of the test data contained in each volume is presented below.

Volume	Wind Tunnel	Wing	Test Data
I	4T	None	Tunnel-empty surveys at M_∞ = .80, .85, .90, .95, 1.0, 1.025, 1.05, 1.10, 1.15
		4-percent	Mach-number surveys at M_∞ = .80, .85, .90, .95, 1.0, 1.025, 1.05, 1.10, 1.15 and $\alpha = 0^o$
		4-percent and 6-percent	Force and Moment and Pressure data at M_∞ = 0.80, 0.85, 0.90, 0.925, 0.95, 0.975, 1.0, 1.025, 1.05, 1.10, 1.15
II	4T	4-percent	Flow-field survey data at M_∞ = 0.925 and $\alpha = 0^o, \pm 2^o, \pm 5^o$
III	4T	4-percent	Flow-field survey data at M_∞ = 0.975 and $\alpha = 0^o, \pm 2^o, \pm 5^o$
IV	4T	4-percent	Flow-field survey data at M_∞ = 1.025 and $\alpha = 0^o, \pm 2^o, \pm 5^o$
V	4T	6-percent	Flow-field survey data at M_∞ = 0.925, .975, 1.025, and $\alpha = 0^o, \pm 2^o, \pm 5^o$
VI	16T	None	Tunnel-empty surveys at M_∞ = .925, .975, 1.025
		4-percent	Force and Moment and Pressure data at M_∞ = .80, .85, .90, .925, .95 .975, 1.0, 1.025, 1.05, 1.10, 1.15 and $\alpha = 0^o, \pm 2^o, \pm 5^o$
		4-percent	Flow-field survey data at M_∞ = 0.925, 0.975, 1.025 and $\alpha = 0^o, \pm 5^o$

References 1 and 2 are reports prepared by AEDC describing data reduction procedures used in the 4T and 16T Wind Tunnel facilities, respectively, to determine force and moment, pressure, and flow-field data. The data uncertainties given in this report were provided by AEDC and meet the data quality requirements that were outlined in reference 3.

2. PURPOSE AND SCOPE OF TEST PROGRAM

The purpose of the test program is to obtain experimental data which will aid in the development and evaluation of a theoretical method for predicting flow fields about three-dimensional configurations characteristic of modern fighter/bombers flying in the transonic range. The scope of the test program is to obtain data on a simplified wing-body combination with two different sets of wings at subcritical, supercritical, and supersonic speeds for several values of angle of attack. The effects of wall interference in the 4T Tunnel will also be studied by repeating several of the test conditions in the 16T Propulsion Wind Tunnel. Detailed information regarding these wind tunnels can be found in reference 4. The bulk of the testing was done in the 4T Tunnel and the data obtained from these tests comprise the major part of this data report. This volume, Volume I, summarizes the data, presents the 4T Tunnel-empty and Mach-number surveys, and presents some uncertainties associated with the data. Also, the force and moment and pressure data obtained in the 4T Tunnel are presented in this volume. The 4T Tunnel flow-field survey data are presented in Volumes II through V and all data obtained in the 16T Tunnel are presented in Volume VI.

3. TEST HARDWARE

The test installation consisted of a wing-body combination in whose proximity a flow-field survey probe was located. Flow-field survey, pressure distribution, and force and moment data were obtained for two wing-body combinations during the test program.

The flow-field survey data was obtained using the AEDC 0.25-inch diameter conical probe with a 20^{0} semi-apex angle. The body is an aluminum fuselage which was bored out and mounted on the 1.5-inch, 500 lb. AEDC 6-1.50-1.12 M-A balance. The force and moment data on the wing-body combinations were obtained using this balance. The entire body contained 25 axially-aligned pressure orifices, 6 on the nose portion and 19 on the fuselage section, from which pressure data on the body surface were obtained.

Figure 1 shows the wing-body combination with the 4-percent wing in the 4T Tunnel with the conical flow-field probe supported on the captive trajectory system (CTS, see ref. 4). Figure 2 is a drawing of the wing-body model which includes the positions of the 25 pressure orifices along the body centerline. Sketches and coordinates of the 4-percent and 6-percent thick airfoils which were used to obtain data are given in figure 3. A detailed sketch of the entire conical flow-field probe and of the probe tip are shown in figure 4. It is also noted that 0.0035-inch, #150 Carbolum grit was used on the nose tip and wing leading edges to trip the boundary layer. Figure 5 shows the positions and width of the grit for both the body nose and wings.

4. DEFINITION OF AXES

A conventional set of orthogonal body-fixed axes is used as a frame of reference for the inner flow-field surveys near the body. The origin of the wing-body combination system is at the tip of the nose, as shown in figure 6(a). The pressure orifices are on the same side that probe measurements are taken, which is on the negative Z axis side. As seen by a person positioned on the pressure orifices and looking toward the nose, the X axis is positive aft, the Y axis is positive to the left and the Z axis is positive down.

A conventional set of orthogonal tunnel-fixed axes is used as a frame of reference for the outer flow-field surveys near the

tunnel walls and is shown in figure 6(b). The origin of this reference frame is the nose of the model body when the wing-body is at an angle of attack of 0°. It is at this position that the body-fixed axes and tunnel-fixed axes coincide.

The sign convention adopted for the upwash and sidewash angles is shown in figure 7. In the figure, the wing-body is shown in the tunnel at a negative angle of attack to show the flow components with respect to the walls of the tunnel. In the body-axis system, positive angular values of sidewash and upwash correspond to positive values of their respective velocity components. As seen by a person positioned on the pressure orifices and looking forward, positive upwash is a downward flow and positive sidewash is an outward flow along the left wing panel.

5. DESCRIPTION OF TESTS

Tests for which experimental data are reported herein are of three general types: (1) flow-field survey tests, (2) pressure-distribution tests and (3) force and moment tests. The tests have been conducted at nominal free-stream Mach numbers of 0.80 to 1.15 and at a nominal Reynolds number per foot of 3.0×10^{6}. Tests were conducted with a 4-percent thick and a 6-percent thick airfoil to investigate thickness effects on the flow field generated by the wing-body combination.

A note is made here with respect to the positioning of the probe. The X and XT coordinates given in the tabulated data are the axial positions of the probe static pressure orifices in the body-axis and tunnel-axis systems, respectively. The Y and Z coordinates indicate the lateral and vertical positions of the probe longitudinal axis in the body-axis coordinate systems, while YT and ZT indicate the lateral and vertical positions of the probe longitudinal axis in the tunnel-axis coordinate system.

5.1 Flow-Field Survey Tests

Flow-field survey tests were conducted with the tunnel empty and with two wing-body configurations at several angles of attack. The conical probe used in the tests was calibrated at nominal Mach numbers of 0.80, 0.85, 0.90, 1.0, 1.025, 1.05, 1.10, and 1.15. Tunnel-empty surveys were made at the same Mach numbers to investigate the uniformity of the free-stream conditions in the region of the wing-body model. Mach-number surveys were also taken in regions of particular interest with the wing-body model in the tunnel. The aforementioned conical probe was used to obtain flow-field velocity components and upwash and sidewash angles, as well as other quantities, at various locations of interest. The velocities and angles were calculated using five pressures measured with the probe. One is a total pressure, located on the tip of the probe, and the other four consist of two orthogonal pairs of static pressures located on the surface of the conical probe. These pressures have also been used to deduce other local-flow quantities, such as Mach number and total pressure. Flow-field surveys were obtained at Mach numbers 0.925, 0.975, and 1.025 for both the 4-percent thick and 6-percent thick wing-body combinations. Data were taken at specified spanwise and chordwise positions for angles of attack of 0^o, $\pm 2^o$, and $\pm 5^o$.

The specific flow-field survey tests which were performed and the final data which were obtained are presented in Volumes I through VI of this report. Volumes I through V contain the data obtained in the 4T Tunnel and Volume VI contains data obtained in the 16T Tunnel. Volume I contains Tunnel-empty and Mach-number survey data at Mach numbers 0.80, 0.85, 0.90, 0.95, 1.0, 1.025, 1.05, 1.10 and 1.15. Volumes II, III, and IV contain data for the 4-percent thick wing-body combination at Mach numbers 0.925, 0.975, and 1.025, respectively, and Volume V contains data at the same Mach numbers for the 6-percent thick wing-body combination. Volume VI contains data for the 4-percent thick wing-body combination at the same Mach numbers.

5.2 Pressure Distribution Tests

Axial pressure distributions were obtained along the body surface for Mach numbers 0.80, 0.85, 0.90, 0.925, 0.95, 0.975, 1.0, 1.025, 1.05, 1.10 and 1.15 with the 4-percent thick wing-body combination and for Mach numbers 0.925, 0.975, and 1.025 with the 6-percent thick wing-body combination. The pressure distributions were obtained along the bottom of the body ($\alpha = 0^o$, 2^o, and 5^o) and along the top of the body ($\alpha = -2^o$ and -5^o).

The specific pressure distribution tests which were performed and the final data which were obtained are presented in this volume, Volume I, for the 4T Tunnel and in Volume VI for the 16T Tunnel.

5.3 Force and Moment Tests

Force and moment data were obtained for the 4-percent thick wing-body combination at Mach numbers 0.80, 0.85, 0.90, 0.925, 0.95, 0.975, 1.0, 1.025, 1.05, 1.10 and 1.15 and for the 6-percent thick wing-body combination at Mach numbers 0.925, 0.975, and 1.025. These data were obtained for $\alpha = 0^o$, $\pm 2^o$, and $\pm 5^o$.

The specific force and moment tests which were performed and the final data which were obtained are presented in this volume, Volume I, for the 4T Tunnel and in Volume VI for the 16T Tunnel.

6. SYMMETRY OF TESTS

This section of the data report contains an assessment of the symmetry of the flow-field survey tests previously described in this report. Specifically, this refers to comparisons of data taken at points whose Z values are identical, but whose Y locations differ in sign only. These comparisons are a good measure of the exactness of positioning the wing-body configuration and probe with respect to one another, the symmetry of the configuration and flow field, and a check of the data reduction procedure.

Figure 8 shows comparisons of pressure (C_p), upwash (AAL), and sidewash (SWL) for positions (4, -1), (-4, -1) and (7, -1), (-7, -1) at $\alpha = 0^o$ and $M_\infty = 0.925$ (4-percent thick wing). Figure 9 shows the same comparisons at $M_\infty = 1.025$. These comparisons were carried out for all Mach numbers and angles of attack and for both wings. With the exception of the upwash comparisons at the outboard positions, all comparisons for pressure, upwash and sidewash were very good. Several explanations for the poor upwash comparisons at the outboard positions are offered. First, the regions in which the comparisons are poor are regions in which the measured angle is very small, and the difference in the measurements is usually within the accuracy of the data ($\pm$ 0.40^o, see DATA UNCERTAINTIES section). Second, the wings on either side of the fuselage have slightly different maximum thickness at the tips. Also, the difference between the vertical coordinates of the wing trailing edge at the tip and root is not the same on both sides. These differences, which are given in figure 10, could effect upwash measurements, although it is felt that it would be a very small effect. It is also possible that the flow field of the tunnel is not symmetric in the region of interest, as is shown in reference 5. A small difference in tunnel-empty upwash in the region of interest could easily cause the differences seen in upwash at the outboard wing position.

The overall good agreement shown in the comparisons indicates accurate positioning of the wing-body configuration in the tunnel and of the probe with respect to the wing-body model, and lends confidence to both the test procedures and data reduction schemes used to obtain this data.

7. DATA UNCERTAINTIES

Uncertainties in the aerodynamic coefficients, local conditions, flow angles, and probe position for the 4T Wind Tunnel were provided by ARO and are presented below.

Force and Moment Data

Uncertainty (±), Absolute

C_N	C_Y	C_A	C_m	C_n	C_ℓ
0.004	0.002	0.004	0.004	0.001	0.003

Probe Position

Uncertainty (±), Absolute

X, in.	Y, in.	Z, in.	α_{probe}, deg.
0.05	0.05	0.05	0.15

Flow Angles

Uncertainty (±), Absolute

AAL, deg	SWL, deg
0.25	0.25

Local Conditions

Uncertainty (±), Absolute

C_P
0.03

Examining the uncertainties associated with probe angle (α_{probe}) and upwash angle (AAL), there exists a maximum possible uncertainty of $\pm 0.40^\circ$. This is the uncertainty mentioned in Section 6 that could account for the disagreement of upwash angle at the outboard position.

8. TUNNEL-EMPTY SURVEYS

Figure 11 shows the grids used in the tunnel-empty surveys, which were taken at Mach numbers 0.80, 0.85, 0.90, 0.95, 1.0, 1.025, 1.05 and 1.10 and at a nominal free-stream Reynolds number per foot of 3.0×10^6 to ascertain the quality of the tunnel-empty flow field.

An example of the type of results derived from the tunnel-empty surveys is the XT traverse at YT = 0.0, ZT = -14.14. The sidewash at this particular location should always be zero, since it lies in the plane of symmetry of the configuration. Examining the sidewash at M_∞ = 0.90 and 0.95, as shown in figure 12, it is seen to be nonzero. It could be argued at first that the "nonzero" sidewash is due to probe error, but closer inspection shows a consistent nonzero sidewash throughout the Mach number range. With the model in the tunnel, the same nonzero sidewash distribution is present throughout the Mach number and angle of attack range, as shown in figure 12. The consistency of these results offers reasonable proof that the tunnel-empty flow field has a slightly positive sidewash angle along (YT,ZT) = (0.0, -14.14). The nonzero sidewash is relatively small, however, and will therefore have very little effect on the data taken in this region. In this same manner, the tunnel-empty upwash and sidewash in other parts of the flow field can be examined to determine their effects on the data taken in those regions.

The tunnel-empty surveys at M_∞ = 0.80, 0.85, 0.90, 0.95, 1.0, 1.025, 1.05 and 1.10 are summarized in Table II. Columns one and two indicate the page number and part number, respectively, of the tabulated data for each tunnel-empty survey. Column four indicates the free-stream Mach number. Columns five, six and seven indicate the XT, YT, and ZT ranges, respectively, for each traverse. Column nine indicates the increment for the axis along which the traverse is being carried out. All positions are relative to the tunnel origin, which is located at the tip of the wing-body model

when the model is at $\alpha = 0^o$.

The data are presented in tabular form on pages 1 through 36 at the end of this volume. The heading on each page contains the test number, the part number, the Reynolds number per foot, the angle of attack of the model (not applicable for these tests), the type of wing attached to the model (none for these tests), and the (YT,ZT),(XT,YT), or (XT,ZT) coordinates at which the probe traverse is carried out. Also included are the run and survey numbers and the date on which the data were recorded.

Below the heading information are the data obtained during each test. The first two columns indicate the sequential indexing number for referencing data obtained during one part (POINT) and the location of the probe in the tunnel axis XT, YT, or ZT direction. The wind tunnel free-stream quantities are in columns three through seven, and are Mach number (M), velocity (VM, ft/sec), total pressure (PT, psfa), dynamic pressure (Q, psf), and total temperature (TT, ^oF). Following these quantities are local quantities as measured by the probe or calculated from probe measurements. These local quantities are Mach number (ML), the ratio of local to free-stream velocity (VML/VM), the ratio of local to free-stream total pressure (PTL/PT), pressure coefficient (CPL), the ratio of local velocity components in the tunnel axis X, Y, and Z directions, respectively, to the free-stream velocity (UT/VM, VT/VM and WT/VM, respectively), and the upwash and sidewash angles referenced to tunnel-axis coordinates (AATL and SWTL, respectively). The positive sense of the upwash and sidewash is shown in figure 7.

9. MACH-NUMBER SURVEYS

The Mach-number surveys were taken at Mach numbers 0.80, 0.85, 0.90, 0.95, 1.0, 1.025, 1.05, 1.10 and 1.15 at a nominal Reynolds number per foot of 3.0×10^6. These tests were taken with the 4-percent thick wing-body configuration at $\alpha = 0^o$ along an X traverse at Y = 3.0, Z = -2.0 and -1.0. The purpose of these

tests was to establish appropriate subcritical and supercritical test conditions for the flow-field surveys.

The data are presented in tabular form on pages 37 through 56 of this volume. The heading on each page contains the test number, the part number, the Reynolds number per foot, the angle of attack of the model, the type of wing attached to the model (4-percent thick wing for the Mach-number surveys), and the Y and Z coordinates at which the X traverse is carried out. Also included are the run and survey numbers and the date on which data were recorded.

Below the heading information are the data obtained during each test. The first two columns indicate the sequential indexing number for referencing data obtained during one part (POINT) and the location of the probe in the body-axis X direction. The wind tunnel free-stream quantities are in columns three through seven and are Mach number (M), velocity (VM, ft/sec), total pressure (PT, psfa), dynamic pressure (Q, psf), and total temperature (TT, ^{o}F). Columns eight through sixteen contain local quantities which were either measured by the probe or calculated from probe measurements. These local quantities are Mach number (ML), the ratio of local to free-stream velocity (VML/VM), the ratio of local to free-stream total pressure (PTL/PT), the pressure coefficient (CPL), the ratio of the local velocity components in the body-axis X, Y, and Z directions, respectively, to the free-stream velocity (UL/VM, VL/VM, and WL/VM, respectively), and the upwash and sidewash angles referenced to body-axis coordinates (AAL and SWL, respectively). The positive sense of the upwash and sidewash is shown in figure 7.

10. FORCE AND MOMENT AND PRESSURE TESTS

This section presents the force and moment and pressure data at M_∞ = 0.80, 0.85, 0.90, 0.925, 0.95, 0.975, 1.0, 1.025, 1.05, 1.10 and 1.15 for the 4-percent thick wing-body and at M_∞ = 0.925,

0.975 and 1.025 for the 6-percent thick wing-body. Data for both wings is presented at $\alpha = 0^{\circ}, \pm 2^{\circ}, \pm 5^{\circ}$. These tests, performed at a nominal Reynolds number per foot of 3.0×10^{6}, are outlined in Table IV of this volume. The tabulated data are at the end of this volume beginning on the page numbered 57.

10.1 Description of Tests

The aerodynamic coefficients on the wing-body models were obtained during the force and moment tests using the 1.5-inch, 500 lb. AEDC balance #6-1.50-0.50-1.12 M-A. The fuselage was bored out and mounted on this balance. The surface-pressure data were obtained from the 25 pressure orifices on the body surface. The orifices are labeled 1 through 25 with number 1 being closest to the nose tip. The data are arranged such that there are two pages for each Mach number. The first presents the force and moment data for the angle-of-attack range ($0^{\circ}, \pm 2^{\circ}, \pm 5^{\circ}$) and the second presents the pressure data obtained at each angle of attack.

The force and moment and pressure tests are summarized in Table IV. Columns one and two indicate the page and part numbers, respectively, of the tabulated data. Column four indicates the Mach number. Column six indicates the type of data given on each page; F & M indicates force and moment data and P indicates pressure data. Column eight indicates which wing (4-percent thick or 6-percent thick) was attached to the body for each particular run.

10.2 Description of Data

The data are presented in tabular form on pages 57 through 84 of this volume. As previously mentioned, each Mach number has two pages of data associated with it. The first page contains force and moment data and the second contains pressure data. The heading on both pages is identical and contains the test number, the part

number, the free-stream Mach number, total pressure, static pressure, Reynolds number per foot, velocity, dynamic pressure, and total temperature. Also included are the type of wing attached to the body (4-percent thick or 6-percent thick), the run number and the survey number.

Below the heading on the first page of each Mach number section are the data obtained during each force and moment test. The results for the force and moment tests include the wing-body model angle of attack, the normal-force coefficient (CNF), the side-force coefficient (CY), the axial-force coefficient corrected for base effects (CAF), the pitching-moment coefficient (CLMF), the yawing-moment coefficient (CLN), the rolling-moment coefficient (CLL), and the base axial-force coefficient (CAB). The positive sense of these forces and moments is shown in figure 13.

Below the heading on the second page of each Mach number section are the data obtained during each pressure test. Column one indicates the orifice at which the pressure coefficient was measured. Columns two through six indicate the pressure coefficient at $\alpha = -5^{o}$, -2^{o}, 0^{o}, 2^{o} and 5^{o}, respectively, at each orifice location. The locations of the pressure orifices are shown in figure 2.

REFERENCES

1. Rittenhouse, L. E. and Kaupp, H.: Procedures and Computer Program for Conducting Force Tests in the PWT 4T Facility. Revised Sept. 5, 1973.

2. Reichenau, D. E.: AFFDL Wing-Body Flowfield Study Test. PWT 16T Facility Project Criteria, Project No. P41T-M9A, Test No. TF-445, Mar. 10, 1977.

3. AEDC 4T Wind Tunnel Test Plan for the Study of Transonic Flow Fields About Aircraft with Application to External Stores. AFOSR Contract No. F44620-75-C-0047, Project No. 9781-01, Nov. 1976.

4. Test Facilities Handbook, Tenth Edition, Arnold Engineering Development Center, Arnold Air Force Station, TN, May 1974.

5. Luchuk, W.: Flow Angle Measurements Using a 2-Inch Span Cruciform-Wing Force Model. Presented at the 45th Semi-Annual Supersonic Tunnel Association, Albuquerque, New Mexico, Apr. 13-14, 1976.

TABLE I.- SUMMARY OF TABULATED DATA IN VOLUME I

Re/ft = 3.0×10^6

Table (1)	Test (2)	Pages (3)
II	Tunnel-empty surveys	1-36
III	Mach-number surveys	37-56
IV	Force and moment and pressure data	57-84

TABLE II.- TUNNEL-EMPTY SURVEYS

1	2	3	4	5	6	7	8	9	10
Page No.	Part No.		Mach No.	XT Range inches	YT Range inches	ZT Range inches		Δ inches	
1	34		0.80	-6.000 to 24.000	0.000	-14.142		2.000	
2	34			-6.000 to 24.000	0.000	0.000		2.000	
3	34			-6.000 to 24.000	14.142	0.000		2.000	
4	33			14.000	14.000 to -14.000	0.000		2.000	
5	33			14.000	0.000	14.000 to -14.000		2.000	
6	37		0.85	-6.000 to 24.000	0.000	-14.142		2.000	
7	37			-6.000 to 24.000	0.000	0.000		2.000	
8	37			-6.000 to 24.000	14.142	0.000		2.000	
9	36			14.000	14.000 to -14.000	0.000		2.000	
10	36			14.000	0.000	14.000 to -14.000		2.000	

TABLE II.- CONTINUED

1	2	3	4	5	6	7	8	9	10
Page No.	Part No.		Mach No.	XT Range inches	YT Range inches	ZT Range inches		Δ inches	
11	39		0.90	-6.000 to 24.000	0.000	-14.142		2.000	
12	39			-6.000 to 24.000	0.000	0.000		2.000	
13	39			-6.000 to 24.000	14.142	0.000		2.000	
14	38			14.000	14.000 to -14.000	0.000		2.000	
15	38			14.000	0.000	14.000 to -14.000		2.000	
16	41		0.95	-6.000 to 24.000	0.000	-14.142		2.000	
17	41			-6.000 to 24.000	0.000	0.000		2.000	
18	41			-6.000 to 24.000	14.142	0.000		2.000	
19	40			14.000	14.000 to -14.000	0.000		2.000	
20	40			14.000	0.000	14.000 to -14.000		2.000	

TABLE II.- CONTINUED

	1	2	3	4	5	6	7	8	9	10
	Page No.	Part No.		Mach No.	XT Range inches	YT Range inches	ZT Range inches		Δ inches	
	21	43		1.00	-6.000 to 24.000	0.000	-14.142		2.000	
	22	43			-6.000 to 24.000	14.142	0.000		2.000	
	23	43			-6.000 to 24.000	14.142	0.000		2.000	
	24	42			14.000	14.000 to -14.000	0.000		2.000	
	25	42			14.000	0.000	14.000 to -14.000		2.000	
	26	45		1.025	-6.000 to 24.000	0.000	-14.142		2.000	
	27	45			-6.000 to 24.000	0.000	0.000		2.000	
	28	45			-6.000 to 24.000	14.142	0.000		2.000	
	29	44			14.000	14.000 to -14.000	0.000		2.000	
	30	44			14.000	0.000	14.000 to -14.000		2.000	

TABLE II.- CONCLUDED

	1	2	3	4	5	6	7	8	9	10
	Page No.	Part No.		Mach No.	XT Range inches	YT Range inches	ZT Range inches		Δ inches	
	31	47		1.05	-6.000 to 24.000	0.000	-14.142		2.000	
	32	47			-6.000 to 24.000	0.000	0.000		2.000	
	33	47			-6.000 to 24.000	14.142	0.000		2.000	
	34	46			14.000	14.000 to -14.000	0.000		2.000	
	35	46			14.000	0.000	14.000 to -14.000		2.000	
	36	48		1.10	14.000	0.000	14.000 to -14.000		2.000	

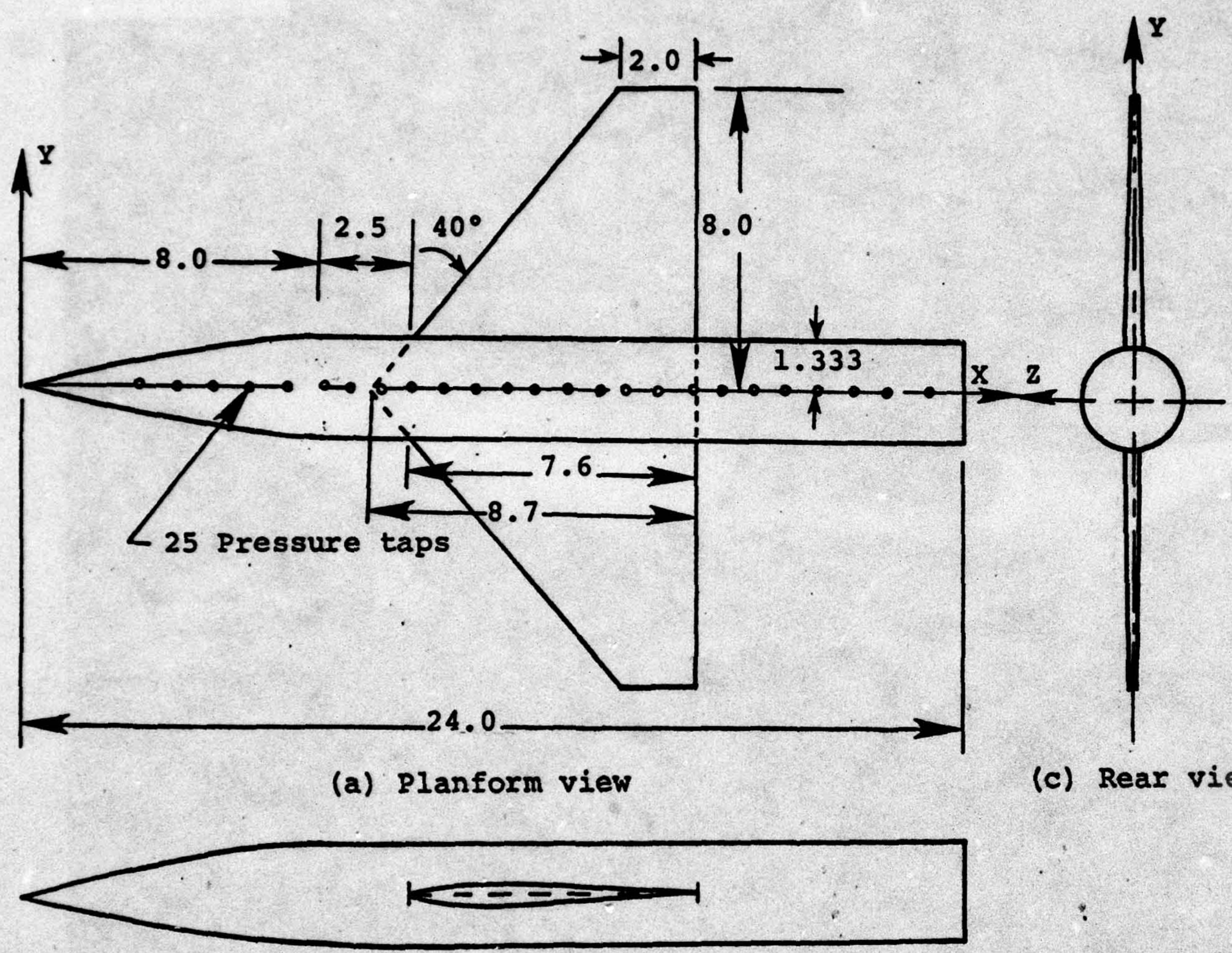
TABLE III.- MACH-NUMBER SURVEYS

1	2	3	4	5	6	7	8	9	10
Page No.	Part No.		Mach No.	Initial X inches	Final X inches	ΔX inches		Y inches	Z inches
37	85		0.80	10.333	19.000	0.333		3.000	-2.000
38	86		↓	↓	↓	↓		↓	-1.000
39	93		0.85						-2.000
40	94		↓						-1.000
41	96		0.90						-2.000
42	97		↓						-1.000
43	99		0.95						-2.000
44	100		↓						-1.000
45	102		1.00						-2.000
46	103		↓						-1.000
47	105		1.025						-2.000
48	106		↓						-1.000
49	108		1.050						-2.000
50	109		↓						-1.000
51	111		1.10						-2.000
52	112		↓						-1.000
53	114		1.15						-2.000
54	115		↓						-1.000
							α		
55	928		0.85	11.000	19.000	0.666	5.0	4.000	-1.000
56	929		↓	↓	↓	↓	-5.0	↓	↓

TABLE IV.- FORCE AND MOMENT AND PRESSURE TESTS

1	2	3	4	5	6	7	8	9	10
Page No.	Part No.		Mach No.		Type of Data		Wing		
57	73		0.80		F & M		4%		
58	↓		↓		P		↓		
59	74		0.85		F & M				
60	↓		↓		P				
61	75		0.90		F & M				
62	↓		↓		P				
63	938		0.925		F & M				
64	↓		↓		P				
65	76		0.95		F & M				
66	↓		↓		P				
67	939		0.975		F & M				
68	↓		↓		P				
69	77		1.00		F & M				
70	↓		↓		P				
71	78		1.025		F & M				
72	↓		↓		P				
73	79		1.05		F & M				
74	↓		↓		P				
75	80		1.10		F & M				
76	↓		↓		P				
77	81		1.15		F & M				
78	↓		↓		P		↓		
79	945		0.925		F & M		6%		
80	↓		↓		P		↓		
81	946		0.975		F & M				
82	↓		↓		P				
83	947		1.025		F & M				
84	↓		↓		P		↓		

Figure 1.- Conical flow-field survey probe on CTS and wing-body combination with 4-percent thick airfoil.



(a) Planform view

(b) Side view

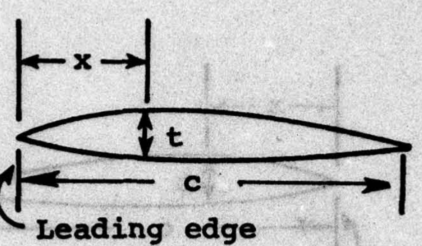
(c) Rear view

BODY COORDINATES

x, in.	r, in.
0	0
0.5	0.162
1.0	0.313
1.5	0.453
2.0	0.583
2.5	0.703
3.0	0.813
3.5	0.912
4.0	1.000
4.5	1.078
5.0	1.146
5.5	1.203
6.0	1.250
6.5	1.287
7.0	1.313
7.5	1.328
8.0	1.333
24.0	1.333

ALL DIMENSIONS IN INCHES

Figure 2.-Wing-body combination.



x/c, %	t/2c, %
0	0
2.5	0.325
5.0	0.548
7.5	0.736
10.0	0.900
15.0	1.175
20.0	1.399
25.0	1.576
30.0	1.726
35.0	1.837
40.0	1.921
45.0	1.974
50.0	1.998
55.0	1.989
60.0	1.955
65.0	1.885
70.0	1.777
75.0	1.620
80.0	1.406
85.0	1.085
90.0	0.738
95.0	0.369
100.0	0

$t_{max} = 2(7.6)\ (0.01998)$
$= 0.304"$ @ root

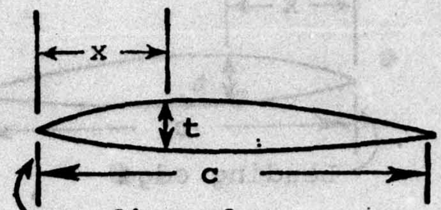
$t_{max} = 2(2)\ (0.01998)$
$= 0.080"$ @ tip

Leading-edge radius: 0.1 percent c

Trailing-edge radius: 0.01 percent c

(a) 4-percent thick airfoil

Figure 3.-Coordinates of airfoil sections.



x/c, %	t/2c, %
0	0
0.5	0.464
0.75	0.563
1.25	0.718
2.5	0.981
5.0	1.313
7.5	1.591
10.0	1.824
15.0	2.194
20.0	2.474
25.0	2.687
30.0	2.842
35.0	2.945
40.0	2.996
45.0	2.992
50.0	2.925
55.0	2.793
60.0	2.602
65.0	2.364
70.0	2.087
75.0	1.775
80.0	1.437
85.0	1.083
90.0	0.727
95.0	0.370
100.0	0.013

$$t_{max} = 2(7.6)\ (2.996)$$
$$= 0.455" \quad @ \text{ root}$$

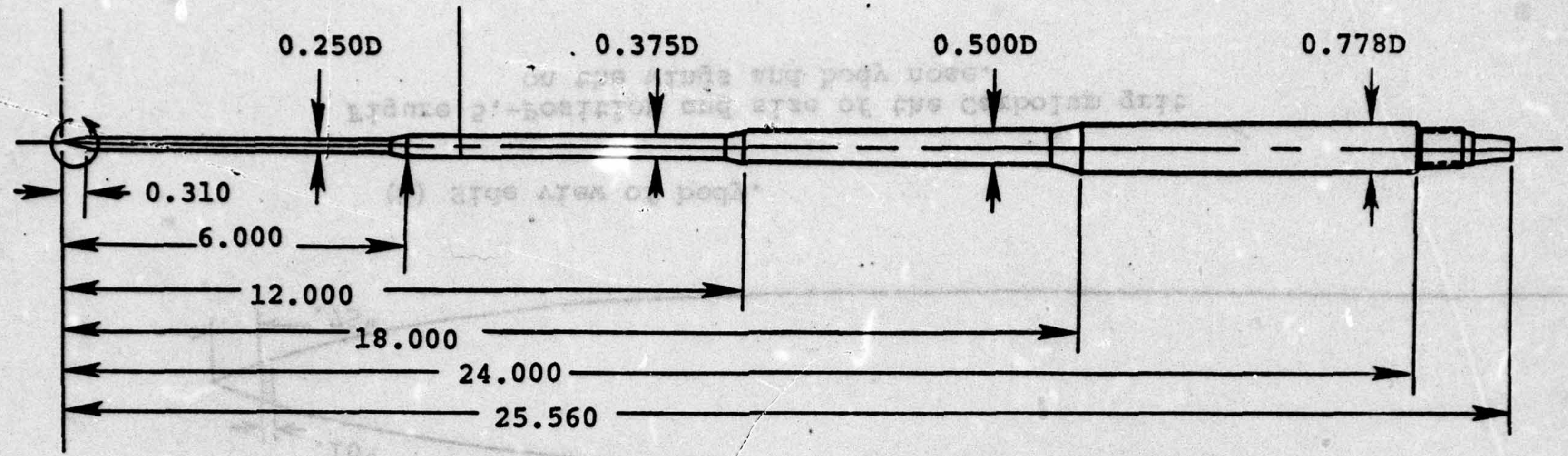
$$t_{max} = 2(2)\ (2.996)$$
$$= 0.120" \quad @ \text{ tip}$$

Leading-edge radius: 0.229 percent c

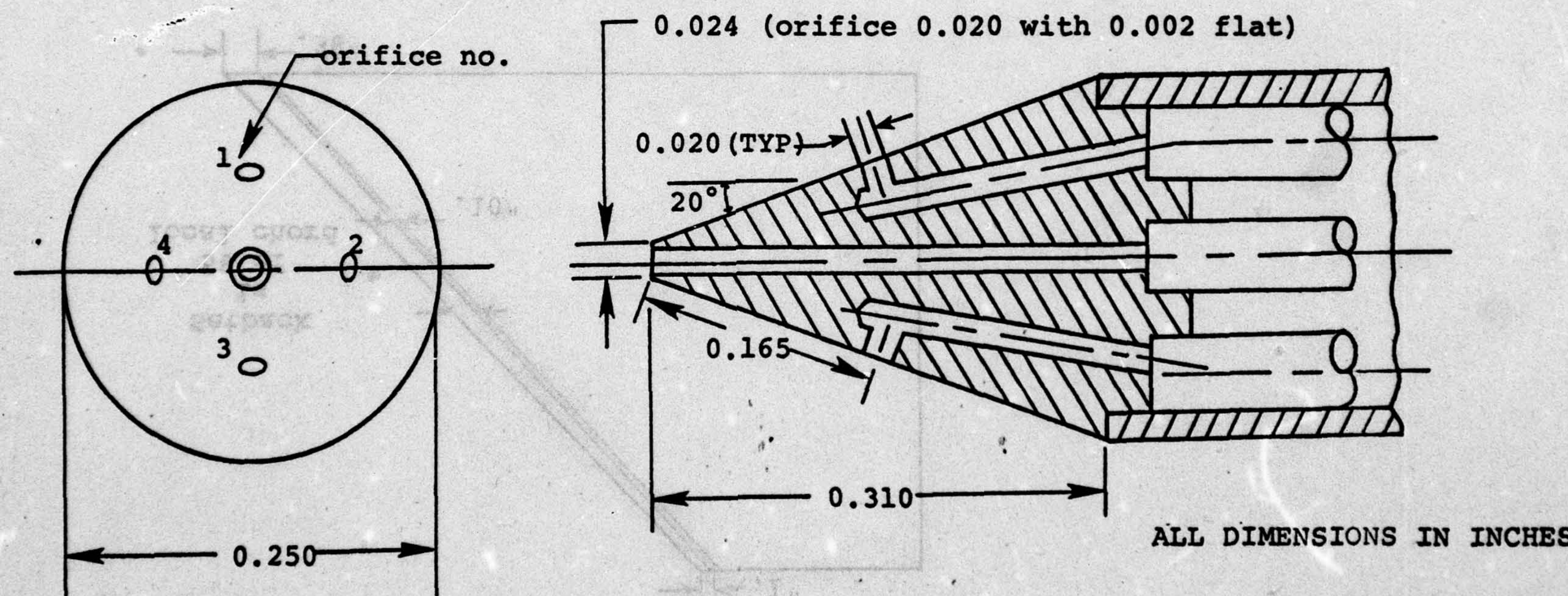
Trailing-edge radius: 0.014 percent c

(b) 6-percent thick airfoil (NACA 65A006)

Figure 3.-Concluded.

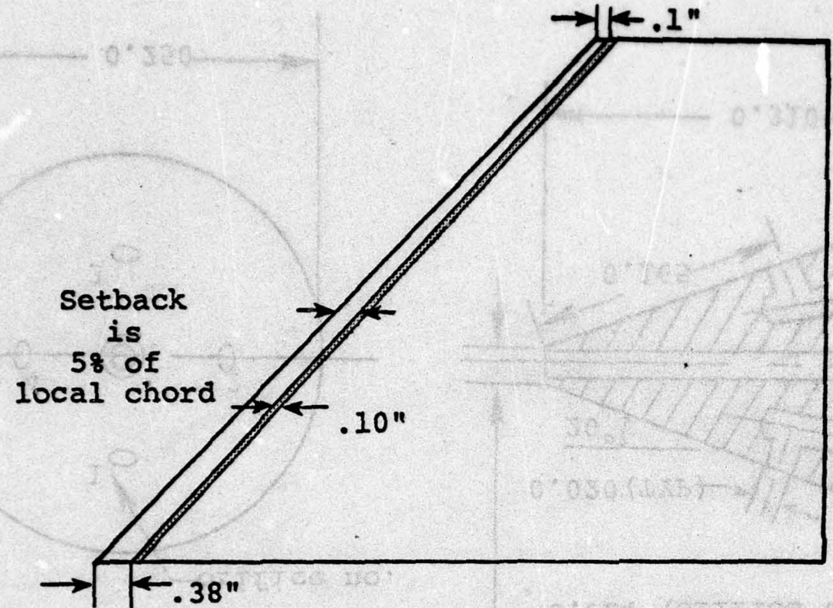


(a) Overall view of the probe.

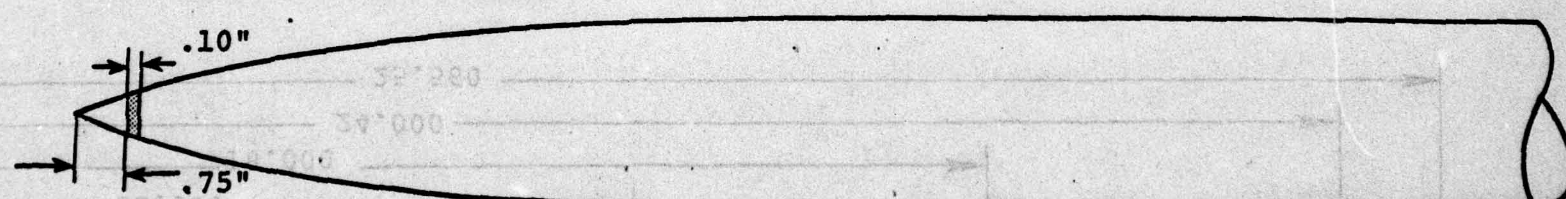


(b) Details of the 40° probe tip.

Figure 4.-Details and Dimensions of the 40° apex angle probe.

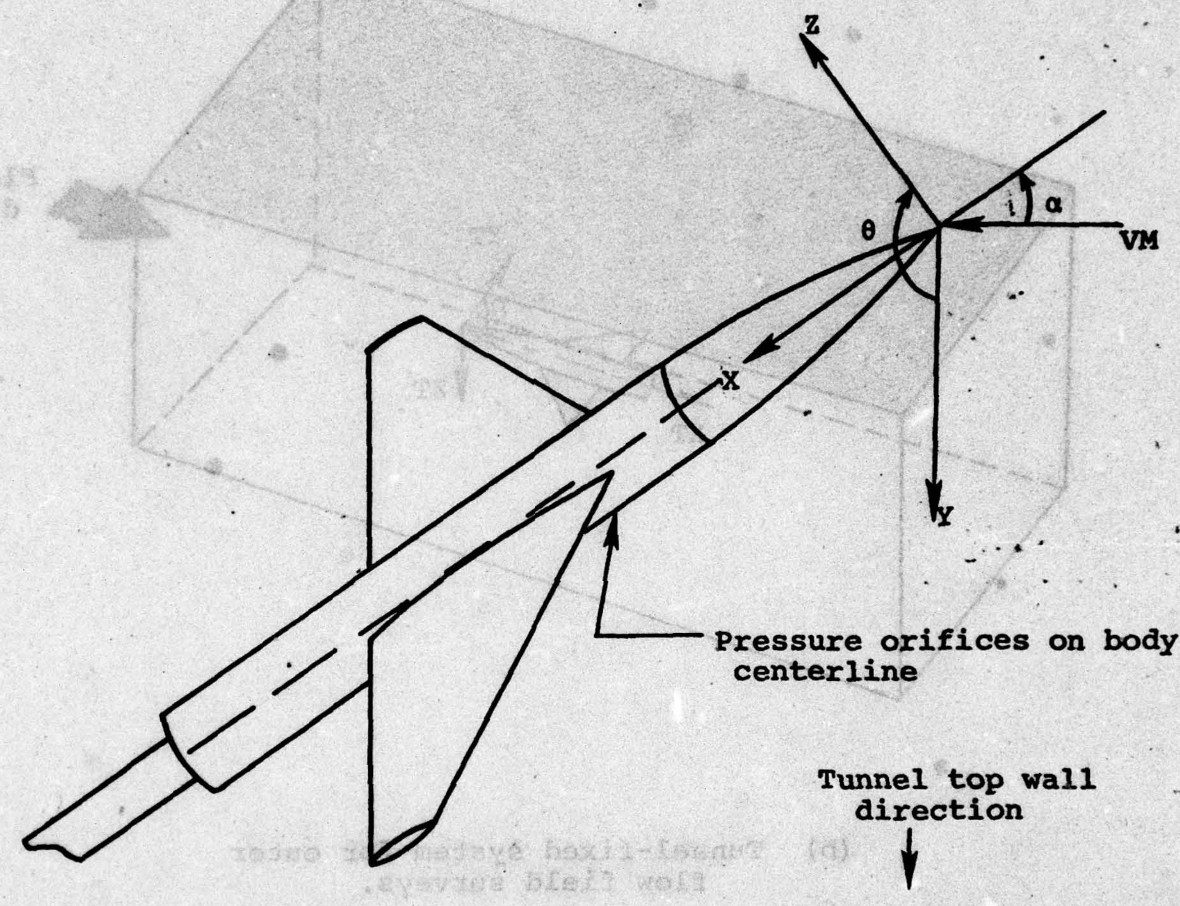


(a) Planform view of wing.



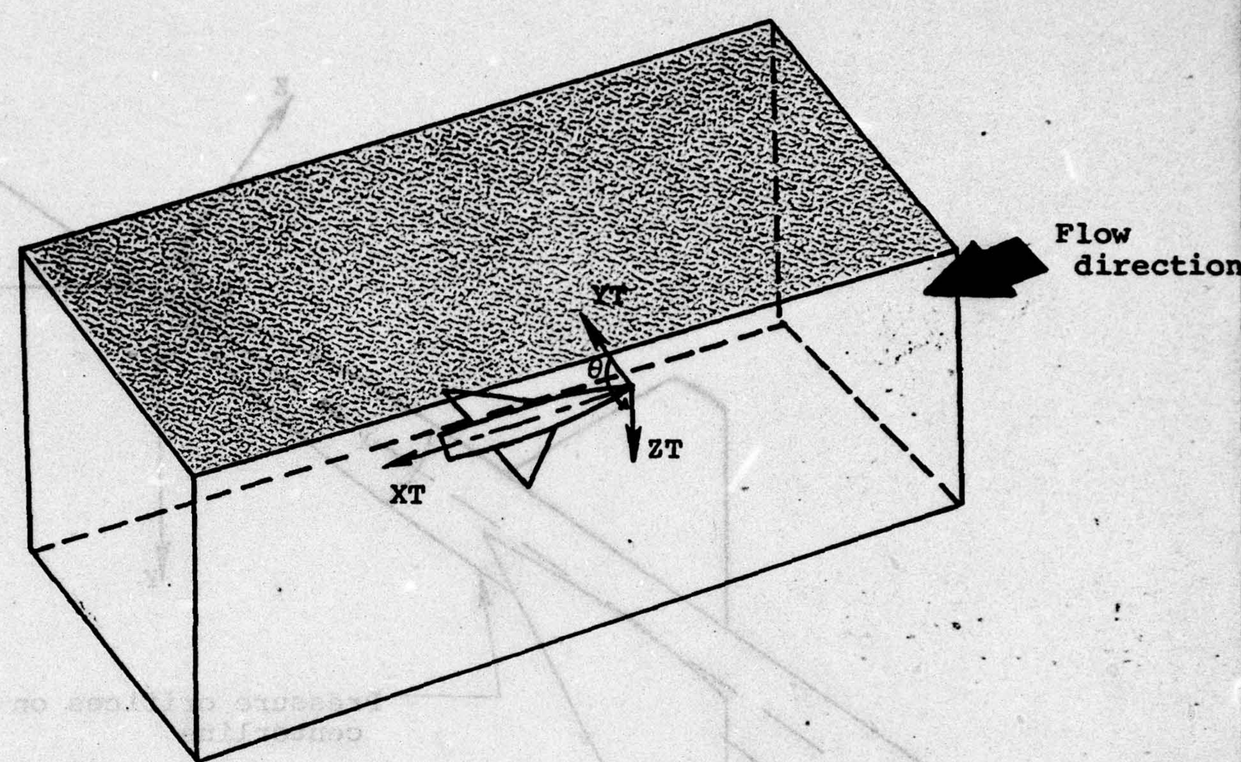
(b) Side view of body.

Figure 5.-Position and size of the Carbolum grit on the wings and body nose.



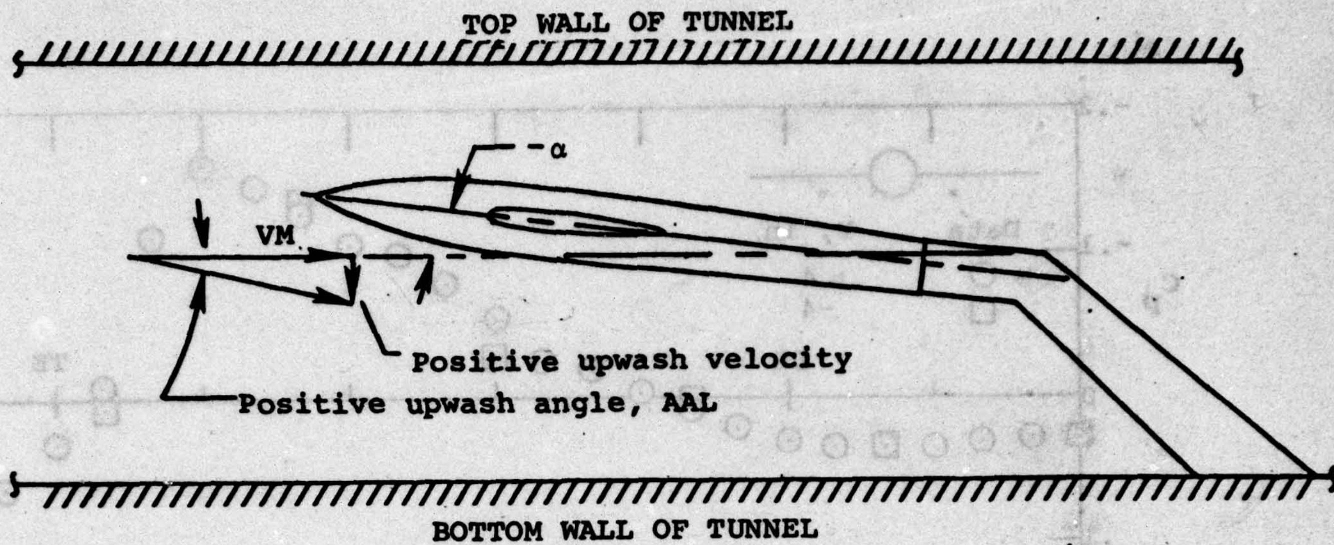
(a) Body-fixed system for inner flow field surveys.

Figure 6.-Coordinate systems.

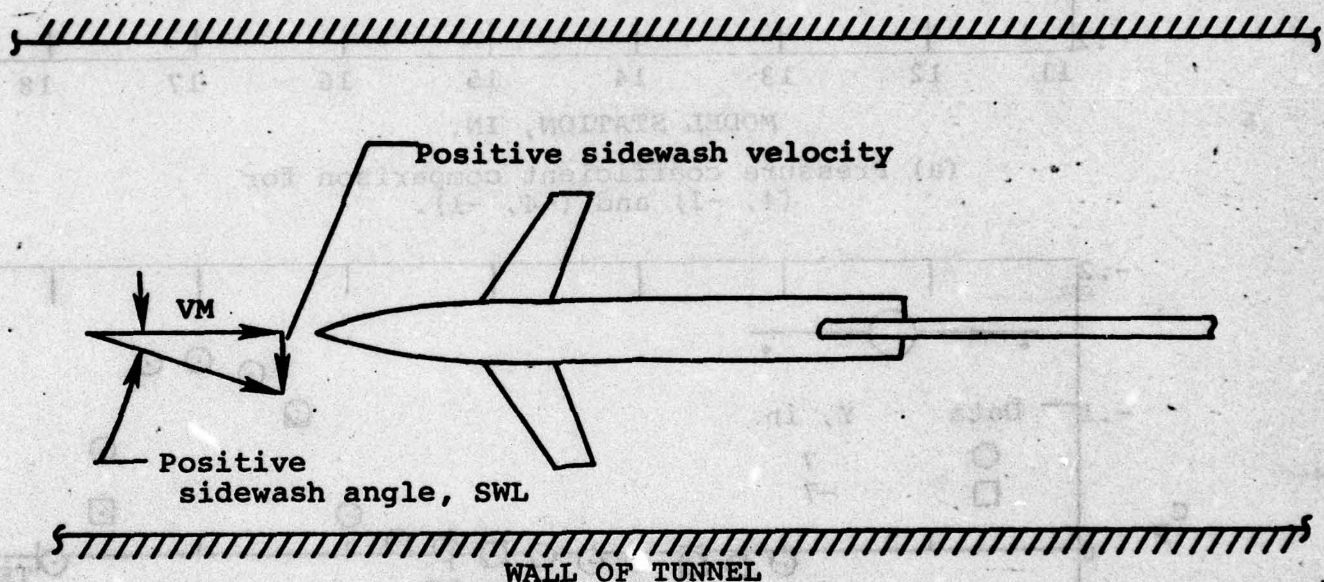


(b) Tunnel-fixed system for outer flow field surveys.

Figure 6.- Concluded.

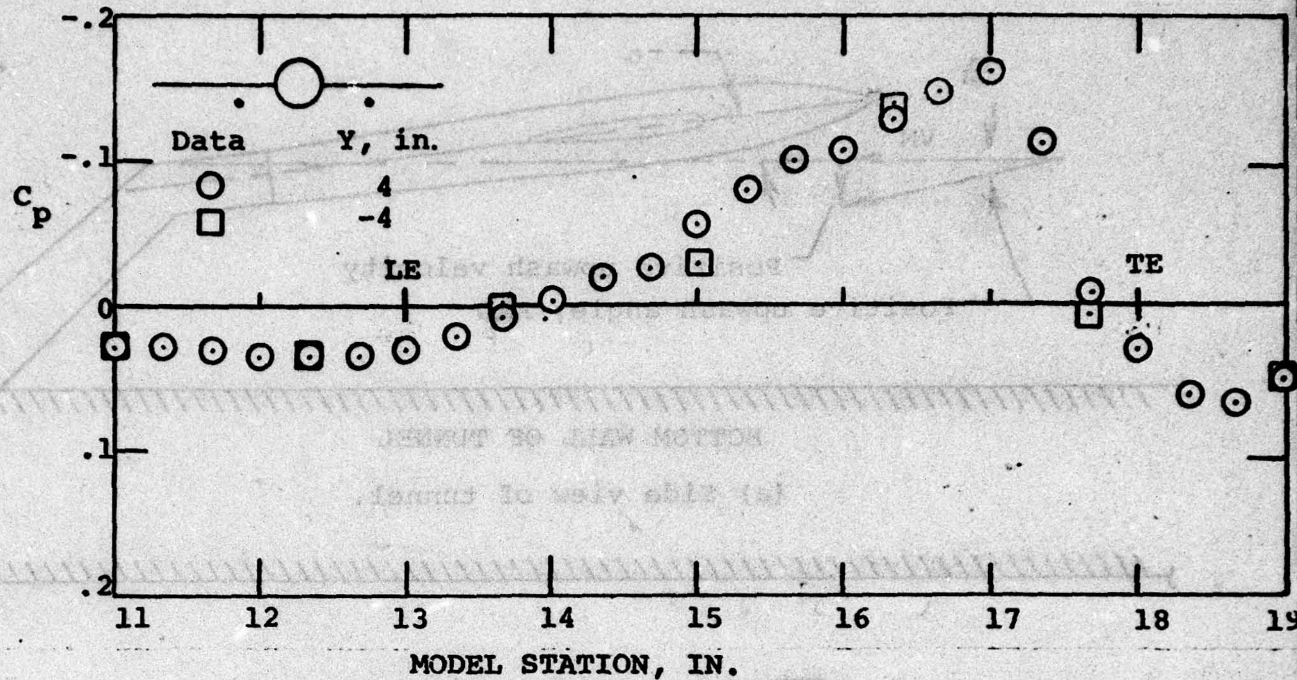


(a) Side view of tunnel.

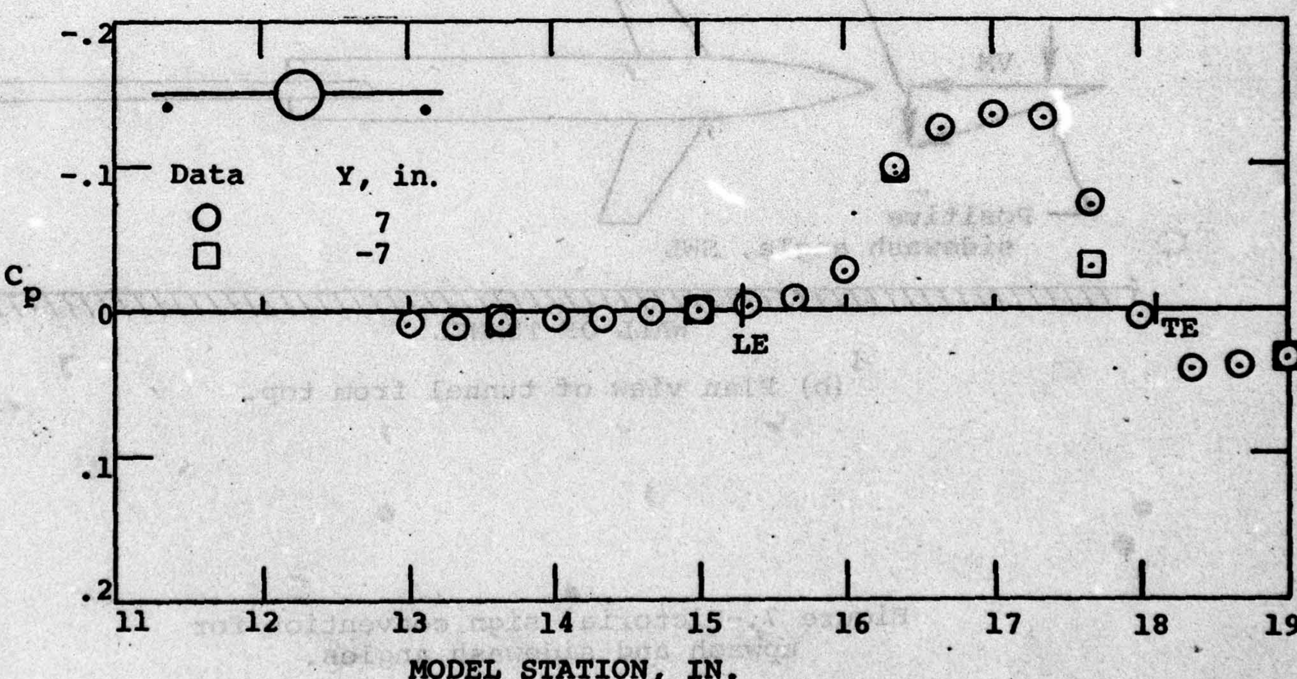


(b) Plan view of tunnel from top.

Figure 7.-Pictorial sign convention for upwash and sidewash angles.

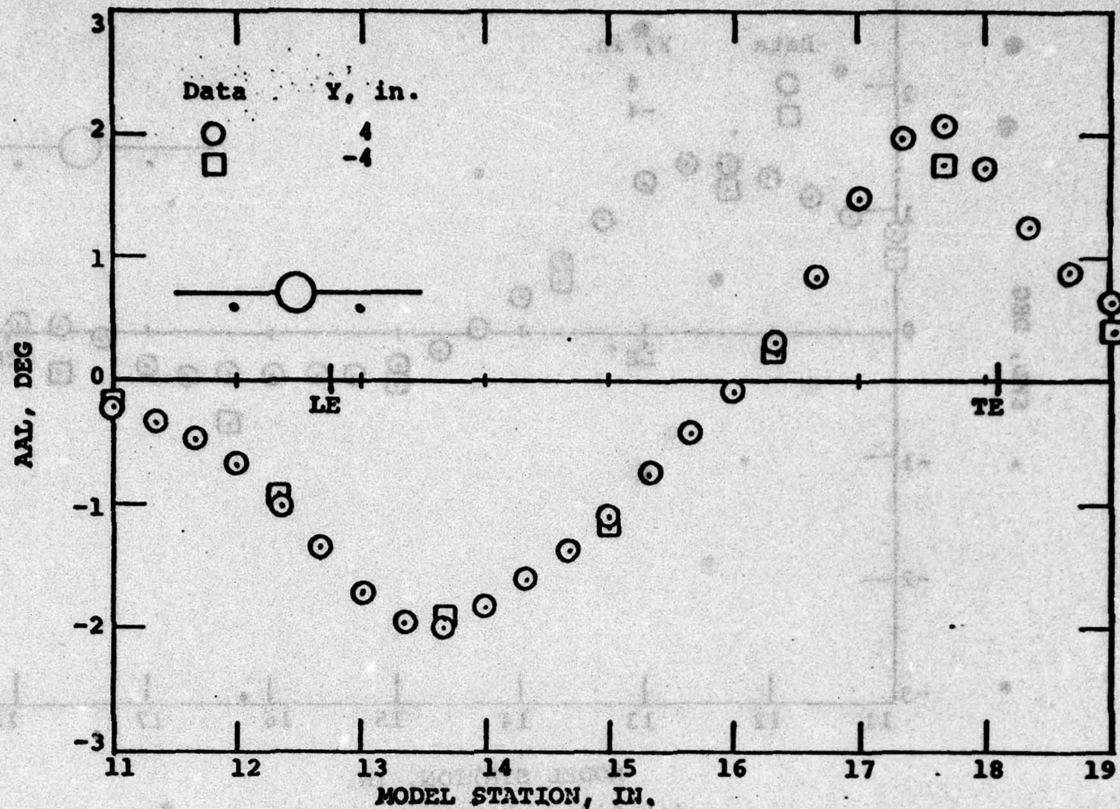


(a) Pressure coefficient comparison for (4, -1) and (-4, -1).

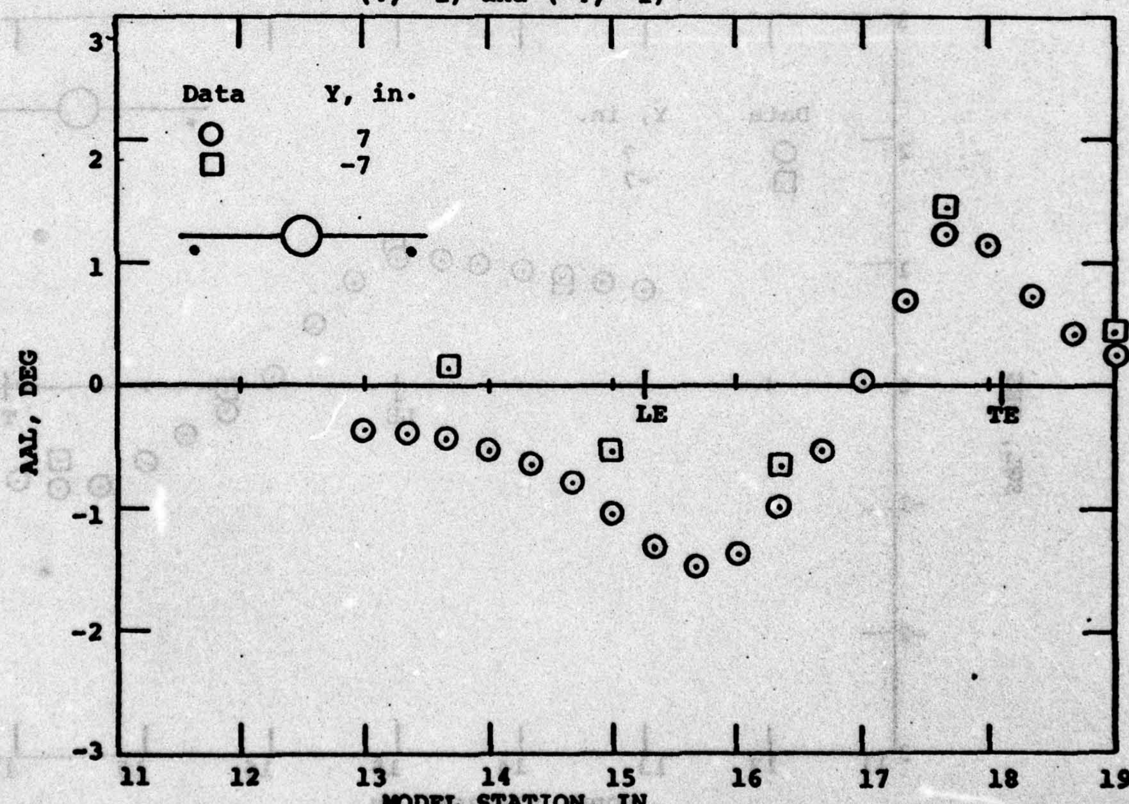


(b) Pressure coefficient comparison for (7, -1) and (-7, -1).

Figure 8.-Symmetry comparisons for 4-percent thick wing-body combination at $Z = -1.0$ in., $\alpha = 0°$, $M_\infty = 0.925$.

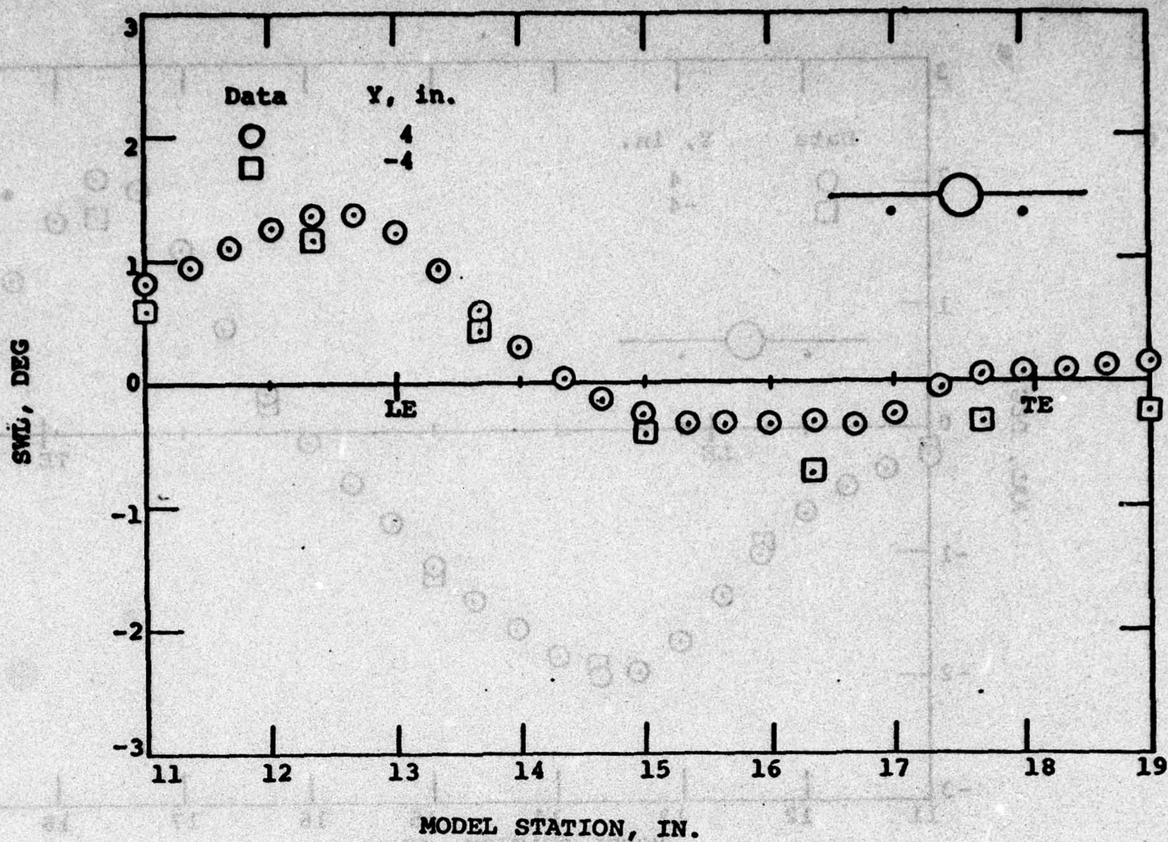


(c) Upwash comparison for
(4, -1) and (-4, -1).

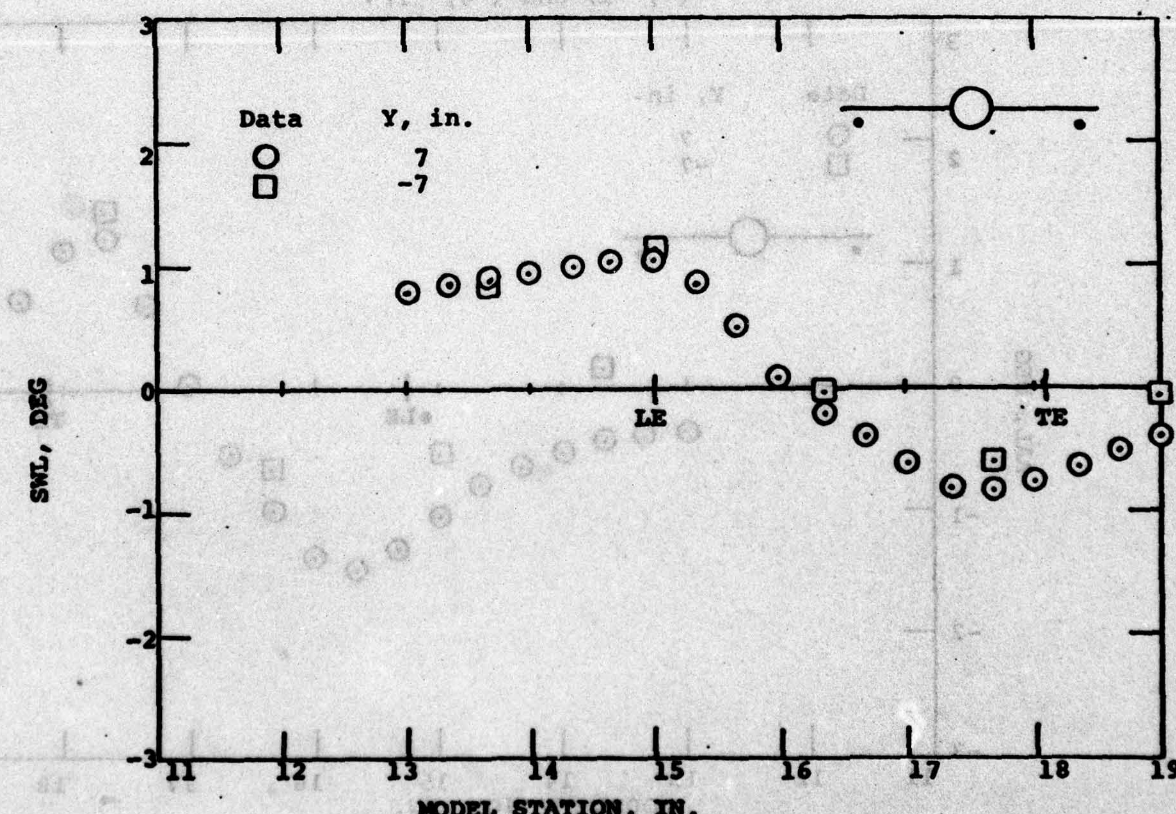


(d) Upwash comparison for
(7, -1) and (-7, -1).

Figure 8.-Continued.

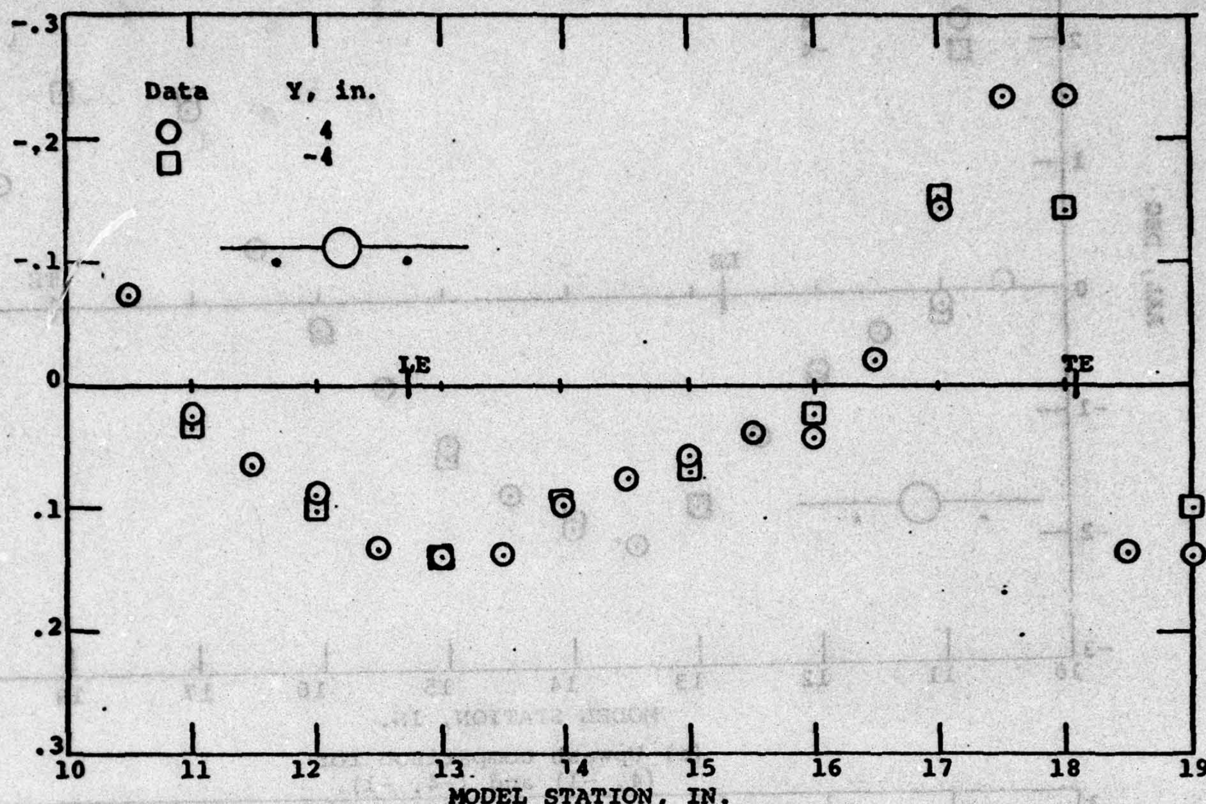


(e) Sidewash comparison for
(4, -1) and (-4, -1).

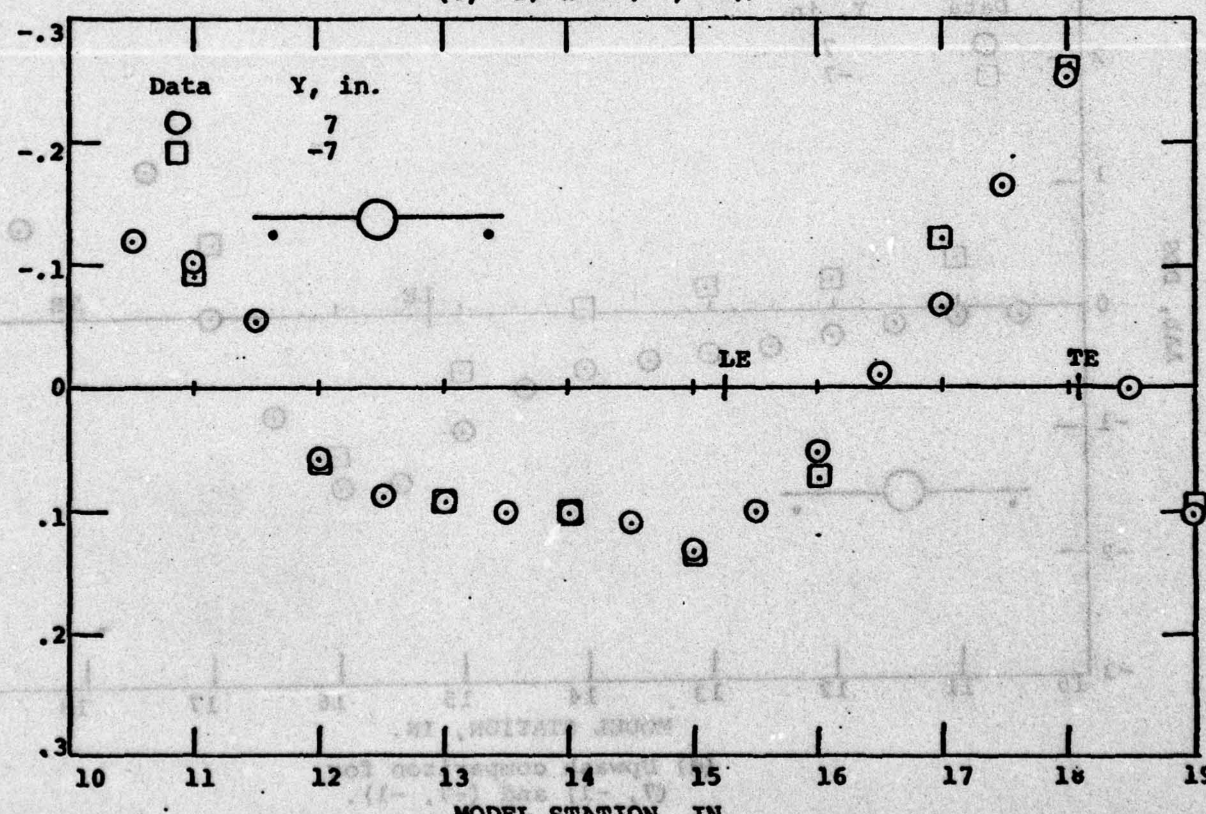


(f) Sidewash comparison for
(7, -1) and (-7, -1).

Figure 8.-Concluded.

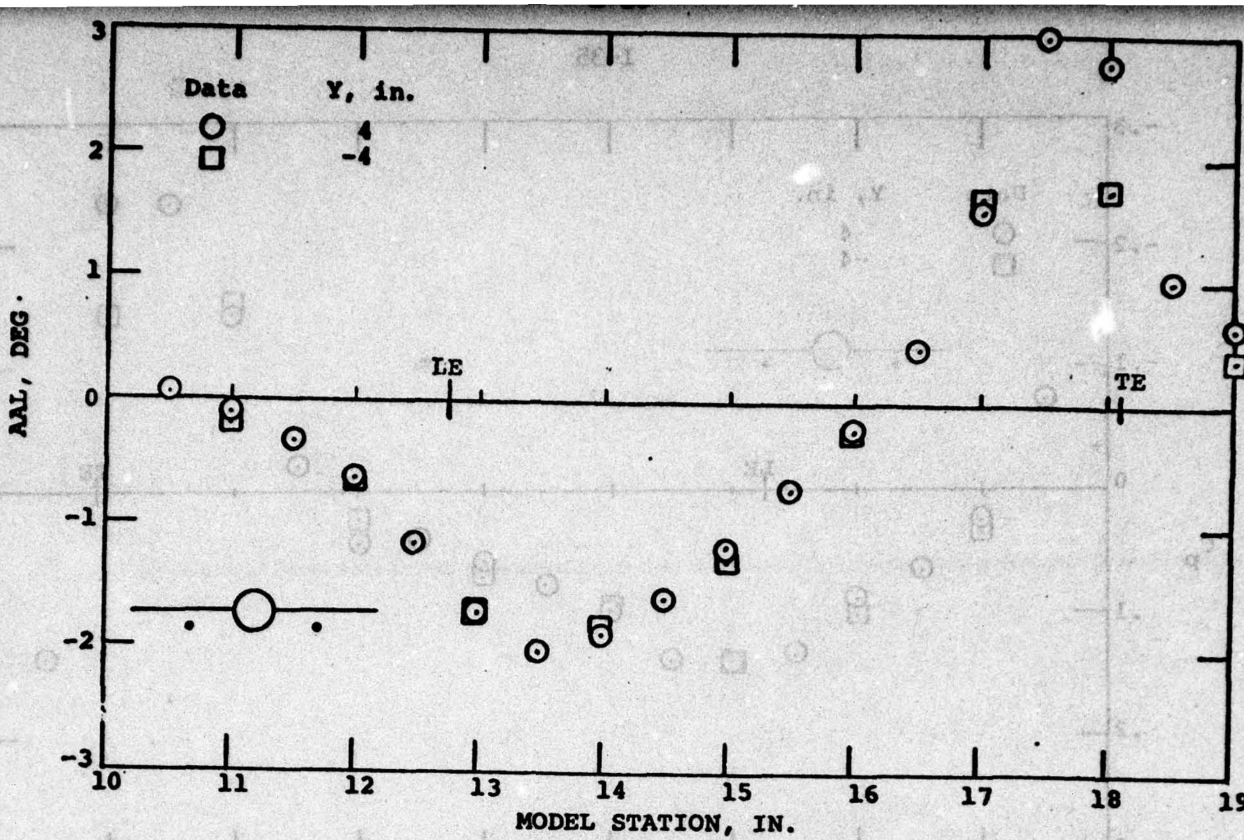


(a) Pressure coefficient comparison for (4, -1) and (-4, -1).

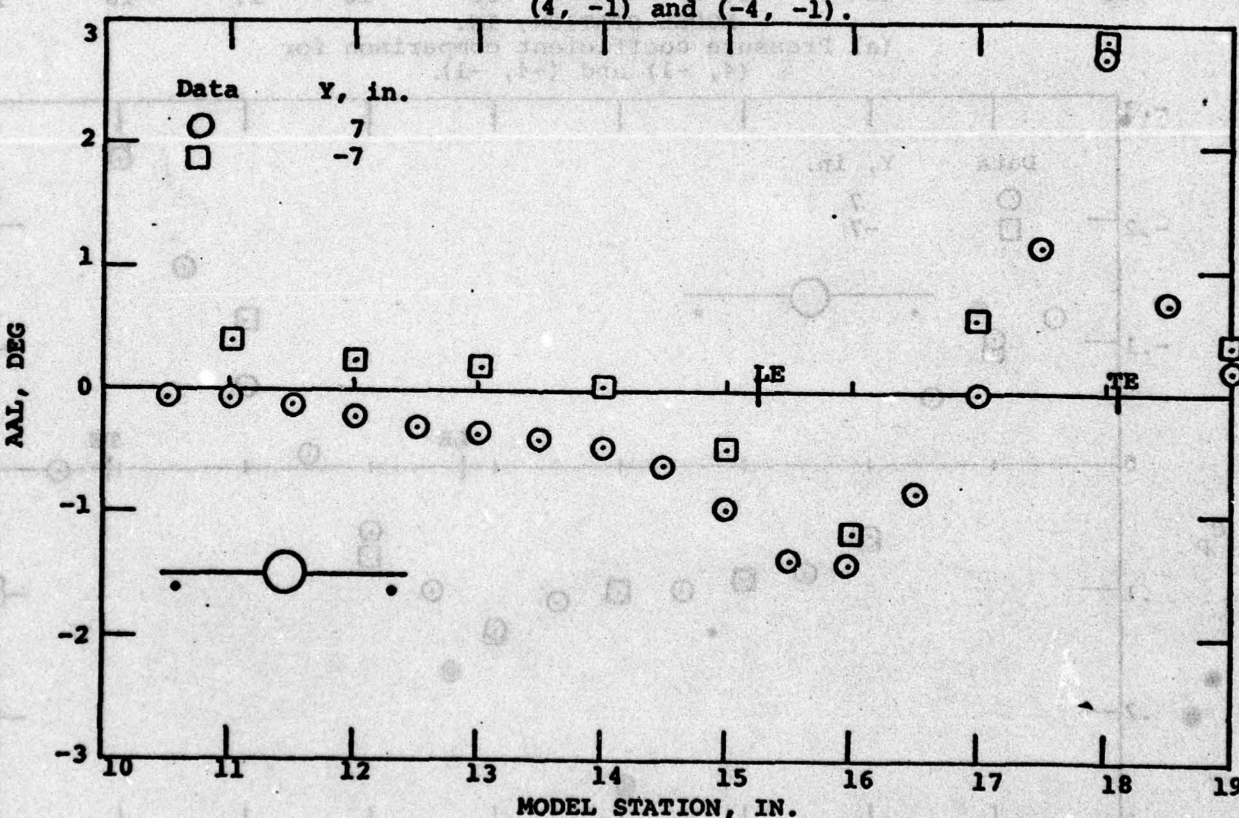


(b) Pressure coefficient comparison for (7, -1) and (-7, -1).

Figure 9.-Symmetry comparisons for 4-percent thick wing-body combination at $Z = -1.0$ in., $\alpha = 0°$, $M_\infty = 1.025$.

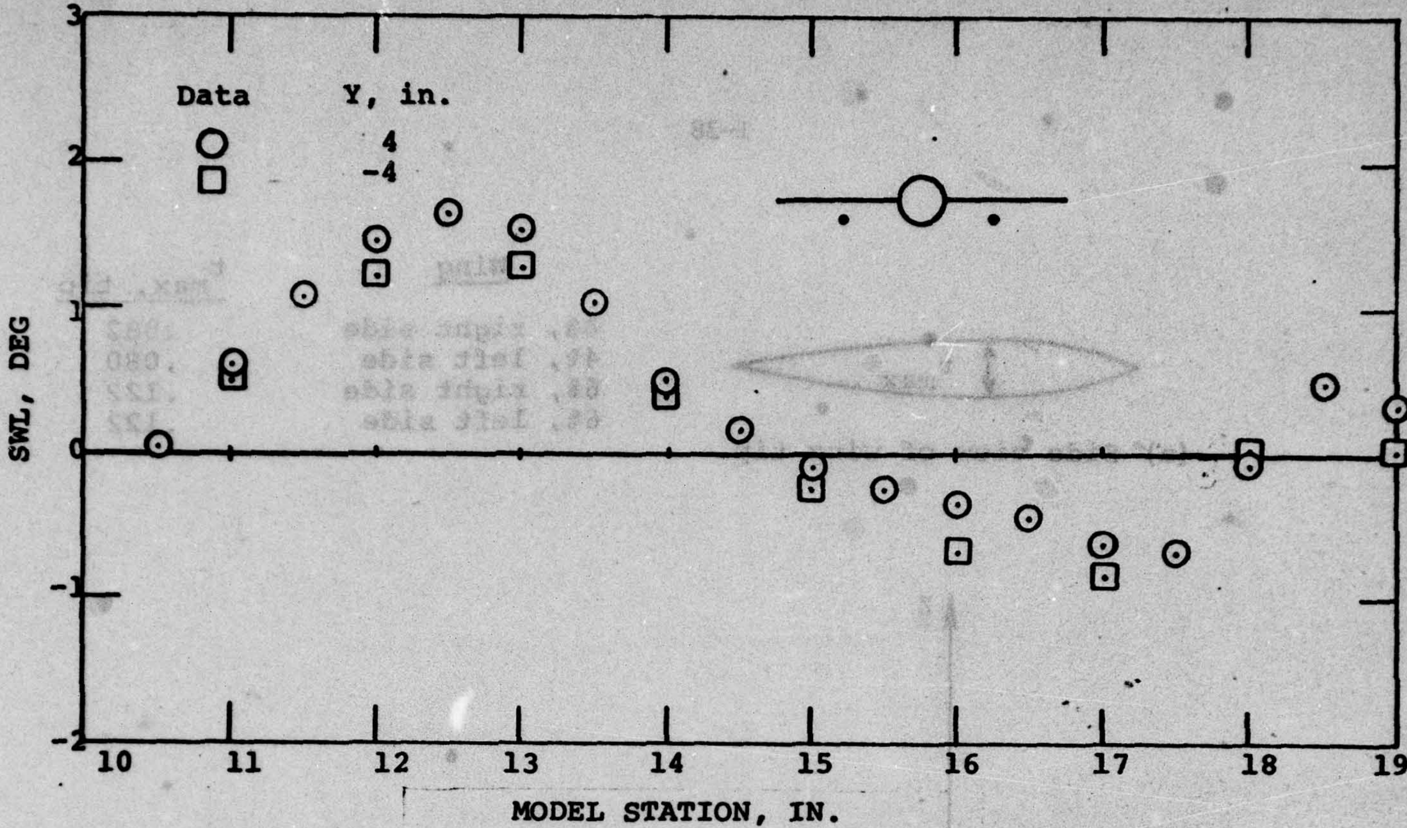


(c) Upwash comparison for
(4, -1) and (-4, -1).

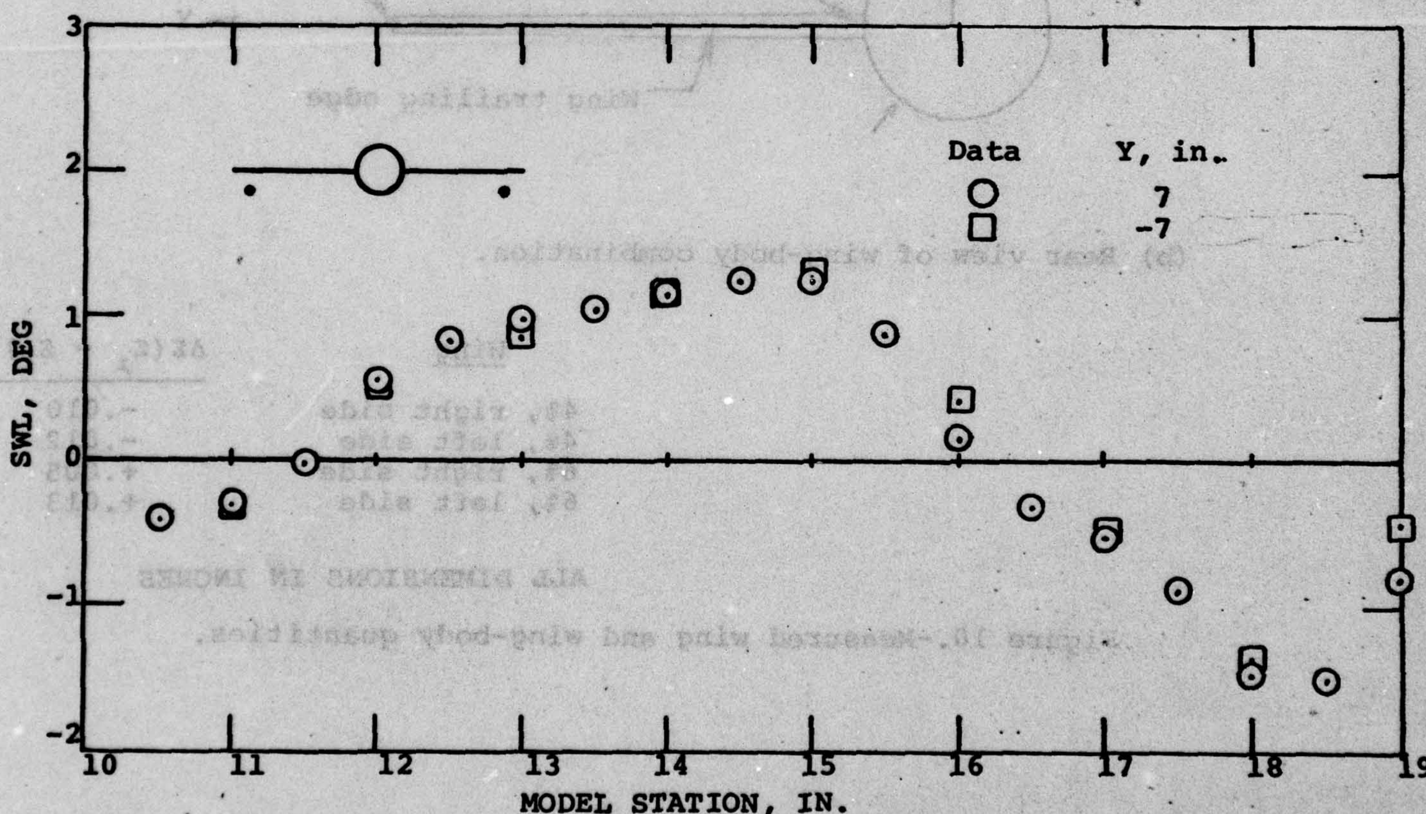


(d) Upwash comparison for
(7, -1) and (-7, -1).

Figure 9.- Continued.

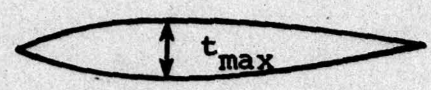


(e) Sidewash comparison for (4, -1) and (-4, -1).



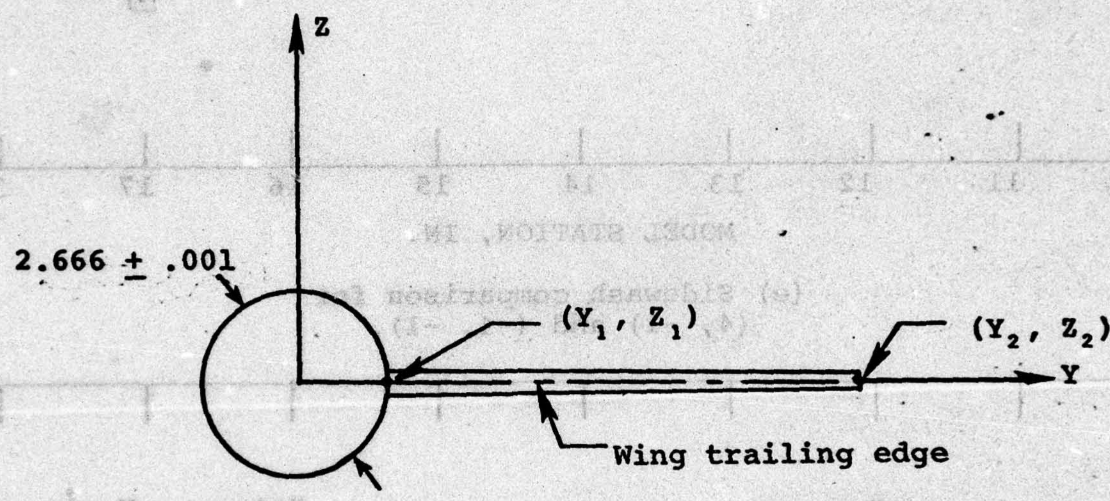
(f) Sidewash comparison for (7, -1) and (-7, -1).

Figure 9.-Concluded.



Wing	$t_{max, tip}$
4%, right side	.082
4%, left side	.080
6%, right side	.122
6%, left side	.122

(a) Side view of wing tip.

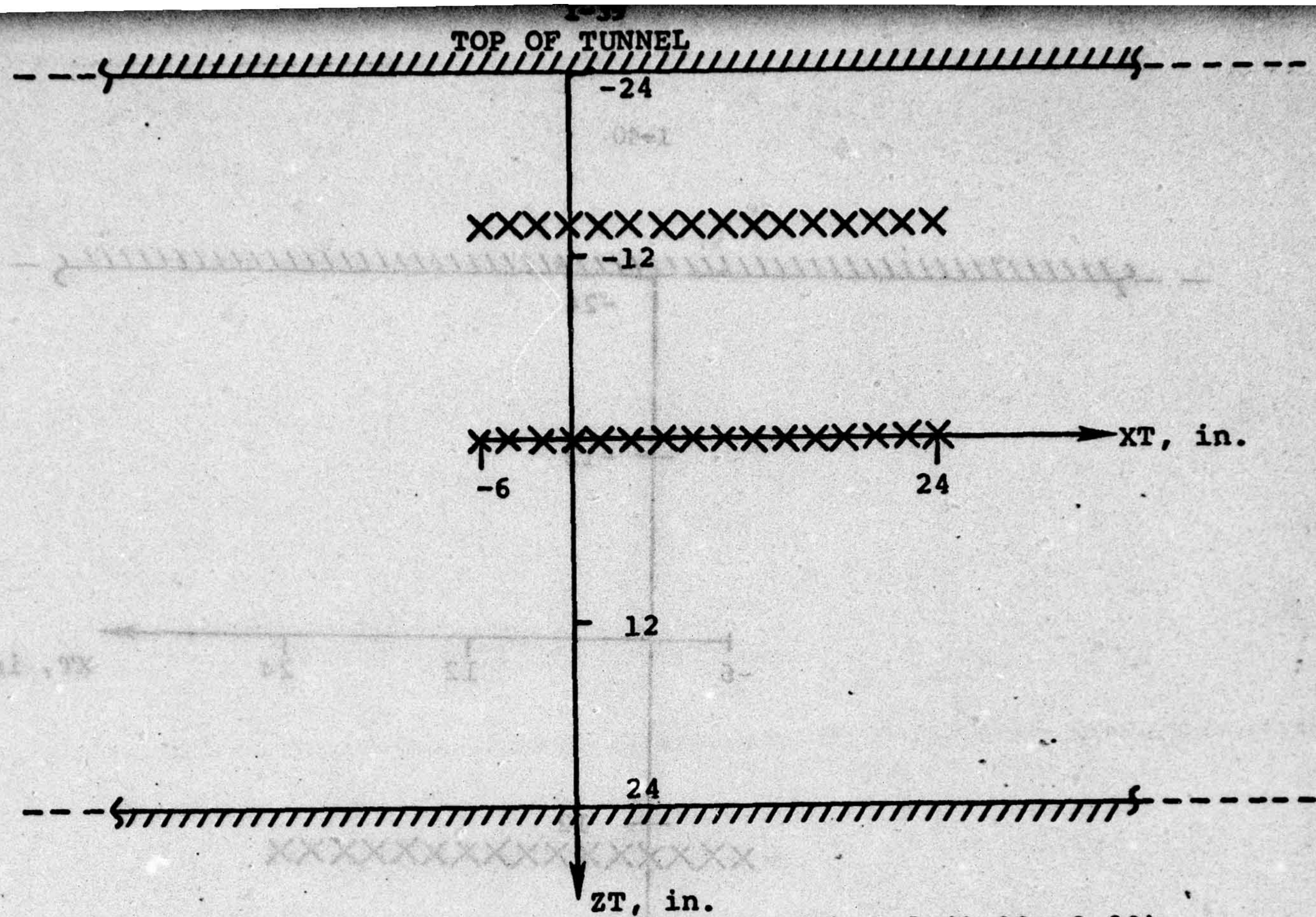


(b) Rear view of wing-body combination.

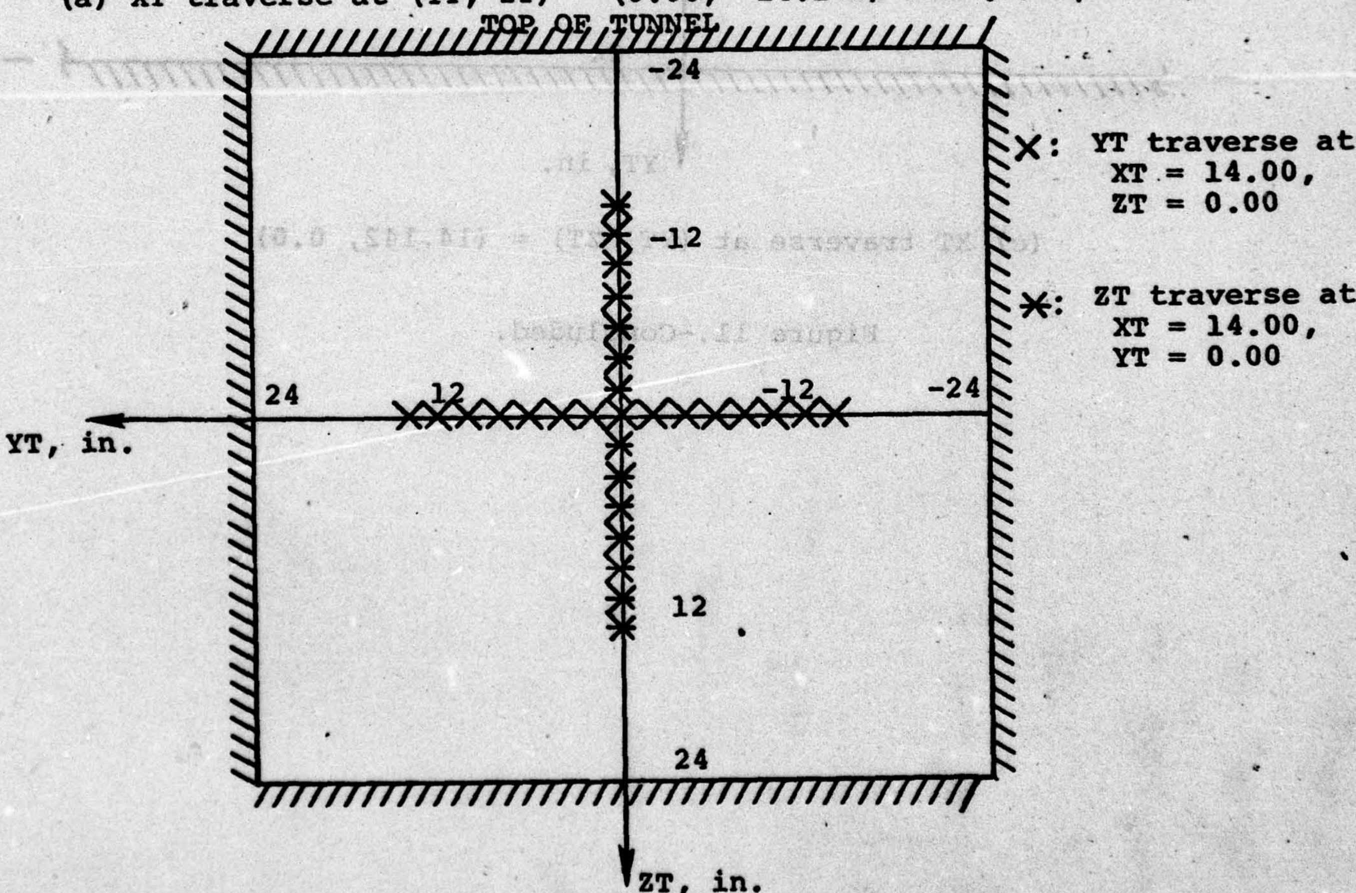
Wing	$\Delta Z (Z_2 - Z_1)$
4%, right side	-.010
4%, left side	-.012
6%, right side	+.005
6%, left side	+.013

ALL DIMENSIONS IN INCHES

Figure 10.-Measured wing and wing-body quantities.

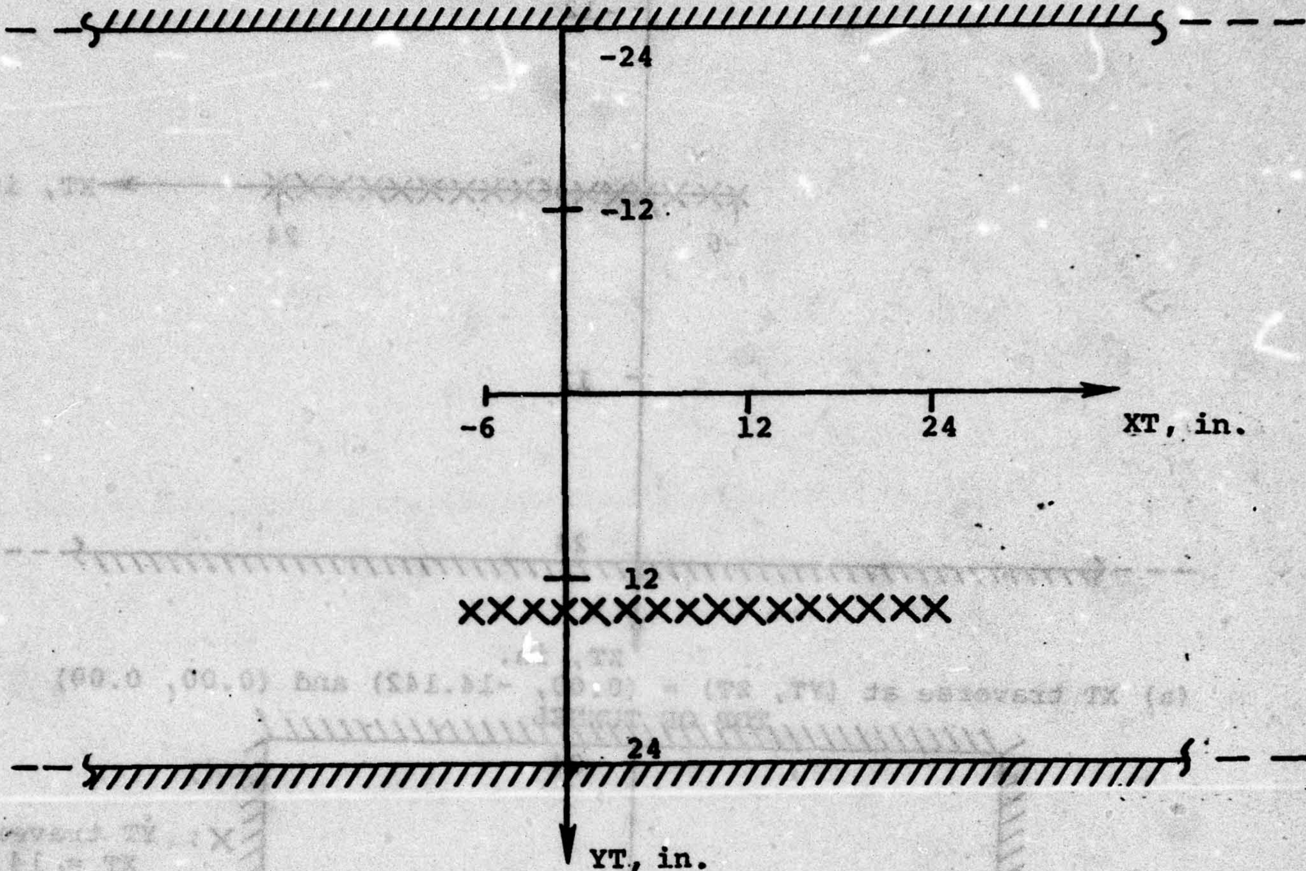


(a) XT traverse at (YT, ZT) = (0.00, -14.142) and (0.00, 0.00)



(b) YT and ZT traverses

Figure 11.- Tunnel-Empty Survey grids.



(c) XT traverse at (YT, ZT) = (14.142, 0.0)

Figure 11.-Concluded.

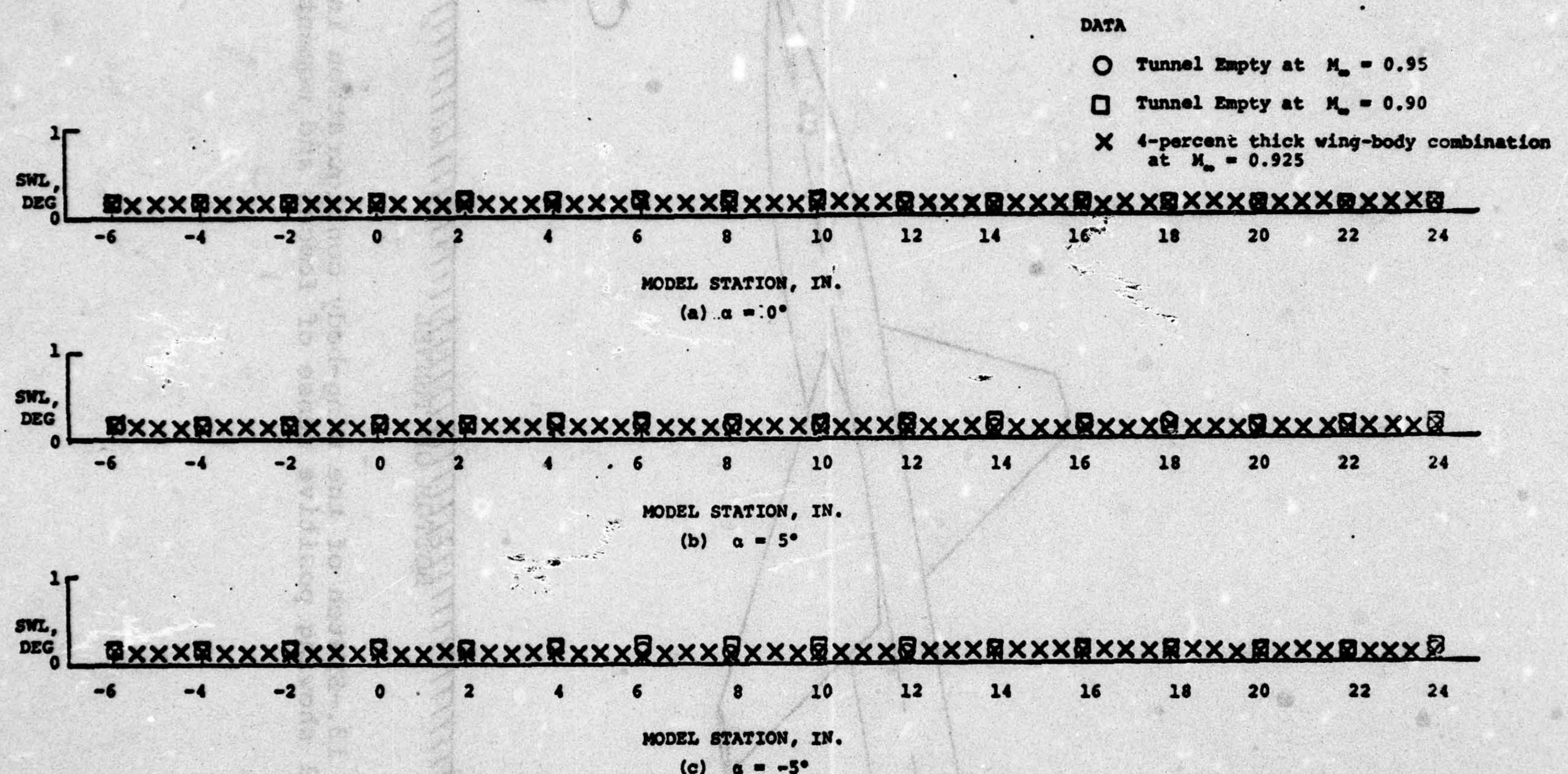


Figure 12.-Local Sidewash SWL at YT = 0.0 in., ZT = -14.14 in.

41

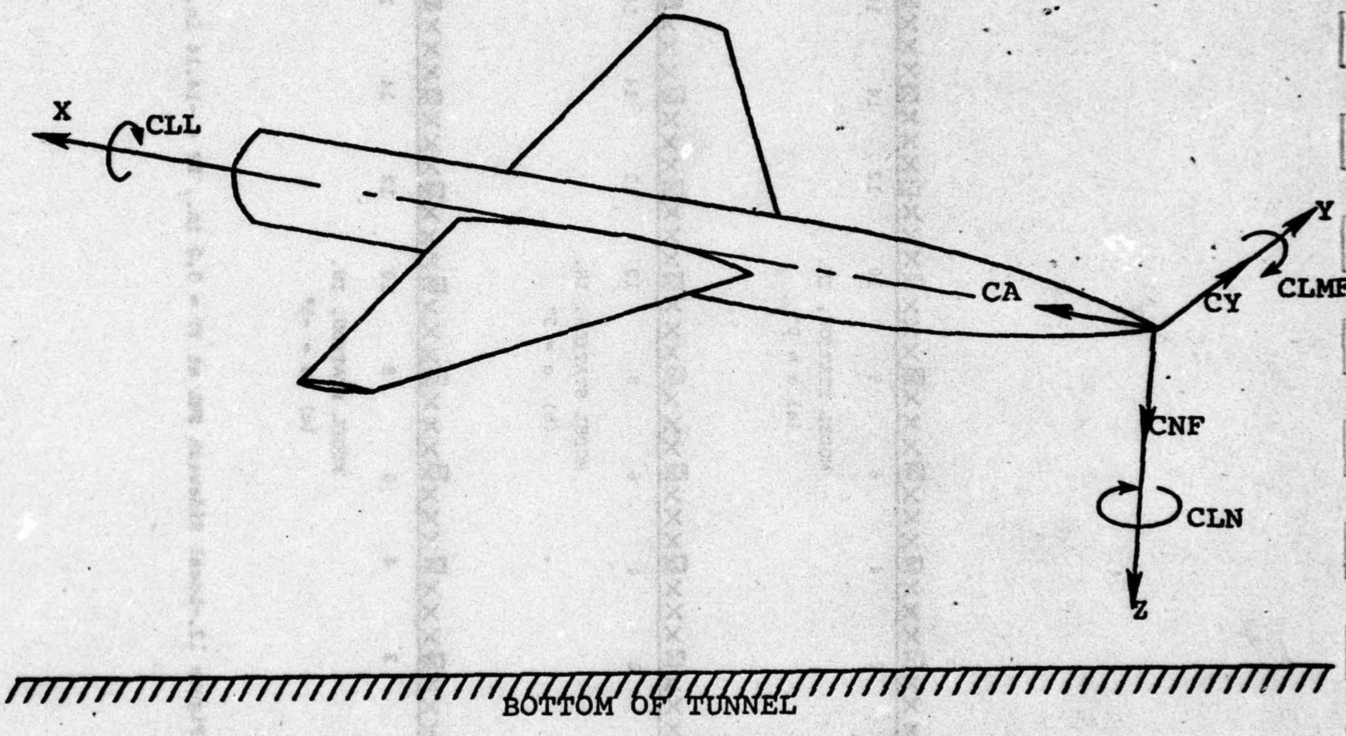


Figure 13.-Sketch of the wing-body configuration in the tunnel showing positive sense of forces and moments.

TEST TC-484 | PART 34 | RCX10-6 3.007 | ALFA **** | WING NONE | YT 0.00 | ZT -14.14 | PWW SURVEY 1-701 | DATE 2- 2-77 | AEDC PROPULSION WIND TUNNEL TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BNTL
5	-6.000	0.802	861.76	1544.9	455.7	82.1	0.806	1.004	1.000	-0.008	1.004	0.003	0.007	0.41	0.18
6	-4.000	0.806	865.97	1541.1	457.3	82.7	0.811	1.006	1.001	-0.010	1.006	0.003	0.007	0.42	0.15
7	-2.000	0.798	858.70	1544.0	453.0	82.9	0.806	1.009	1.001	-0.017	1.009	0.003	0.008	0.43	0.16
8	0.000	0.798	858.49	1546.0	453.2	82.9	0.807	1.009	1.001	-0.015	1.009	0.003	0.007	0.40	0.16
9	2.000	0.801	861.37	1547.5	455.5	83.4	0.806	1.006	1.001	-0.011	1.006	0.003	0.006	0.34	0.17
10	4.000	0.800	863.73	1548.1	455.3	83.2	0.809	1.009	1.000	-0.018	1.009	0.003	0.005	0.28	0.16
11	6.000	0.802	861.88	1547.7	456.2	82.9	0.808	1.007	1.001	-0.012	1.007	0.003	0.005	0.26	0.15
12	8.000	0.805	865.18	1546.7	457.9	83.3	0.812	1.007	1.001	-0.013	1.007	0.003	0.004	0.22	0.18
13	10.000	0.804	864.30	1545.3	456.9	83.4	0.811	1.008	1.001	-0.014	1.008	0.003	0.003	0.20	0.16
14	12.000	0.803	863.05	1547.2	455.7	83.2	0.808	1.006	1.001	-0.009	1.006	0.003	0.003	0.17	0.16
15	14.000	0.804	863.87	1548.8	457.8	83.2	0.807	1.004	1.001	-0.007	1.004	0.003	0.003	0.17	0.15
16	16.000	0.799	858.98	1547.9	454.3	82.7	0.803	1.005	1.001	-0.008	1.004	0.002	0.003	0.19	0.10
17	18.000	0.804	863.71	1546.0	456.8	83.3	0.807	1.004	1.000	-0.007	1.004	0.002	0.004	0.22	0.11
18	20.000	0.803	863.97	1547.6	457.2	83.8	0.805	1.002	1.001	-0.002	1.002	0.001	0.004	0.23	0.08
19	22.000	0.799	860.07	1550.9	455.4	83.6	0.797	0.997	1.001	0.008	0.997	0.001	0.005	0.28	0.08
20	24.000	0.800	859.94	1547.1	454.5	83.1	0.803	1.004	1.001	-0.005	1.004	0.002	0.008	0.29	0.09

TEST	PART	REX10-6	ALPHA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	34	3.001	****	NONE	0.00	0.00	1-702	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
22	-6.000	0.799	858.60	1547.9	454.0	82.7	0.809	1.011	0.999	-0.026	1.011	0.004	0.008	0.43	0.20
23	-4.000	0.799	859.72	1545.2	453.5	82.9	0.804	1.005	1.001	-0.008	1.005	0.003	0.007	0.37	0.18
24	-2.000	0.799	859.77	1546.1	453.9	83.3	0.804	1.005	1.001	-0.008	1.005	0.003	0.007	0.39	0.18
25	0.000	0.799	859.73	1548.2	454.6	83.2	0.803	1.004	1.000	-0.008	1.004	0.003	0.006	0.35	0.17
26	2.000	0.799	859.14	1548.4	454.1	83.5	0.804	1.006	1.001	-0.010	1.006	0.003	0.006	0.35	0.15
27	4.000	0.800	860.04	1547.7	454.9	82.9	0.802	1.002	1.002	-0.000	1.002	0.003	0.006	0.32	0.17
28	6.000	0.802	861.34	1544.6	455.2	82.3	0.808	1.007	1.001	-0.010	1.007	0.003	0.005	0.31	0.15
29	8.000	0.800	861.12	1543.9	454.0	83.8	0.805	1.005	1.001	-0.007	1.005	0.002	0.006	0.33	0.12
30	10.000	0.799	860.24	1550.1	454.9	84.4	0.802	1.003	1.000	-0.006	1.003	0.002	0.005	0.30	0.14
31	12.000	0.802	862.24	1550.4	456.9	83.4	0.807	1.005	1.001	-0.008	1.005	0.002	0.006	0.32	0.12
32	14.000	0.801	860.59	1547.6	455.3	82.7	0.807	1.007	1.001	-0.012	1.007	0.002	0.006	0.32	0.13
33	16.000	0.808	867.63	1545.1	459.4	83.2	0.809	1.002	1.001	-0.001	1.002	0.002	0.005	0.29	0.09
34	18.000	0.808	868.38	1545.1	459.8	83.4	0.810	1.002	1.001	-0.001	1.002	0.002	0.005	0.30	0.11
35	20.000	0.808	868.60	1548.4	460.5	84.2	0.814	1.006	1.001	-0.012	1.006	0.002	0.005	0.30	0.11
36	22.000	0.813	872.52	1548.2	463.7	83.3	0.817	1.005	1.001	-0.007	1.005	0.001	0.005	0.31	0.09
37	24.000	0.801	861.01	1548.6	455.3	83.4	0.803	1.003	1.001	-0.003	1.003	0.002	0.006	0.32	0.10

TEST	PART	RPX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	34	2.985	****	NONE	14.14	0.00	1-703	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
39	-6.000	0.798	859.87	1551.7	455.0	84.5	0.830	1.035	0.998	-0.075	1.035	0.004	0.008	0.46	0.20
40	-4.000	0.799	860.51	1552.0	455.6	84.4	0.802	1.003	1.001	-0.005	1.003	0.002	0.003	0.15	0.14
41	-2.000	0.801	862.74	1552.1	457.1	84.6	0.804	1.003	1.001	-0.004	1.003	0.002	0.002	0.12	0.14
42	0.000	0.798	860.07	1554.2	455.9	84.5	0.802	1.004	1.000	-0.008	1.004	0.002	0.002	0.12	0.12
43	2.000	0.801	862.30	1552.9	457.1	84.5	0.803	1.003	1.000	-0.004	1.003	0.002	0.002	0.11	0.11
44	4.000	0.801	862.18	1551.3	456.7	84.3	0.805	1.005	1.001	-0.007	1.005	0.002	0.002	0.11	0.12
45	6.000	0.799	858.84	1552.4	455.3	84.7	0.802	1.004	1.001	-0.006	1.004	0.002	0.001	0.08	0.10
46	8.000	0.802	863.05	1550.6	457.1	84.3	0.803	1.001	1.001	-0.000	1.001	0.002	0.001	0.06	0.09
47	10.000	0.804	863.89	1549.7	457.9	83.5	0.807	1.004	1.001	-0.005	1.004	0.001	0.001	0.08	0.06
48	12.000	0.804	864.78	1546.9	457.3	84.2	0.808	1.005	1.001	-0.007	1.005	0.001	0.002	0.09	0.06
49	14.000	0.803	864.09	1550.0	457.5	86.6	0.809	1.007	1.000	-0.013	1.007	0.001	0.001	0.08	0.06
50	16.000	0.806	866.67	1546.3	458.5	84.1	0.810	1.004	1.001	-0.006	1.004	0.001	0.001	0.04	0.08
51	18.000	0.803	864.33	1550.1	457.6	84.7	0.804	1.001	1.000	-0.002	1.001	0.001	0.001	0.06	0.04
52	20.000	0.799	859.51	1551.5	455.0	83.9	0.804	1.006	1.001	-0.011	1.006	0.001	0.001	0.05	0.07
53	22.000	0.803	864.06	1549.6	457.8	83.8	0.806	1.002	1.001	-0.002	1.002	0.001	0.001	0.06	0.06
54	24.000	0.801	861.30	1547.5	455.5	83.2	0.804	1.003	1.001	-0.004	1.003	0.001	0.001	0.05	0.05

TEST	PART	REX10-6	ALFA	WING	XT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	33	2.995	****	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	YT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
23	14.000	0.796	853.29	1536.7	448.6	79.7	0.804	1.010	1.000	-0.019	1.010	0.001	0.005	0.30	0.08
24	12.000	0.798	855.93	1532.9	449.3	79.9	0.804	1.007	1.002	-0.010	1.007	-0.002	-0.005	-0.26	-0.09
25	10.000	0.803	850.32	1535.9	452.9	80.1	0.805	1.003	1.001	-0.003	1.003	-0.001	-0.001	-0.07	-0.08
26	8.000	0.797	854.92	1537.3	449.9	79.7	0.802	1.006	1.001	-0.009	1.006	-0.001	-0.002	-0.13	-0.04
27	6.000	0.799	856.19	1534.1	450.0	79.5	0.803	1.004	1.001	-0.005	1.004	0.002	-0.004	-0.21	0.14
28	4.000	0.800	858.04	1533.7	450.8	80.0	0.803	1.003	1.001	-0.004	1.003	0.002	-0.001	-0.06	0.10
29	2.000	0.796	856.43	1536.4	450.7	79.9	0.803	1.004	1.001	-0.007	1.004	0.003	0.002	0.11	0.17
30	0.000	0.800	858.09	1536.9	451.6	80.5	0.801	1.001	1.001	-0.000	1.001	0.002	0.005	0.29	0.12
31	-2.000	0.801	858.87	1538.7	452.6	80.7	0.805	1.005	1.000	-0.009	1.005	-0.000	0.001	0.05	-0.08
32	-4.000	0.799	857.87	1539.8	451.8	81.4	0.803	1.004	1.001	-0.006	1.004	0.000	0.000	0.00	0.02
33	-6.000	0.799	857.40	1540.3	451.9	80.9	0.804	1.005	1.001	-0.008	1.005	-0.001	0.005	0.31	-0.04
34	-8.000	0.799	857.45	1537.7	451.4	80.5	0.802	1.003	1.001	-0.003	1.003	0.000	0.003	0.20	0.01
35	-10.000	0.801	858.40	1536.1	451.8	80.2	0.804	1.004	1.001	-0.005	1.004	0.001	0.004	0.24	0.06
36	-12.000	0.800	856.24	1537.0	451.7	80.6	0.804	1.005	1.001	-0.008	1.005	0.002	0.002	0.14	0.13
37	-14.000	0.800	858.45	1538.5	452.1	81.0	0.804	1.004	1.000	-0.008	1.004	0.000	0.002	0.14	0.01

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	PEX10-6	ALPA	WING	XP	YT	RUN SURVEY	DATE
TC-484	33	2.996	0000	NONE	14.00	0.00	1- 1	2- 2-77

POINT	ZT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
7	14.000	0.799	854.97	1527.8	448.2	77.9	0.801	1.003	1.001	-0.003	1.003	0.001	0.005	0.20	0.05
8	12.000	0.799	855.16	1530.5	449.0	78.1	0.803	1.004	1.001	-0.007	1.004	0.003	0.003	0.18	0.18
9	10.000	0.798	855.02	1531.1	448.9	78.4	0.800	1.002	1.001	-0.001	1.002	0.006	-0.002	-0.11	0.33
10	8.000	0.799	855.35	1530.2	448.7	78.7	0.803	1.004	1.001	-0.006	1.004	0.006	-0.002	-0.11	0.37
11	6.000	0.798	855.01	1531.0	448.8	78.6	0.801	1.003	1.001	-0.005	1.003	0.006	-0.005	-0.27	0.33
12	4.000	0.801	857.54	1530.6	450.4	78.7	0.802	1.002	1.001	0.000	1.001	0.005	-0.003	-0.16	0.30
13	2.000	0.800	857.15	1531.7	450.4	78.8	0.802	1.002	1.001	-0.002	1.002	0.002	0.001	0.05	0.14
14	0.000	0.798	856.38	1531.8	449.5	78.3	0.803	1.004	1.001	-0.005	1.004	0.002	0.005	0.29	0.09
15	-2.000	0.802	858.81	1529.8	451.3	78.4	0.807	1.005	1.001	-0.008	1.005	0.003	0.006	0.33	0.20
16	-4.000	0.803	858.88	1528.7	451.0	78.4	0.805	1.003	1.001	-0.003	1.003	0.002	0.005	0.29	0.13
17	-6.000	0.801	857.84	1529.5	450.5	78.4	0.805	1.004	1.001	-0.005	1.004	-0.001	0.004	0.26	-0.04
18	-8.000	0.801	857.07	1530.2	450.6	78.7	0.805	1.004	1.001	-0.006	1.004	-0.002	0.004	0.21	-0.10
19	-10.000	0.799	855.27	1530.2	449.0	78.1	0.802	1.004	1.002	-0.004	1.004	-0.001	0.004	0.20	-0.04
20	-12.000	0.798	854.26	1528.9	448.7	79.3	0.802	1.004	1.001	-0.004	1.004	0.003	0.004	0.23	0.15
21	-14.000	0.798	855.67	1533.8	449.5	79.7	0.801	1.003	1.001	-0.005	1.003	0.003	0.004	0.21	0.14

TEST	PART	REX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	37	3.001	****	NONE	0.00	-14.14	1-701	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	PL	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
5	-6.000	0.856	917.81	1519.5	483.1	88.6	0.861	1.005	1.001	-0.009	1.005	0.003	0.007	0.42	0.19
6	-4.000	0.851	913.09	1523.9	481.4	88.4	0.860	1.009	1.001	-0.016	1.009	0.003	0.008	0.43	0.19
7	-2.000	0.846	909.27	1524.5	478.4	88.4	0.855	1.009	1.001	-0.017	1.009	0.004	0.007	0.43	0.20
8	0.000	0.848	909.45	1522.7	478.6	88.6	0.857	1.009	1.001	-0.017	1.009	0.003	0.007	0.40	0.19
9	2.000	0.842	904.52	1522.5	475.1	88.6	0.853	1.012	1.002	-0.020	1.012	0.003	0.006	0.36	0.18
10	4.000	0.840	902.25	1523.8	474.0	88.5	0.851	1.011	1.001	-0.020	1.011	0.003	0.005	0.31	0.18
11	6.000	0.837	900.33	1523.4	472.4	88.8	0.848	1.011	1.001	-0.019	1.011	0.003	0.005	0.29	0.18
12	8.000	0.838	900.26	1523.6	472.8	88.1	0.847	1.009	1.001	-0.016	1.009	0.003	0.004	0.24	0.16
13	10.000	0.841	902.59	1519.5	473.2	87.9	0.847	1.006	1.001	-0.010	1.006	0.003	0.003	0.20	0.18
14	12.000	0.846	907.42	1520.8	476.9	88.0	0.851	1.006	1.001	-0.009	1.006	0.003	0.004	0.22	0.16
15	14.000	0.859	919.46	1519.6	484.5	88.0	0.861	1.002	1.001	-0.002	1.002	0.002	0.003	0.20	0.13
16	16.000	0.883	947.79	1518.8	499.3	88.0	0.883	0.999	1.000	0.002	0.999	0.002	0.004	0.21	0.13
17	18.000	0.908	964.46	1517.8	513.4	87.3	0.904	0.996	1.001	0.010	0.996	0.002	0.034	0.24	0.12
18	20.000	0.928	982.87	1515.6	523.9	87.7	0.921	0.993	1.002	0.016	0.993	0.002	0.005	0.26	0.14
19	22.000	0.925	976.02	1523.4	522.1	88.0	0.919	0.999	1.001	0.004	0.999	0.003	0.005	0.28	0.15
20	24.000	0.832	896.15	1526.3	469.5	88.0	0.836	1.005	1.001	-0.006	1.005	0.002	0.006	0.32	0.10

TEST	PART	RNX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	37	2.982	****	NONE	0.00	0.00	1-702	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
22	-6.000	0.848	909.70	1519.2	477.8	88.1	0.860	1.012	1.001	-0.024	1.012	0.003	0.007	0.42	0.19
23	-4.000	0.849	910.55	1521.9	479.6	87.5	0.855	1.005	1.001	-0.009	1.005	0.003	0.007	0.39	0.19
24	-2.000	0.851	912.51	1519.8	480.1	87.8	0.857	1.006	1.001	-0.010	1.006	0.004	0.007	0.39	0.21
25	0.000	0.850	911.11	1521.1	479.7	87.6	0.856	1.007	1.001	-0.011	1.007	0.004	0.006	0.34	0.20
26	2.000	0.848	909.44	1518.3	477.8	87.4	0.856	1.008	1.002	-0.011	1.008	0.003	0.006	0.33	0.18
27	4.000	0.845	907.95	1521.9	477.1	88.4	0.853	1.008	1.001	-0.015	1.008	0.003	0.006	0.33	0.19
28	6.000	0.843	904.59	1524.0	475.8	88.1	0.850	1.008	1.001	-0.013	1.008	0.003	0.006	0.34	0.18
29	8.000	0.842	904.35	1522.7	475.4	88.0	0.849	1.007	1.002	-0.010	1.006	0.003	0.006	0.33	0.18
30	10.000	0.847	908.69	1520.2	477.5	88.0	0.852	1.005	1.001	-0.008	1.005	0.003	0.005	0.31	0.18
31	12.000	0.854	914.80	1519.5	481.4	88.0	0.858	1.004	1.001	-0.006	1.004	0.003	0.006	0.31	0.16
32	14.000	0.863	924.03	1520.2	487.7	88.1	0.867	1.004	1.001	-0.005	1.004	0.003	0.006	0.33	0.17
33	16.000	0.876	935.29	1517.3	494.6	87.4	0.879	1.003	1.002	-0.002	1.003	0.003	0.005	0.28	0.18
34	18.000	0.891	949.73	1514.8	502.4	88.0	0.893	1.002	1.002	-0.002	1.002	0.003	0.005	0.29	0.17
35	20.000	0.888	946.31	1517.5	501.4	88.0	0.895	1.007	1.000	-0.015	1.007	0.003	0.005	0.30	0.17
36	22.000	0.844	906.74	1527.2	477.7	88.6	0.848	1.004	1.001	-0.006	1.004	0.003	0.006	0.34	0.18
37	24.000	0.854	915.74	1520.0	482.0	88.3	0.858	1.004	1.002	-0.005	1.304	0.003	0.006	0.34	0.16

TEST	PART	RFX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	37	3.013	0000	NONE	14.14	0.00	1-703	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
39	-6.000	0.850	919.43	1524.2	485.9	88.7	0.855	0.997	0.999	0.005	0.997	0.004	0.007	0.40	0.21
40	-4.000	0.856	917.38	1523.0	483.8	89.0	0.861	1.006	1.001	-0.009	1.006	0.003	0.003	0.18	0.18
41	-2.000	0.853	915.38	1524.2	482.7	89.1	0.860	1.006	1.001	-0.011	1.006	0.003	0.003	0.15	0.16
42	0.000	0.849	911.14	1524.3	479.9	89.0	0.856	1.007	1.002	-0.012	1.007	0.003	0.003	0.14	0.14
43	2.000	0.843	905.34	1525.0	476.6	88.2	0.850	1.007	1.002	-0.011	1.007	0.002	0.003	0.15	0.11
44	4.000	0.840	902.43	1522.4	474.0	88.0	0.846	1.006	1.001	-0.010	1.006	0.002	0.002	0.13	0.12
45	6.000	0.838	901.55	1521.2	472.8	88.3	0.845	1.006	1.001	-0.010	1.006	0.002	0.002	0.14	0.12
46	8.000	0.840	902.46	1522.2	473.8	88.3	0.846	1.006	1.001	-0.009	1.006	0.002	0.002	0.12	0.10
47	10.000	0.844	906.28	1522.3	476.4	88.3	0.850	1.006	1.001	-0.009	1.005	0.002	0.002	0.10	0.11
48	12.000	0.850	911.73	1521.4	479.9	88.0	0.855	1.004	1.001	-0.007	1.004	0.002	0.002	0.12	0.13
49	14.000	0.859	920.00	1517.3	484.1	88.1	0.863	1.004	1.002	-0.003	1.004	0.002	0.002	0.12	0.11
50	16.000	0.870	930.20	1517.7	491.0	88.1	0.876	1.006	1.001	-0.010	1.006	0.002	0.001	0.08	0.12
51	18.000	0.840	910.64	1526.9	480.9	88.0	0.848	0.998	1.001	0.005	0.998	0.002	0.002	0.10	0.09
52	20.000	0.854	916.84	1520.0	483.0	87.9	0.858	1.003	1.002	-0.002	1.003	0.002	0.001	0.09	0.10
53	22.000	0.848	909.89	1521.3	476.7	87.7	0.855	1.007	1.002	-0.010	1.007	0.002	0.002	0.10	0.12
54	24.000	0.846	907.70	1518.0	476.4	87.6	0.855	1.009	1.002	-0.015	1.009	0.002	0.002	0.10	0.09

TEST	PART	REX10-6	ALFA	WING	XT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	36	2.989	****	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	YT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
22	14.000	0.845	906.13	1517.7	475.4	87.3	0.854	1.009	1.001	-0.017	1.009	0.002	0.006	0.32	0.09
24	12.000	0.851	912.30	1518.7	479.8	87.4	0.856	1.005	1.000	-0.011	1.005	-0.000	-0.002	-0.09	-0.01
25	10.000	0.851	910.87	1518.1	479.1	86.7	0.856	1.006	1.001	-0.010	1.006	-0.000	-0.000	-0.02	-0.02
26	8.000	0.853	913.24	1516.7	480.0	87.1	0.858	1.005	1.001	-0.008	1.005	-0.000	-0.001	-0.06	-0.01
27	6.000	0.848	909.43	1518.6	478.2	86.8	0.853	1.004	1.002	-0.006	1.004	0.003	-0.003	-0.14	0.17
28	4.000	0.851	911.79	1518.0	479.3	87.3	0.854	1.003	1.001	-0.004	1.003	0.002	-0.000	-0.02	0.12
29	2.000	0.849	910.12	1520.0	478.8	87.3	0.857	1.008	1.001	-0.015	1.008	0.003	0.003	0.16	0.10
30	0.000	0.851	912.27	1522.3	480.5	88.1	0.856	1.005	1.000	-0.009	1.005	0.003	0.006	0.34	0.16
31	-2.000	0.848	909.73	1521.1	478.5	88.0	0.854	1.006	1.001	-0.009	1.006	0.001	0.001	0.07	0.04
32	-4.000	0.851	912.01	1520.9	480.0	88.0	0.855	1.005	1.001	-0.008	1.005	0.001	0.001	0.05	0.07
33	-6.000	0.845	906.10	1522.6	476.8	87.5	0.854	1.010	1.002	-0.016	1.010	0.000	0.006	0.32	0.03
34	-8.000	0.850	910.51	1518.9	478.7	87.3	0.856	1.007	1.001	-0.012	1.007	0.001	0.005	0.26	0.06
35	-10.000	0.851	911.46	1516.8	478.8	87.2	0.856	1.005	1.001	-0.008	1.005	0.002	0.004	0.25	0.12
36	-12.000	0.849	910.17	1519.2	478.4	87.6	0.856	1.008	1.001	-0.014	1.008	0.003	0.003	0.16	0.15
37	-14.000	0.849	910.45	1519.4	478.5	87.9	0.855	1.007	1.001	-0.011	1.007	0.001	0.003	0.16	0.06

TEST	PART	REX10-6	ALFA	WING	XT	YT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	36.	7.998	****	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	ZT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
6	14.000	0.848	909.67	1518.6	478.1	87.2	0.854	1.005	1.001	-0.007	1.005	0.002	0.005	0.31	0.09
7	12.000	0.851	911.55	1516.9	479.0	87.0	0.855	1.004	1.001	-0.007	1.004	0.004	0.003	0.18	0.21
8	10.000	0.850	910.51	1518.8	479.0	86.8	0.855	1.005	1.001	-0.010	1.005	0.006	-0.001	-0.08	0.37
9	8.000	0.847	907.86	1517.3	476.7	86.9	0.854	1.007	1.002	-0.011	1.007	0.007	-0.001	-0.08	0.38
10	6.000	0.850	910.56	1516.5	478.3	86.8	0.854	1.004	1.001	-0.005	1.004	0.006	-0.003	-0.19	0.34
11	4.000	0.848	908.08	1518.7	477.4	86.6	0.855	1.008	1.000	-0.015	1.008	0.006	-0.003	-0.15	0.35
12	2.000	0.851	911.12	1516.0	478.4	86.9	0.853	1.003	1.001	-0.003	1.003	0.003	0.002	0.09	0.17
13	0.000	0.853	913.26	1516.2	479.9	86.9	0.857	1.004	1.001	-0.007	1.004	0.003	0.005	0.30	0.16
14	-2.000	0.851	911.70	1517.6	479.0	87.0	0.857	1.007	1.001	-0.011	1.007	0.004	0.006	0.34	0.22
15	-4.000	0.851	911.64	1518.8	479.6	87.0	0.857	1.006	1.001	-0.010	1.006	0.003	0.005	0.31	0.17
16	-6.000	0.850	910.02	1518.6	478.6	86.7	0.855	1.005	1.001	-0.009	1.005	0.001	0.005	0.28	0.04
17	-8.000	0.850	910.95	1517.0	478.8	86.9	0.855	1.004	1.001	-0.007	1.004	-0.001	0.005	0.27	-0.05
18	-10.000	0.850	910.73	1516.6	478.5	86.7	0.856	1.006	1.001	-0.011	1.006	0.001	0.005	0.27	0.05
19	-12.000	0.851	910.85	1517.6	478.9	86.7	0.855	1.004	1.001	-0.008	1.004	0.003	0.005	0.27	0.17
20	-14.000	0.848	908.57	1517.5	477.3	86.8	0.852	1.005	1.001	-0.008	1.005	0.003	0.005	0.26	0.14

TEST	PART	REX10-6	ALPA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	30	2.996	0000	NONE	0.00	-14.14	1-701	2- 2-77	TRANSONIC 4T

POINT	XP	M	V	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BNTL
5	-6.000	0.904	963.43	1496.5	501.8	90.5	0.906	1.000	1.000	-0.000	1.000	0.003	0.007	0.39	0.17
6	-4.000	0.899	959.59	1495.5	500.6	90.4	0.905	1.006	1.001	-0.011	1.006	0.003	0.007	0.39	0.17
7	-2.000	0.894	954.13	1495.5	497.7	90.6	0.899	1.005	1.001	-0.008	1.005	0.003	0.006	0.36	0.17
8	0.000	0.885	947.44	1495.6	493.6	90.4	0.893	1.008	1.000	-0.012	1.008	0.003	0.006	0.33	0.19
9	2.000	0.891	951.72	1492.8	495.1	90.4	0.898	1.008	1.002	-0.011	1.007	0.003	0.005	0.31	0.17
10	4.000	0.896	956.10	1493.3	498.5	90.0	0.900	1.003	1.002	-0.004	1.003	0.003	0.005	0.27	0.19
11	6.000	0.893	952.88	1492.9	496.4	89.9	0.898	1.005	1.001	-0.008	1.005	0.003	0.004	0.23	0.19
12	8.000	0.904	963.25	1491.8	502.4	90.0	0.907	1.003	1.001	-0.003	1.003	0.003	0.003	0.18	0.19
13	10.000	0.887	947.25	1495.0	493.6	89.8	0.892	1.005	1.001	-0.009	1.005	0.003	0.003	0.16	0.17
14	12.000	0.868	929.84	1493.7	481.9	90.1	0.874	1.007	1.001	-0.012	1.007	0.003	0.004	0.21	0.15
15	14.000	0.883	943.49	1488.9	489.6	89.7	0.881	0.998	1.002	0.008	0.998	0.002	0.002	0.14	0.14
16	16.000	0.912	968.78	1488.1	505.0	89.8	0.909	0.998	1.001	0.005	0.998	0.003	0.004	0.21	0.15
17	18.000	0.866	929.64	1497.5	492.0	90.7	0.872	1.006	1.001	-0.009	1.006	0.002	0.004	0.25	0.12
18	20.000	0.854	916.44	1494.9	473.8	89.9	0.852	0.999	1.001	0.005	0.999	0.002	0.005	0.27	0.13
19	22.000	0.874	935.80	1493.9	485.8	90.1	0.870	0.996	1.000	0.009	0.996	0.002	0.005	0.28	0.10
20	24.000	0.883	943.81	1496.4	491.6	90.3	0.870	0.988	1.001	0.026	0.988	0.003	0.004	0.26	0.15

TEST	PART	REX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	39	2.997	****	NONE	0.00	0.00	1-702	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
22	-6.000	0.845	909.14	1494.9	449.5	90.9	0.875	1.031	1.000	-0.062	1.031	0.003	0.005	0.30	0.17
23	-4.000	0.866	927.51	1485.4	470.1	89.3	0.897	1.021	1.001	-0.041	1.021	0.003	0.006	0.33	0.19
24	-2.000	0.853	915.46	1493.1	472.7	89.4	0.877	1.025	1.001	-0.048	1.025	0.004	0.006	0.34	0.20
25	0.000	0.834	897.44	1491.1	459.9	89.8	0.860	1.027	1.001	-0.053	1.027	0.003	0.006	0.34	0.18
26	2.000	0.866	928.46	1511.5	486.8	90.0	0.893	1.026	1.001	-0.050	1.026	0.003	0.006	0.35	0.18
27	4.000	0.868	930.01	1492.5	481.9	89.6	0.890	1.021	1.000	-0.042	1.021	0.003	0.005	0.29	0.19
28	6.000	0.841	904.43	1500.1	467.6	89.4	0.859	1.018	1.001	-0.035	1.018	0.003	0.005	0.28	0.16
29	8.000	0.864	925.46	1487.5	477.6	89.3	0.882	1.018	1.001	-0.034	1.018	0.003	0.005	0.29	0.18
30	10.000	0.848	910.53	1493.0	469.9	88.6	0.864	1.016	1.002	-0.028	1.016	0.003	0.005	0.29	0.18
31	12.000	0.830	892.34	1488.2	456.6	88.0	0.829	0.999	1.001	0.004	0.999	0.002	0.004	0.26	0.13
32	14.000	0.858	919.22	1484.5	472.9	88.5	0.860	1.002	1.000	-0.004	1.002	0.002	0.005	0.28	0.14
33	16.000	0.902	964.31	1510.2	507.3	93.6	0.903	1.001	1.001	0.000	1.001	0.003	0.005	0.31	0.10
34	18.000	0.914	975.38	1501.3	511.3	93.4	0.913	0.999	1.002	0.005	0.999	0.003	0.005	0.29	0.16
35	20.000	0.917	978.21	1504.8	514.2	93.4	0.916	0.999	1.002	0.006	0.999	0.003	0.005	0.30	0.10
36	22.000	0.930	990.26	1502.7	520.8	93.3	0.923	0.993	1.003	0.018	0.993	0.002	0.006	0.33	0.14
37	24.000	0.937	995.02	1498.8	522.7	92.5	0.928	0.992	1.002	0.019	0.992	0.003	0.006	0.36	0.18

TEST	PART	PFX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	39	2,969	oooo	NONE	14,14	0,00	1-703	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
39	-6.000	0.862	977.44	1515.2	485.5	93.3	0.875	1.012	0.994	-0.036	1.012	0.004	0.008	0.45	0.20
40	-4.000	0.903	964.34	1507.6	507.2	92.2	0.904	1.001	1.002	0.001	1.001	0.002	0.003	0.15	0.12
41	-2.000	0.897	958.90	1506.3	503.4	92.2	0.901	1.003	1.001	-0.004	1.003	0.002	0.002	0.13	0.13
42	0.000	0.891	953.00	1502.2	498.5	91.9	0.894	1.003	1.002	-0.004	1.003	0.002	0.002	0.12	0.11
43	2.000	0.893	954.64	1499.1	498.9	91.1	0.895	1.002	1.002	0.000	1.002	0.002	0.002	0.10	0.11
44	4.000	0.895	955.07	1497.3	499.0	90.2	0.897	1.002	1.001	-0.002	1.002	0.002	0.002	0.11	0.12
45	6.000	0.898	957.26	1492.9	499.3	89.6	0.900	1.002	1.001	-0.002	1.002	0.002	0.002	0.13	0.10
46	8.000	0.900	958.57	1491.4	499.4	90.0	0.901	1.002	1.001	-0.002	1.002	0.002	0.002	0.12	0.09
47	10.000	0.902	960.88	1493.6	502.0	89.2	0.904	1.001	1.001	-0.001	1.001	0.002	0.002	0.10	0.10
48	12.000	0.907	966.12	1487.0	507.7	89.8	0.907	1.000	1.002	0.003	1.000	0.002	0.002	0.11	0.10
49	14.000	0.915	973.10	1492.5	508.7	90.0	0.914	0.999	1.000	0.002	0.999	0.002	0.002	0.11	0.11
50	16.000	0.922	978.94	1492.6	512.4	89.7	0.917	0.995	1.001	0.011	0.995	0.001	0.002	0.13	0.08
51	18.000	0.925	981.41	1490.1	513.5	89.1	0.923	0.998	1.000	0.004	0.998	0.002	0.002	0.12	0.10
52	20.000	0.929	985.91	1490.3	515.7	89.8	0.924	0.995	1.002	0.013	0.995	0.002	0.002	0.13	0.11
53	22.000	0.926	982.92	1493.5	515.3	89.3	0.923	0.997	1.001	0.008	0.997	0.002	0.002	0.14	0.12
54	24.000	0.901	957.77	1488.1	499.3	87.3	0.903	1.002	1.001	-0.001	1.002	0.002	0.002	0.10	0.10

TEST	PART	REX10-6	ALFA	WING	XT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	38	2.997	****	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	YT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BNTL
21	14.000	0.898	957.83	1492.6	499.3	90.0	0.901	1.003	1.000	-0.005	1.003	0.002	0.005	0.28	0.09
23	12.000	0.897	957.16	1489.6	497.9	90.0	0.901	1.003	1.004	0.001	1.003	0.001	-0.001	-0.06	0.05
24	10.000	0.898	957.64	1492.6	499.2	90.0	0.900	1.002	1.002	-0.001	1.002	-0.000	-0.000	-0.01	-0.01
25	8.000	0.899	958.92	1492.7	499.9	90.3	0.901	1.002	1.001	-0.002	1.002	0.000	-0.001	-0.08	0.02
26	6.000	0.899	958.68	1493.0	500.2	89.6	0.901	1.002	1.001	-0.001	1.002	0.004	-0.002	-0.14	0.21
27	4.000	0.900	958.87	1490.9	499.9	89.1	0.901	1.001	1.001	-0.001	1.001	0.003	-0.001	-0.04	0.14
28	2.000	0.900	958.95	1488.0	499.1	88.9	0.902	1.002	1.001	-0.001	1.001	0.004	0.002	0.13	0.22
29	0.000	0.901	960.23	1491.3	500.7	89.4	0.904	1.002	1.001	-0.003	1.002	0.003	0.005	0.30	0.18
30	-2.000	0.899	958.17	1489.1	498.9	89.0	0.902	1.003	1.002	-0.002	1.003	0.001	0.001	0.07	0.05
31	-4.000	0.900	958.75	1488.9	499.1	89.3	0.902	1.002	1.001	-0.003	1.002	0.001	0.001	0.03	0.08
32	-6.000	0.896	958.28	1489.8	498.9	89.6	0.902	1.003	1.002	-0.003	1.003	0.001	0.005	0.28	0.04
33	-8.000	0.898	956.93	1489.8	498.1	89.5	0.901	1.003	1.002	-0.003	1.003	0.002	0.004	0.25	0.11
34	-10.000	0.897	956.72	1488.5	497.5	89.1	0.900	1.003	1.002	-0.002	1.003	0.002	0.004	0.26	0.13
35	-12.000	0.897	956.08	1489.1	497.6	89.1	0.900	1.002	1.001	-0.002	1.002	0.003	0.003	0.17	0.15
36	-14.000	0.897	956.48	1489.5	497.8	89.3	0.901	1.003	1.001	-0.005	1.003	0.001	0.003	0.17	0.07

TEST	PART	REX10-6	ALFA	WING	XT	YT	RUN	SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-404	38	3.905	****	NONE	14.00	0.00	1-	1	2- 2-77	TRANSONIC 4T

POINT	ZT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
5	14.000	0.899	957.34	1497.2	499.6	88.8	0.902	1.003	1.001	-0.005	1.003	0.002	0.005	0.27	0.13
6	12.000	0.897	955.95	1489.8	497.0	88.7	0.899	1.002	1.002	-0.001	1.002	0.004	0.002	0.13	0.23
7	10.000	0.897	955.96	1485.8	496.7	88.6	0.900	1.002	1.002	-0.001	1.002	0.007	-0.001	-0.09	0.38
8	8.000	0.900	959.94	1488.3	498.7	89.8	0.902	1.002	1.000	-0.004	1.002	0.007	-0.001	-0.09	0.38
9	6.000	0.900	959.14	1495.5	501.2	89.8	0.903	1.003	1.001	-0.005	1.003	0.006	-0.004	-0.21	0.34
10	4.000	0.898	957.40	1496.1	500.6	89.4	0.901	1.003	1.002	-0.003	1.003	0.006	-0.002	-0.14	0.36
11	2.000	0.897	955.56	1498.7	497.2	89.0	0.899	1.002	1.002	-0.001	1.002	0.003	0.001	0.06	0.19
12	0.000	0.897	956.27	1488.2	497.1	89.6	0.900	1.003	1.001	-0.003	1.003	0.003	0.005	0.20	0.18
13	-2.000	0.897	957.18	1492.0	498.7	90.0	0.901	1.003	1.001	-0.005	1.003	0.004	0.005	0.31	0.22
14	-4.000	0.897	956.37	1495.2	499.5	89.6	0.900	1.003	1.001	-0.004	1.003	0.003	0.004	0.26	0.17
15	-6.000	0.896	955.19	1493.2	497.9	89.9	0.898	1.003	1.002	-0.002	1.003	0.001	0.004	0.25	0.05
16	-8.000	0.895	955.10	1488.9	496.5	90.0	0.897	1.002	1.001	-0.001	1.002	-0.000	0.004	0.25	-0.02
17	-10.000	0.895	955.36	1493.2	498.0	90.0	0.899	1.003	1.001	-0.005	1.003	0.001	0.004	0.23	0.05
18	-12.000	0.896	954.76	1495.0	498.2	90.0	0.897	1.002	1.001	-0.004	1.002	0.003	0.004	0.23	0.19
19	-14.000	0.894	954.01	1493.8	497.6	89.6	0.896	1.002	1.001	-0.001	1.002	0.003	0.004	0.23	0.18

TEST	PART	REX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	41	2.971	****	NONE	0.00	-14.14	1-701	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
5	-6.000	0.925	978.51	1450.0	498.5	86.0	0.924	1.000	1.000	0.001	1.000	0.003	0.007	0.40	0.16
6	-4.000	0.905	960.63	1456.0	490.3	85.8	0.911	1.005	1.001	-0.008	1.005	0.003	0.007	0.41	0.17
7	-2.000	0.895	951.56	1448.9	483.1	85.9	0.901	1.005	1.002	-0.007	1.005	0.003	0.007	0.38	0.17
8	0.000	0.934	986.44	1447.6	503.5	85.7	0.926	0.993	1.001	0.016	0.993	0.003	0.007	0.39	0.16
9	2.000	0.943	994.65	1452.6	509.9	85.7	0.934	0.992	1.001	0.017	0.992	0.003	0.006	0.36	0.17
10	4.000	0.945	996.66	1450.8	510.2	86.1	0.936	0.992	1.002	0.019	0.992	0.003	0.006	0.33	0.19
11	6.000	0.946	997.99	1450.9	510.9	86.1	0.937	0.992	1.001	0.018	0.992	0.003	0.005	0.29	0.17
12	8.000	0.947	998.23	1451.3	511.3	85.9	0.937	0.992	1.002	0.019	0.992	0.003	0.005	0.27	0.13
13	10.000	0.951	1002.28	1446.6	512.0	85.8	0.937	0.987	1.002	0.029	0.987	0.002	0.005	0.27	0.13
14	12.000	0.958	1008.23	1450.8	516.9	85.7	0.941	0.984	1.000	0.031	0.984	0.002	0.005	0.29	0.12
15	14.000	0.964	1013.69	1448.6	519.1	85.7	0.943	0.981	1.001	0.040	0.981	0.002	0.005	0.30	0.12
16	16.000	0.966	1014.93	1448.2	519.6	85.8	0.944	0.981	1.001	0.039	0.981	0.002	0.005	0.31	0.13
17	18.000	0.960	1010.67	1452.3	518.3	86.5	0.944	0.985	1.001	0.031	0.985	0.003	0.006	0.32	0.15
18	20.000	0.965	1015.28	1452.3	521.0	86.2	0.942	0.980	1.001	0.043	0.980	0.002	0.006	0.33	0.11
19	22.000	0.948	998.85	1451.0	511.8	85.5	0.935	0.988	1.001	0.025	0.988	0.002	0.005	0.31	0.12
20	24.000	0.948	998.66	1447.4	510.8	84.8	0.936	0.989	1.002	0.026	0.989	0.002	0.006	0.33	0.11

TEST	PART	RNX10-6	ALPA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	41	3.001	****	NONE	0.00	0.00	1-702	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
22	-6.000	0.946	997.52	1450.7	511.0	85.4	0.934	0.988	1.002	0.026	0.988	0.003	0.007	0.40	0.19
23	-4.000	0.951	1001.59	1446.7	512.0	85.2	0.938	0.988	1.001	0.026	0.988	0.003	0.007	0.38	0.17
24	-2.000	0.948	999.20	1447.9	510.9	85.4	0.937	0.990	1.001	0.022	0.990	0.003	0.007	0.38	0.17
25	0.000	0.945	995.68	1449.3	509.5	85.2	0.936	0.992	1.001	0.018	0.992	0.003	0.006	0.37	0.17
26	2.000	0.945	996.44	1447.0	509.0	85.6	0.935	0.991	1.002	0.020	0.991	0.003	0.006	0.35	0.18
27	4.000	0.947	997.34	1449.8	510.7	85.1	0.937	0.991	1.001	0.019	0.991	0.003	0.006	0.35	0.16
28	6.000	0.949	999.50	1449.1	511.6	85.2	0.937	0.990	1.001	0.023	0.990	0.003	0.006	0.33	0.15
29	8.000	0.950	1000.37	1446.0	511.0	85.2	0.937	0.989	1.002	0.025	0.989	0.003	0.006	0.32	0.15
30	10.000	0.956	1006.64	1449.7	515.4	86.0	0.941	0.986	1.000	0.027	0.986	0.003	0.006	0.34	0.15
31	12.000	0.953	1003.51	1453.2	515.2	85.4	0.940	0.988	1.001	0.026	0.988	0.003	0.006	0.34	0.18
32	14.000	0.955	1005.31	1447.6	514.2	85.4	0.940	0.987	1.002	0.030	0.987	0.003	0.006	0.34	0.15
33	16.000	0.957	1006.38	1446.3	514.6	85.0	0.940	0.985	1.001	0.032	0.985	0.002	0.006	0.32	0.12
34	18.000	0.961	1011.14	1449.9	518.0	86.0	0.943	0.984	1.001	0.033	0.984	0.002	0.006	0.35	0.13
35	20.000	0.957	1007.20	1454.3	517.4	85.9	0.942	0.986	1.001	0.028	0.986	0.003	0.006	0.37	0.15
36	22.000	0.958	1008.06	1447.1	515.5	85.5	0.941	0.985	1.003	0.036	0.985	0.002	0.006	0.35	0.12
37	24.000	0.964	1012.72	1445.6	517.6	85.4	0.942	0.981	1.001	0.039	0.981	0.002	0.006	0.34	0.14

TEST	PART	REX10-6	ALPA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	41	2.997	****	NONE	14.14	0.00	1-703	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BATL
39	-6.000	0.945	996.58	1449.4	510.0	85.4	0.907	0.966	1.004	0.075	0.965	0.003	0.007	0.39	0.20
40	-4.000	0.945	995.68	1448.1	509.2	85.1	0.934	0.990	1.001	0.021	0.990	0.002	0.003	0.19	0.13
41	-2.000	0.947	997.44	1447.6	509.9	85.4	0.935	0.990	1.001	0.022	0.990	0.002	0.003	0.16	0.11
42	0.000	0.945	996.17	1450.3	510.1	85.2	0.934	0.990	1.001	0.021	0.990	0.002	0.003	0.17	0.11
43	2.000	0.945	996.06	1446.9	508.7	85.6	0.934	0.991	1.001	0.021	0.991	0.002	0.003	0.15	0.11
44	4.000	0.949	1000.59	1449.8	511.7	86.0	0.937	0.989	1.001	0.023	0.989	0.002	0.003	0.15	0.11
45	6.000	0.950	1001.41	1452.0	513.3	86.3	0.938	0.990	1.002	0.024	0.990	0.002	0.003	0.16	0.11
46	8.000	0.950	1001.07	1448.3	512.0	85.6	0.937	0.988	1.002	0.027	0.988	0.002	0.002	0.14	0.10
47	10.000	0.954	1004.62	1446.9	513.5	85.6	0.939	0.987	1.001	0.028	0.987	0.001	0.002	0.13	0.09
48	12.000	0.953	1002.97	1448.7	513.3	85.5	0.939	0.988	1.001	0.026	0.988	0.002	0.002	0.13	0.10
49	14.000	0.954	1003.95	1449.4	514.1	85.4	0.941	0.988	1.002	0.026	0.988	0.002	0.002	0.14	0.11
50	16.000	0.959	1008.93	1448.3	516.5	85.4	0.941	0.984	1.001	0.033	0.984	0.002	0.003	0.18	0.10
51	18.000	0.961	1010.34	1449.3	517.5	85.7	0.943	0.985	1.001	0.033	0.985	0.002	0.002	0.13	0.09
52	20.000	0.956	1005.98	1450.8	515.6	85.5	0.938	0.984	1.001	0.034	0.984	0.002	0.002	0.14	0.10
53	22.000	0.965	1014.41	1446.9	518.9	85.6	0.944	0.981	1.002	0.041	0.981	0.002	0.002	0.14	0.10
54	24.000	0.969	1017.84	1447.6	521.1	85.4	0.961	0.992	1.001	0.017	0.992	0.002	0.002	0.14	0.12

TEST	PART	REX10-6	ALPA	WING	XT	ZT	RUN	SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	40	2.081	****	NONE	14.00	0.00	1-	1	2- 2-77	TRANSONIC 4T

POINT	YT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
23	14.000	0.949	1002.16	1455.5	513.7	88.5	0.939	0.991	1.003	0.072	0.991	0.002	0.005	0.30	0.11
25	12.000	0.949	1001.75	1456.6	514.3	87.6	0.937	0.990	1.004	0.028	0.990	0.001	-0.001	-0.06	0.05
26	10.000	0.950	1002.79	1451.9	513.4	87.3	0.938	0.989	1.002	0.026	0.989	0.000	0.000	0.02	0.00
27	8.000	0.954	1005.90	1458.4	517.4	88.4	0.940	0.988	1.001	0.026	0.988	0.001	-0.001	-0.05	0.04
28	6.000	0.955	1008.01	1455.8	517.1	88.4	0.940	0.986	1.002	0.030	0.986	0.003	-0.001	-0.08	0.18
29	4.000	0.955	1007.40	1456.6	517.3	87.9	0.939	0.986	1.001	0.029	0.986	0.003	0.000	0.02	0.15
30	2.000	0.954	1006.66	1454.7	516.3	87.7	0.939	0.987	1.002	0.029	0.987	0.003	0.003	0.16	0.20
31	0.000	0.952	1004.41	1453.8	516.8	87.6	0.938	0.987	1.001	0.027	0.987	0.003	0.006	0.32	0.17
32	-2.000	0.953	1005.28	1452.4	514.6	87.9	0.938	0.987	1.001	0.027	0.987	0.001	0.002	0.13	0.07
33	-4.000	0.955	1006.81	1453.9	516.7	87.6	0.940	0.987	1.001	0.026	0.987	0.001	0.001	0.08	0.07
34	-6.000	0.950	1002.90	1456.2	514.9	87.5	0.938	0.989	1.002	0.025	0.989	0.001	0.006	0.33	0.06
35	-8.000	0.949	1001.60	1451.3	517.5	87.7	0.936	0.988	1.002	0.026	0.988	0.002	0.005	0.27	0.09
36	-10.000	0.950	1002.14	1455.5	514.3	87.2	0.938	0.990	1.001	0.072	0.990	0.002	0.005	0.31	0.13
37	-12.000	0.949	1000.79	1454.1	513.7	87.0	0.936	0.989	1.001	0.024	0.989	0.003	0.004	0.21	0.17
38	-14.000	0.948	1000.08	1453.1	512.7	86.4	0.937	0.990	1.001	0.022	0.990	0.001	0.004	0.21	0.08

TEST	PART	RNX10-6	ALFA	WING	XT	YT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	40	2.991	0000	NONE	14.00	0.00	1- 1	3- 2-77	TRANSONIC 4T

POINT	ZT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BNTL
7	14.000	0.948	1001.17	1453.8	512.8	87.9	0.936	0.989	1.002	0.024	0.989	0.002	0.005	0.29	0.11
8	12.000	0.950	1003.13	1457.2	515.0	88.7	0.938	0.989	1.001	0.023	0.989	0.004	0.003	0.19	0.22
9	10.000	0.951	1004.06	1460.7	516.5	88.6	0.938	0.989	1.001	0.024	0.989	0.006	-0.001	-0.07	0.36
10	8.000	0.949	1002.88	1458.4	515.1	88.6	0.937	0.989	1.002	0.026	0.989	0.007	-0.001	-0.03	0.38
11	6.000	0.949	1002.20	1458.2	514.9	88.3	0.937	0.990	1.002	0.024	0.990	0.006	-0.003	-0.16	0.35
12	4.000	0.948	1000.88	1456.5	513.9	87.4	0.937	0.990	1.002	0.024	0.990	0.006	-0.002	-0.12	0.35
13	2.000	0.949	1001.72	1456.6	514.1	87.9	0.937	0.989	1.001	0.023	0.989	0.003	0.002	0.14	0.19
14	0.000	0.949	1001.89	1459.7	514.8	88.1	0.937	0.990	1.001	0.023	0.989	0.003	0.005	0.30	0.16
15	-2.000	0.948	1001.72	1459.6	514.5	88.5	0.936	0.989	1.001	0.023	0.989	0.003	0.006	0.34	0.20
16	-4.000	0.948	1002.13	1461.3	515.7	88.5	0.937	0.989	1.001	0.023	0.989	0.003	0.006	0.32	0.19
17	-6.000	0.947	1000.61	1458.2	513.8	88.5	0.935	0.989	1.002	0.025	0.989	0.001	0.005	0.30	0.05
18	-8.000	0.947	1001.22	1458.6	514.2	89.5	0.936	0.989	1.001	0.023	0.989	-0.000	0.005	0.30	-0.02
19	-10.000	0.947	1000.79	1460.3	514.7	88.4	0.935	0.989	1.001	0.023	0.989	0.001	0.005	0.29	0.03
20	-12.000	0.946	999.79	1458.6	513.7	88.0	0.934	0.989	1.001	0.024	0.989	0.003	0.005	0.29	0.15
21	-14.000	0.946	998.97	1455.5	512.3	87.7	0.933	0.989	1.002	0.026	0.989	0.003	0.005	0.30	0.18

TEST	PART	REX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	43	3.002	****	NONE	0.00	-14.14	1-701	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
12	-6.000	0.998	1042.91	1426.5	526.8	84.9	0.980	0.985	1.001	0.032	0.985	0.003	0.007	0.39	0.17
13	-4.000	0.998	1043.34	1424.6	525.8	85.5	0.980	0.985	1.001	0.032	0.985	0.003	0.007	0.40	0.17
14	-2.000	0.998	1042.54	1427.5	526.9	85.1	0.984	0.988	1.001	0.025	0.988	0.003	0.007	0.41	0.18
15	0.000	0.994	1039.45	1424.1	524.1	85.0	0.979	0.987	1.002	0.029	0.987	0.003	0.007	0.40	0.18
16	2.000	0.995	1041.63	1423.2	524.7	85.4	0.983	0.989	1.001	0.025	0.988	0.003	0.006	0.38	0.18
17	4.000	0.997	1042.44	1427.9	526.8	85.5	0.986	0.990	1.001	0.020	0.990	0.003	0.006	0.35	0.19
18	6.000	0.996	1042.42	1426.8	526.1	86.2	0.983	0.988	1.002	0.026	0.988	0.003	0.006	0.33	0.16
19	8.000	0.999	1044.79	1429.6	528.2	86.4	0.983	0.987	1.001	0.028	0.986	0.003	0.006	0.32	0.17
20	10.000	1.000	1045.22	1431.3	529.4	85.6	0.984	0.987	1.002	0.030	0.986	0.003	0.006	0.33	0.18
21	12.000	0.998	1047.90	1423.0	525.3	85.4	0.978	0.983	1.002	0.037	0.983	0.003	0.006	0.32	0.18
22	14.000	1.004	1048.41	1426.1	529.3	85.3	0.986	0.985	1.000	0.031	0.985	0.003	0.005	0.31	0.18
23	16.000	1.008	1051.35	1426.8	531.1	85.1	0.982	0.978	1.002	0.046	0.978	0.003	0.006	0.33	0.16
24	18.000	1.005	1051.90	1424.3	530.2	85.5	0.983	0.979	1.001	0.043	0.979	0.002	0.006	0.35	0.14
25	20.000	1.010	1053.14	1427.2	532.1	85.1	0.986	0.980	1.001	0.041	0.980	0.003	0.006	0.35	0.15
26	22.000	1.006	1049.73	1425.4	529.7	85.2	0.984	0.982	1.002	0.039	0.982	0.003	0.008	0.37	0.15
27	24.000	1.003	1047.21	1424.2	527.9	85.4	0.982	0.983	1.001	0.036	0.983	0.002	0.006	0.35	0.14

TEST	PART	RNX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	43	2.999	****	NONE	0.00	0.00	1-702	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	C	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BATL
29	-6.000	1.000	1045.01	1428.3	528.0	86.0	0.979	0.983	1.002	0.037	0.983	0.003	0.007	0.40	0.19
30	-4.000	0.998	1043.89	1430.3	528.1	86.3	0.983	0.988	1.001	0.027	0.988	0.003	0.007	0.39	0.20
31	-2.000	0.998	1043.46	1430.4	528.0	86.1	0.981	0.986	1.001	0.030	0.986	0.003	0.006	0.37	0.17
32	0.000	0.997	1042.93	1428.1	526.9	86.1	0.982	0.987	1.001	0.027	0.987	0.003	0.007	0.39	0.18
33	2.000	0.998	1043.47	1429.1	527.6	86.0	0.983	0.988	1.002	0.027	0.988	0.003	0.006	0.37	0.20
34	4.000	0.998	1043.41	1427.5	527.0	85.8	0.981	0.986	1.002	0.031	0.986	0.003	0.006	0.35	0.17
35	6.000	0.998	1042.65	1425.5	525.1	85.4	0.980	0.985	1.002	0.032	0.985	0.003	0.006	0.35	0.17
36	8.000	0.999	1043.36	1424.4	526.1	85.3	0.982	0.986	1.002	0.030	0.986	0.003	0.006	0.33	0.18
37	10.000	1.000	1044.82	1425.1	527.0	85.5	0.982	0.985	1.002	0.033	0.985	0.003	0.006	0.34	0.19
38	12.000	1.004	1048.58	1428.7	530.0	86.0	0.983	0.983	1.001	0.036	0.983	0.003	0.006	0.35	0.17
39	14.000	1.004	1048.75	1429.9	530.4	86.2	0.983	0.982	1.001	0.037	0.982	0.003	0.006	0.36	0.17
40	16.000	1.007	1051.84	1429.7	531.7	86.6	0.981	0.979	1.001	0.044	0.979	0.003	0.006	0.35	0.18
41	18.000	1.005	1049.65	1428.7	530.5	86.1	0.982	0.981	1.002	0.041	0.981	0.003	0.006	0.38	0.17
42	20.000	1.005	1049.72	1426.4	529.7	86.1	0.982	0.981	1.002	0.041	0.981	0.003	0.007	0.38	0.17
43	22.000	1.004	1049.16	1429.5	530.4	86.4	0.984	0.984	1.001	0.035	0.984	0.003	0.006	0.37	0.16
44	24.000	1.001	1046.86	1429.9	529.4	86.3	0.982	0.984	1.002	0.035	0.984	0.002	0.006	0.37	0.14

TEST	PART	RE X 10-6	ALFA	WING	YT	ZT	RUN	SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	43	3.000	oooo	NONE	14.14	0.00	1-703		2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
46	-6.000	0.999	1043.80	1429.6	527.9	86.0	0.977	0.982	0.998	0.034	0.982	0.003	0.007	0.39	0.19
47	-4.000	0.998	1044.71	1431.5	529.0	86.2	0.982	0.986	1.001	0.030	0.986	0.003	0.003	0.19	0.17
48	-2.000	0.997	1042.87	1428.7	526.9	86.0	0.981	0.986	1.002	0.030	0.986	0.003	0.003	0.19	0.18
49	0.000	0.999	1044.16	1428.4	527.6	86.0	0.981	0.985	1.001	0.030	0.985	0.003	0.003	0.18	0.15
50	2.000	0.999	1044.60	1430.7	528.3	86.5	0.985	0.988	1.001	0.024	0.988	0.003	0.003	0.19	0.15
51	4.000	0.997	1042.94	1429.1	527.1	86.3	0.983	0.988	1.002	0.026	0.988	0.002	0.003	0.17	0.14
52	6.000	0.998	1043.45	1428.0	527.1	86.0	0.980	0.985	1.001	0.032	0.985	0.003	0.003	0.18	0.15
53	8.000	1.000	1045.53	1428.4	528.1	86.4	0.983	0.986	1.002	0.031	0.986	0.002	0.002	0.14	0.13
54	10.000	1.000	1045.65	1429.0	528.5	86.3	0.982	0.985	1.002	0.033	0.985	0.003	0.003	0.17	0.15
55	12.000	1.002	1047.30	1429.5	529.4	86.4	0.981	0.982	1.001	0.037	0.982	0.002	0.002	0.13	0.14
56	14.000	1.003	1047.56	1427.9	529.3	85.7	0.982	0.982	1.001	0.037	0.982	0.003	0.003	0.15	0.17
57	16.000	1.006	1050.55	1427.7	530.5	85.9	0.983	0.981	1.001	0.040	0.981	0.002	0.003	0.15	0.14
58	18.000	1.009	1052.67	1429.0	532.2	85.9	0.985	0.981	1.001	0.041	0.980	0.003	0.003	0.17	0.16
59	20.000	1.008	1052.43	1426.1	530.9	86.0	0.984	0.980	1.002	0.042	0.980	0.003	0.003	0.18	0.16
60	22.000	1.011	1054.80	1429.7	533.3	86.3	0.986	0.980	1.001	0.042	0.980	0.003	0.003	0.18	0.16
61	24.000	1.010	1054.26	1429.6	533.2	86.0	0.985	0.979	1.002	0.045	0.979	0.002	0.003	0.17	0.14

TEST	PART	PFX10-6	ALFA	WING	XT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	47	3.005	0000	NOVF	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	YT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BHTL
21	14.000	0.998	1043.84	1431.5	528.7	86.0	0.983	0.987	0.997	0.021	0.987	0.002	0.006	0.33	0.12
23	12.000	0.997	1042.67	1424.7	525.7	85.5	0.980	0.986	1.004	0.034	0.986	0.001	-0.001	-0.04	0.06
24	10.000	0.999	1043.66	1425.9	526.8	85.3	0.981	0.985	1.002	0.033	0.985	0.000	0.001	0.06	0.02
25	8.000	1.000	1044.89	1427.1	527.8	85.4	0.983	0.986	1.001	0.029	0.986	0.001	0.000	0.01	0.05
26	6.000	0.998	1042.69	1425.0	526.0	85.2	0.980	0.985	1.002	0.033	0.985	0.003	-0.001	-0.06	0.19
27	4.000	1.001	1045.10	1425.8	527.5	85.3	0.984	0.986	1.001	0.030	0.986	0.003	0.001	0.05	0.19
28	2.000	0.999	1044.54	1427.2	527.5	85.7	0.981	0.985	1.001	0.032	0.985	0.004	0.003	0.20	0.21
29	0.000	0.999	1044.42	1428.1	527.9	85.6	0.983	0.986	1.002	0.030	0.986	0.003	0.006	0.34	0.18
30	-2.000	0.999	1043.77	1426.1	526.9	85.4	0.981	0.985	1.002	0.032	0.985	0.001	0.002	0.14	0.08
31	-4.000	0.999	1043.97	1426.1	527.1	85.2	0.982	0.986	1.000	0.030	0.986	0.002	0.002	0.09	0.11
32	-6.000	0.999	1043.80	1426.7	527.3	85.0	0.983	0.986	1.001	0.029	0.986	0.001	0.006	0.35	0.05
33	-8.000	1.000	1044.23	1426.4	527.3	85.3	0.983	0.986	1.001	0.028	0.986	0.002	0.005	0.29	0.11
34	-10.000	0.998	1043.16	1425.7	526.5	85.3	0.982	0.986	1.002	0.030	0.986	0.002	0.005	0.30	0.13
35	-12.000	0.998	1043.11	1427.1	526.9	85.4	0.982	0.987	1.000	0.027	0.987	0.003	0.004	0.22	0.18
36	-14.000	0.998	1042.71	1427.1	526.7	85.4	0.982	0.987	1.002	0.029	0.987	0.001	0.004	0.23	0.08

TEST	PART	REX10-6	ALFA	WING	XT	YT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	42	3.004	0000	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	ZT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BNTL
5	14.000	1.001	1045.66	1428.3	528.6	85.5	0.985	0.987	1.001	0.028	0.987	0.002	0.008	0.31	0.13
6	12.000	1.000	1044.53	1427.8	527.9	85.4	0.982	0.985	1.001	0.031	0.985	0.004	0.004	0.21	0.23
7	10.000	0.999	1044.04	1425.4	526.6	85.8	0.983	0.987	1.002	0.029	0.987	0.006	0.000	0.01	0.35
8	8.000	1.000	1045.60	1428.0	528.1	86.2	0.984	0.987	1.002	0.029	0.987	0.006	0.000	0.00	0.37
9	6.000	1.000	1045.75	1430.9	529.2	86.3	0.983	0.986	1.001	0.031	0.986	0.006	-0.002	-0.12	0.34
10	4.000	0.999	1044.44	1428.8	527.9	86.1	0.983	0.986	1.001	0.029	0.986	0.006	-0.001	-0.06	0.38
11	2.000	0.998	1043.13	1428.0	527.1	85.7	0.983	0.988	1.002	0.027	0.988	0.003	0.003	0.15	0.19
12	0.000	0.999	1043.53	1426.5	526.9	85.4	0.981	0.985	1.001	0.031	0.985	0.003	0.006	0.33	0.17
13	-2.000	0.998	1043.16	1426.0	526.5	85.4	0.982	0.986	1.001	0.029	0.986	0.004	0.006	0.34	0.23
14	-4.000	0.999	1043.82	1427.8	527.5	85.4	0.983	0.986	1.001	0.029	0.986	0.003	0.006	0.34	0.17
15	-6.000	0.997	1042.24	1426.7	526.3	85.4	0.980	0.986	1.001	0.031	0.986	0.001	0.005	0.32	0.06
16	-8.000	0.998	1043.45	1426.0	526.6	85.5	0.982	0.986	1.002	0.031	0.986	0.000	0.005	0.30	0.02
17	-10.000	0.998	1043.61	1424.7	526.9	85.6	0.982	0.986	1.002	0.030	0.986	0.001	0.005	0.32	0.08
18	-12.000	0.999	1043.55	1426.6	527.0	85.4	0.979	0.984	1.001	0.034	0.984	0.003	0.006	0.34	0.18
19	-14.000	0.998	1042.73	1426.0	526.3	85.4	0.982	0.987	1.001	0.029	0.987	0.003	0.006	0.34	0.17

TEST	PART	REX10-6	ALPA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL TRANSONIC 4T
TC-484	45	3.006	****	NONE	0.00	-14.14	1-701	2- 3-77	

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
5	-6.000	1.044	1092.07	1448.2	553.9	95.3	1.014	0.976	1.002	0.050	0.976	0.002	0.007	0.42	0.09
6	-4.000	1.026	1076.79	1449.0	546.9	95.5	0.997	0.977	1.000	0.046	0.977	0.002	0.007	0.42	0.09
7	-2.000	1.020	1072.01	1455.4	547.1	95.4	0.991	0.976	1.001	0.049	0.976	0.002	0.006	0.38	0.10
8	0.000	1.021	1073.14	1450.9	545.9	95.5	0.992	0.976	1.002	0.052	0.976	0.002	0.006	0.34	0.10
9	2.000	1.019	1071.05	1451.6	545.2	95.3	0.993	0.978	1.000	0.043	0.978	0.003	0.006	0.35	0.10
10	4.000	1.015	1067.27	1453.4	544.0	95.4	0.994	0.983	1.000	0.034	0.983	0.003	0.006	0.38	0.10
11	6.000	1.018	1069.77	1451.8	544.5	95.6	0.991	0.978	1.001	0.045	0.978	0.003	0.006	0.36	0.10
12	8.000	1.025	1075.90	1451.7	547.5	95.5	0.999	0.979	1.001	0.043	0.979	0.001	0.005	0.31	0.08
13	10.000	1.032	1082.06	1451.3	550.5	95.2	1.002	0.976	1.001	0.049	0.976	0.001	0.006	0.34	0.09
14	12.000	1.041	1090.37	1449.8	553.7	95.4	1.012	0.977	1.002	0.049	0.977	0.001	0.006	0.33	0.07
15	14.000	1.037	1086.72	1452.2	552.9	95.4	1.011	0.979	1.000	0.042	0.979	0.002	0.006	0.37	0.09
16	16.000	1.043	1091.91	1453.4	555.9	95.3	1.007	0.971	1.000	0.058	0.971	0.001	0.006	0.37	0.08
17	18.000	1.051	1098.65	1451.4	558.0	95.6	1.024	0.979	1.001	0.044	0.979	0.002	0.006	0.36	0.09
18	20.000	1.040	1088.93	1449.4	552.9	95.7	1.001	0.969	1.000	0.062	0.969	0.002	0.006	0.36	0.10
19	22.000	1.033	1087.99	1455.1	552.1	95.7	1.004	0.977	1.001	0.047	0.977	0.001	0.005	0.31	0.05
20	24.000	1.027	1077.48	1453.7	549.1	95.5	0.993	0.973	1.000	0.055	0.973	0.001	0.006	0.33	0.07

TEST	PART	REX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-404	45	3.003	0000	NONE	0.00	0.00	1-702	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BATL
22	-6.000	1.024	1075.00	1453.4	547.8	95.4	0.997	0.979	0.999	0.041	0.979	0.002	0.007	0.41	0.10
23	-4.000	1.020	1071.88	1452.1	545.6	95.8	0.988	0.974	1.002	0.055	0.974	0.003	0.007	0.38	0.20
24	-2.000	1.024	1075.87	1457.7	547.8	95.7	0.988	0.970	1.001	0.060	0.970	0.002	0.006	0.36	0.10
25	0.000	1.022	1073.78	1451.6	546.3	95.8	0.990	0.974	1.000	0.053	0.974	0.002	0.005	0.32	0.08
26	2.000	1.021	1072.77	1456.7	547.8	95.7	0.994	0.978	1.001	0.045	0.978	0.002	0.006	0.33	0.12
27	4.000	1.027	1077.56	1451.3	548.3	95.3	0.996	0.975	1.001	0.052	0.975	0.002	0.005	0.32	0.12
28	6.000	1.031	1081.17	1447.1	548.5	95.1	1.007	0.981	1.002	0.041	0.981	0.002	0.005	0.30	0.10
29	8.000	1.028	1078.95	1450.0	548.4	95.4	1.006	0.982	1.001	0.037	0.982	0.002	0.006	0.33	0.13
30	10.000	1.028	1078.66	1452.7	549.2	95.6	1.007	0.983	1.001	0.035	0.983	0.002	0.005	0.30	0.09
31	12.000	1.032	1082.16	1451.3	550.5	95.2	1.001	0.975	1.001	0.053	0.975	0.002	0.004	0.26	0.11
32	14.000	1.033	1083.43	1450.9	550.8	95.6	1.000	0.973	1.001	0.055	0.973	0.002	0.005	0.31	0.12
33	16.000	1.031	1081.38	1451.3	550.1	95.4	0.995	0.971	1.001	0.060	0.971	0.002	0.005	0.32	0.11
34	18.000	1.033	1083.04	1450.9	550.8	95.2	0.997	0.971	1.002	0.061	0.971	0.002	0.005	0.32	0.12
35	20.000	1.044	1092.07	1448.2	554.0	95.1	1.007	0.971	1.002	0.061	0.971	0.002	0.006	0.34	0.11
36	22.000	1.049	1096.30	1449.8	556.3	95.7	1.014	0.973	1.002	0.057	0.973	0.001	0.006	0.34	0.07
37	24.000	1.024	1075.57	1455.5	548.5	96.1	0.996	0.978	1.001	0.046	0.978	0.001	0.006	0.36	0.08

TEST	BODY	RNX10-6	ALFA	WING	YT	ZT	RUN	SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-604	49	3.002	0.000	WOVF	14.14	0.00	1	703	2- 2-77	TRANSONIC 4T

POINT	XT	M	VV	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
39	-6.000	1.022	1073.33	1453.8	547.1	95.4	1.004	0.986	0.999	0.028	0.986	0.002	0.006	0.35	0.11
40	-4.000	1.024	1075.43	1450.8	547.1	95.3	0.987	0.970	1.002	0.053	0.970	0.001	0.002	0.14	0.05
41	-2.000	1.025	1075.79	1449.8	546.8	95.3	0.989	0.971	1.001	0.060	0.971	0.001	0.003	0.16	0.07
42	0.000	1.024	1075.83	1451.6	547.4	95.7	0.995	0.976	1.002	0.051	0.976	0.001	0.003	0.16	0.08
43	2.000	1.024	1075.26	1452.0	547.4	95.4	0.994	0.975	1.002	0.052	0.975	0.002	0.002	0.14	0.09
44	4.000	1.024	1076.70	1453.3	548.6	95.3	0.997	0.976	1.001	0.049	0.976	0.001	0.002	0.11	0.08
45	6.000	1.029	1079.82	1450.0	548.6	95.8	0.998	0.975	1.002	0.054	0.975	0.001	0.002	0.13	0.06
46	8.000	1.038	1087.50	1452.2	553.2	95.5	1.006	0.975	1.001	0.052	0.975	0.001	0.002	0.14	0.04
47	10.000	1.044	1092.79	1453.2	556.1	95.6	1.016	0.978	1.000	0.045	0.978	0.001	0.002	0.13	0.06
48	12.000	1.047	1095.03	1451.8	556.7	95.2	1.020	0.978	1.002	0.045	0.978	0.000	0.002	0.11	0.02
49	14.000	1.045	1093.79	1447.7	554.2	96.0	1.023	0.983	1.002	0.036	0.983	0.001	0.002	0.12	0.04
50	16.000	1.021	1073.44	1454.2	545.9	96.2	0.998	0.981	1.000	0.038	0.981	0.001	0.002	0.13	0.06
51	18.000	1.017	1069.88	1458.7	547.0	95.9	0.986	0.974	1.000	0.052	0.974	0.002	0.003	0.16	0.14
52	20.000	1.054	1101.64	1452.5	559.7	95.8	1.022	0.975	1.003	0.054	0.975	0.000	0.002	0.12	0.02
53	22.000	1.050	1098.64	1450.0	557.3	96.2	1.014	0.971	1.000	0.058	0.971	0.001	0.002	0.14	0.07
54	24.000	1.019	1071.89	1457.5	547.4	96.3	0.991	0.977	1.001	0.048	0.977	0.002	0.003	0.17	0.13

TEST	PART	RFX10-6	ALFA	WING	XT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	44	2.993	****	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	YT	M	VM	PT	Q	TT	YL	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
22	14.000	1.023	1073.99	1446.0	544.9	94.6	0.994	0.976	1.004	0.054	0.976	0.001	0.004	0.24	0.05
24	12.000	1.025	1075.52	1452.1	547.8	94.9	1.001	0.980	0.999	0.039	0.980	-0.001	-0.000	-0.03	-0.04
25	10.000	1.023	1074.12	1448.2	545.5	95.1	0.989	0.972	1.002	0.059	0.972	-0.001	0.001	0.05	-0.04
26	8.000	1.028	1078.51	1449.9	548.3	95.1	0.983	0.971	1.001	0.059	0.971	-0.000	-0.001	-0.06	-0.02
27	6.000	1.030	1079.72	1451.5	549.6	94.9	0.985	0.972	1.002	0.059	0.972	0.002	-0.001	-0.07	0.10
28	4.000	1.025	1075.38	1449.9	546.8	95.1	0.990	0.972	1.002	0.060	0.972	0.001	0.001	0.03	0.08
29	2.000	1.025	1075.68	1450.1	547.1	95.0	0.991	0.972	1.001	0.057	0.972	0.002	0.003	0.16	0.14
30	0.000	1.025	1075.98	1450.9	547.4	95.3	0.989	0.971	1.001	0.061	0.971	0.002	0.005	0.31	0.12
31	-2.000	1.024	1074.38	1451.2	547.0	94.6	0.989	0.971	1.002	0.060	0.971	0.000	0.002	0.10	0.02
32	-4.000	1.022	1073.18	1445.4	544.2	94.8	0.985	0.970	1.002	0.064	0.970	0.000	0.001	0.09	0.02
33	-6.000	1.022	1073.07	1449.8	545.5	95.3	0.989	0.973	1.003	0.058	0.973	-0.000	0.006	0.32	-0.00
34	-8.000	1.026	1075.72	1451.9	548.0	94.4	0.988	0.969	1.001	0.063	0.969	0.001	0.005	0.28	0.05
35	-10.000	1.025	1075.65	1443.9	544.7	95.2	0.987	0.969	1.003	0.066	0.969	0.001	0.005	0.29	0.09
36	-12.000	1.025	1078.57	1453.1	549.6	95.0	0.987	0.975	1.000	0.050	0.975	0.002	0.004	0.21	0.13
37	-14.000	1.024	1074.96	1448.5	546.0	95.2	0.990	0.972	1.002	0.059	0.972	0.000	0.004	0.21	0.02

TEST	PART	RFX10-6	ALFA	WING	XT	YT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	44	3.006	0000	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	ZT	M	VV	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BETL
6	14.000	1.026	1079.20	1448.5	546.9	93.7	0.995	0.975	1.000	0.050	0.975	0.001	0.004	0.26	0.07
7	12.000	1.028	1077.94	1449.5	548.3	94.7	0.996	0.974	0.999	0.051	0.974	0.003	0.002	0.11	0.17
8	10.000	1.027	1076.54	1448.7	547.6	93.7	0.995	0.974	1.001	0.054	0.974	0.005	-0.001	-0.07	0.20
9	8.000	1.025	1075.78	1443.5	545.1	93.9	0.990	0.970	1.001	0.061	0.970	0.005	-0.001	-0.05	0.29
10	6.000	1.025	1074.20	1448.5	545.9	94.7	0.992	0.972	1.001	0.057	0.972	0.004	-0.002	-0.14	0.26
11	4.000	1.026	1075.44	1449.4	547.2	93.9	0.991	0.971	1.002	0.060	0.971	0.005	-0.002	-0.09	0.27
12	2.000	1.023	1073.04	1447.0	545.1	94.0	0.988	0.970	1.001	0.062	0.970	0.002	0.002	0.12	0.15
13	0.000	1.021	1071.19	1445.1	543.6	93.7	0.987	0.972	1.002	0.059	0.972	0.002	0.005	0.31	0.10
14	-2.000	1.028	1078.29	1449.6	548.5	94.2	0.990	0.969	1.000	0.062	0.969	0.002	0.005	0.31	0.13
15	-4.000	1.026	1076.15	1448.1	546.7	94.6	0.988	0.969	1.001	0.063	0.969	0.002	0.005	0.32	0.13
16	-6.000	1.024	1074.65	1449.0	546.4	94.5	0.988	0.970	1.001	0.061	0.970	-0.000	0.005	0.31	-0.01
17	-8.000	1.023	1073.46	1448.6	545.7	94.5	0.989	0.972	1.001	0.059	0.972	-0.001	0.005	0.31	-0.06
18	-10.000	1.022	1071.97	1447.6	544.7	94.2	0.989	0.974	1.002	0.055	0.974	-0.000	0.005	0.30	-0.03
19	-12.000	1.021	1071.09	1448.0	544.5	94.1	0.988	0.974	1.001	0.055	0.974	0.002	0.005	0.30	0.10
20	-14.000	1.023	1072.74	1444.9	544.1	94.0	0.986	0.970	1.002	0.063	0.970	0.001	0.006	0.33	0.09

TEST	PART	REX10-6	ALFA	WING	YR	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-404	47	2.998	0000	NONE	0.00	0.00	1-707	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
22	-6.000	1.049	1096.90	1445.0	554.7	95.8	1.035	0.990	0.998	0.018	0.990	0.001	0.007	0.40	0.07
23	-4.000	1.045	1094.00	1443.8	553.0	95.8	1.020	0.980	1.001	0.042	0.980	0.002	0.005	0.31	0.13
24	-2.000	1.051	1098.93	1447.7	556.5	96.1	1.018	0.974	1.000	0.052	0.974	0.002	0.006	0.35	0.08
25	0.000	1.047	1095.98	1446.5	554.5	96.3	1.018	0.977	1.002	0.049	0.977	0.001	0.006	0.37	0.07
26	2.000	1.050	1097.87	1443.7	554.6	95.5	1.013	0.970	1.001	0.061	0.970	0.001	0.006	0.36	0.08
27	4.000	1.052	1098.70	1445.3	556.1	95.8	1.022	0.976	1.001	0.049	0.976	0.001	0.006	0.34	0.08
28	6.000	1.055	1102.34	1449.5	558.9	95.9	1.026	0.978	0.999	0.044	0.978	0.002	0.005	0.32	0.11
29	8.000	1.057	1104.32	1444.7	558.0	95.7	1.034	0.982	1.001	0.038	0.982	0.002	0.005	0.32	0.10
30	10.000	1.055	1107.75	1443.3	556.6	96.0	1.027	0.978	1.001	0.045	0.978	0.002	0.005	0.31	0.10
31	12.000	1.057	1103.73	1448.7	559.2	95.9	1.034	0.983	1.001	0.035	0.983	0.002	0.005	0.29	0.12
32	14.000	1.057	1099.85	1447.0	556.8	95.8	1.031	0.983	1.001	0.034	0.983	0.002	0.005	0.30	0.11
33	16.000	1.063	1109.44	1446.0	560.7	95.9	1.042	0.984	1.000	0.033	0.984	0.002	0.006	0.32	0.11
34	18.000	1.057	1104.65	1444.8	557.8	96.3	1.021	0.971	1.000	0.057	0.971	0.002	0.006	0.35	0.12
35	20.000	1.068	1114.12	1448.1	563.3	96.3	1.043	0.981	1.001	0.040	0.981	0.002	0.007	0.38	0.11
36	22.000	1.060	1106.47	1445.5	559.2	96.0	1.033	0.979	1.001	0.044	0.979	0.002	0.006	0.36	0.11
37	24.000	1.058	1105.34	1446.0	558.7	96.3	1.021	0.971	1.000	0.058	0.971	0.001	0.006	0.37	0.08

TEST	PART	REX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL TRANSONIC 4T
TC-464	47	2.997	0000	NONE	14.14	0.00	1-703	2- 2-77	

POINT	XP	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	SWTL
39	-6.000	1.046	1094.97	1445.3	553.9	95.9	1.035	0.991	0.998	0.015	0.991	0.003	0.006	0.34	0.15
40	-4.000	1.049	1097.13	1445.5	555.0	95.8	1.019	0.977	1.001	0.048	0.977	0.001	0.002	0.14	0.03
41	-2.000	1.049	1096.30	1444.8	554.6	95.4	1.017	0.975	1.001	0.052	0.975	0.001	0.002	0.14	0.04
42	0.000	1.049	1097.46	1443.0	556.2	95.7	1.016	0.973	1.001	0.055	0.973	0.001	0.003	0.18	0.03
43	2.000	1.053	1099.98	1446.1	556.6	95.6	1.022	0.976	1.000	0.047	0.976	0.001	0.003	0.18	0.07
44	4.000	1.050	1098.45	1444.6	555.3	96.0	1.021	0.977	1.002	0.048	0.977	0.002	0.002	0.14	0.10
45	6.000	1.055	1137.00	1444.4	556.8	95.9	1.025	0.977	1.001	0.048	0.977	0.001	0.002	0.15	0.08
46	8.000	1.054	1101.90	1446.6	557.5	96.0	1.026	0.978	1.001	0.045	0.978	0.001	0.003	0.15	0.06
47	10.000	1.053	1100.18	1446.1	556.7	95.5	1.027	0.980	1.000	0.041	0.980	0.001	0.002	0.14	0.07
48	12.000	1.057	1104.31	1443.3	557.1	95.9	1.036	0.984	1.001	0.034	0.984	0.001	0.002	0.12	0.06
49	14.000	1.058	1104.43	1444.8	558.2	95.6	1.030	0.978	1.000	0.044	0.978	0.001	0.002	0.11	0.07
50	16.000	1.058	1104.53	1442.6	557.3	95.7	1.036	0.983	1.003	0.037	0.983	0.001	0.002	0.12	0.08
51	18.000	1.056	1103.01	1444.9	557.4	95.8	1.033	0.982	1.002	0.038	0.982	0.000	0.003	0.16	0.02
52	20.000	1.053	1101.00	1444.4	556.2	96.1	1.022	0.975	1.002	0.052	0.975	0.001	0.002	0.13	0.05
53	22.000	1.058	1105.27	1447.1	559.1	96.1	1.029	0.977	1.001	0.048	0.977	0.001	0.002	0.14	0.05
54	24.000	1.055	1107.82	1445.8	557.5	96.4	1.022	0.974	1.001	0.053	0.974	0.001	0.003	0.15	0.08

TEST	PART	REX10-6	ALFA	WING	YT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	47	2.998	****	NONE	0.00	-14.14	1-701	2- 2-77	TRANSONIC 4T

POINT	XT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
5	-6.000	1.057	1099.46	1438.0	553.4	95.6	1.023	0.977	1.001	0.047	0.977	0.001	0.007	0.40	0.07
6	-4.000	1.051	1098.98	1445.8	555.8	96.1	1.027	0.981	1.001	0.038	0.981	0.002	0.007	0.41	0.11
7	-2.000	1.046	1095.72	1449.7	555.6	96.5	1.015	0.975	1.001	0.051	0.975	0.002	0.007	0.40	0.09
8	0.000	1.040	1089.81	1442.6	550.5	95.9	1.008	0.974	1.003	0.055	0.974	0.001	0.007	0.39	0.06
9	2.000	1.044	1093.43	1446.1	553.4	96.0	1.017	0.978	1.000	0.044	0.978	0.002	0.006	0.35	0.09
10	4.000	1.051	1099.43	1449.1	557.9	96.0	1.027	0.981	1.000	0.038	0.981	0.002	0.005	0.32	0.09
11	6.000	1.050	1098.74	1444.9	555.2	96.0	1.021	0.978	1.002	0.047	0.978	0.002	0.005	0.30	0.09
12	8.000	1.058	1102.81	1445.0	557.2	96.2	1.032	0.982	1.000	0.037	0.982	0.001	0.005	0.29	0.07
13	10.000	1.055	1103.01	1446.2	557.8	96.1	1.036	0.985	1.001	0.032	0.985	0.002	0.006	0.32	0.10
14	12.000	1.051	1098.93	1447.1	556.3	96.1	1.017	0.974	1.000	0.052	0.974	0.001	0.006	0.34	0.07
15	14.000	1.062	1108.14	1447.7	560.7	96.0	1.037	0.981	1.001	0.040	0.981	0.002	0.006	0.34	0.09
16	16.000	1.074	1118.37	1444.7	563.9	96.1	1.042	0.976	1.000	0.049	0.976	0.001	0.006	0.36	0.09
17	18.000	1.067	1112.44	1446.7	562.2	96.0	1.043	0.982	1.000	0.036	0.982	0.002	0.006	0.37	0.09
18	20.000	1.056	1103.72	1444.7	557.4	96.3	1.024	0.975	1.002	0.053	0.975	0.001	0.006	0.37	0.07
19	22.000	1.058	1104.99	1447.5	559.1	96.3	1.032	0.980	1.001	0.042	0.980	0.001	0.007	0.40	0.07
20	24.000	1.054	1101.49	1446.3	557.1	96.1	1.024	0.977	1.001	0.048	0.977	0.001	0.006	0.35	0.06

TEST	PART	REX10-6	ALFA	WING	XT	ZT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-404	46	3.995	0000	NOSE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	YT	M	VM	PT	Q	TT	ML	VMI/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
22	14.000	1.048	1096.35	1444.0	554.0	96.0	1.017	0.976	1.002	0.051	0.976	0.000	0.004	0.25	0.03
24	12.000	1.048	1096.17	1445.8	554.6	96.0	1.016	0.975	1.001	0.051	0.975	-0.001	-0.001	-0.03	-0.05
25	10.000	1.048	1096.40	1443.5	553.9	95.8	1.018	0.977	1.002	0.049	0.977	-0.001	0.001	0.05	-0.07
26	8.000	1.050	1098.28	1446.7	555.8	96.2	1.023	0.979	1.001	0.043	0.979	-0.000	0.000	0.00	-0.03
27	6.000	1.050	1098.78	1446.5	555.7	96.3	1.023	0.979	1.001	0.044	0.979	0.002	-0.001	-0.07	0.09
28	4.000	1.049	1097.53	1443.2	554.3	95.8	1.026	0.981	1.002	0.040	0.981	0.001	0.000	0.01	0.00
29	2.000	1.052	1100.35	1446.3	556.6	96.1	1.034	0.985	1.001	0.030	0.985	0.002	0.003	0.15	0.14
30	0.000	1.053	1100.55	1447.9	557.3	96.1	1.032	0.984	1.000	0.032	0.984	0.002	0.005	0.31	0.10
31	-2.000	1.052	1100.20	1446.4	556.6	96.2	1.031	0.984	1.000	0.033	0.984	-0.000	0.002	0.12	-0.00
32	-4.000	1.050	1097.68	1443.5	554.5	95.7	1.031	0.985	1.002	0.031	0.985	0.000	0.001	0.08	0.01
33	-6.000	1.050	1097.84	1446.4	555.6	96.0	1.029	0.984	1.001	0.034	0.984	-0.000	0.005	0.29	-0.01
34	-8.000	1.048	1096.27	1447.0	555.0	96.1	1.022	0.980	1.001	0.042	0.980	0.001	0.004	0.26	0.04
35	-10.000	1.048	1096.34	1445.6	554.6	96.0	1.019	0.977	1.000	0.046	0.977	0.002	0.005	0.29	0.10
36	-12.000	1.046	1094.83	1445.1	553.7	96.0	1.016	0.977	1.001	0.049	0.977	0.003	0.003	0.19	0.18
37	-14.000	1.047	1095.23	1445.7	554.1	96.0	1.017	0.977	1.000	0.047	0.977	0.001	0.004	0.23	0.08

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	REX10-6	ALPA	WING	XT	YT	RUN SURVEY	DATE
TC-484	46	2.997	****	NONE	14.00	0.00	1- 1	2- 2-77

POINT	ZT	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BATL
6	14.000	1.049	1097.44	1444.2	554.6	95.8	1.020	0.977	1.002	0.048	0.977	0.001	0.004	0.23	0.03
7	12.000	1.052	1099.67	1444.8	555.9	95.8	1.022	0.977	1.000	0.047	0.977	0.003	0.002	0.12	0.15
8	10.000	1.053	1101.20	1449.0	558.1	96.0	1.028	0.980	0.999	0.039	0.980	0.004	-0.001	-0.05	0.28
9	8.000	1.052	1098.93	1448.6	557.4	96.0	1.026	0.980	1.000	0.041	0.980	0.005	-0.001	-0.04	0.27
10	6.000	1.050	1097.84	1445.0	555.1	95.8	1.023	0.979	1.000	0.042	0.979	0.004	-0.003	-0.18	0.25
11	4.000	1.048	1096.65	1444.4	554.3	96.0	1.027	0.983	1.002	0.036	0.983	0.005	-0.002	-0.10	0.27
12	2.000	1.047	1095.77	1445.1	554.2	95.8	1.024	0.982	1.002	0.039	0.982	0.002	0.002	0.09	0.14
13	0.000	1.046	1094.34	1444.3	553.3	95.9	1.024	0.983	1.002	0.037	0.983	0.002	0.005	0.27	0.11
14	-2.000	1.044	1092.70	1443.2	552.0	96.0	1.019	0.980	1.001	0.041	0.980	0.003	0.006	0.33	0.18
15	-4.000	1.055	1102.56	1446.0	557.4	96.3	1.031	0.981	1.001	0.039	0.981	0.002	0.005	0.29	0.10
16	-6.000	1.050	1097.85	1445.7	555.3	96.0	1.025	0.980	1.002	0.042	0.980	-0.000	0.005	0.31	-0.01
17	-8.000	1.048	1096.45	1444.4	554.3	95.8	1.020	0.978	1.001	0.046	0.978	-0.001	0.006	0.34	-0.08
18	-10.000	1.046	1095.37	1445.2	553.9	96.2	1.018	0.978	1.002	0.047	0.978	0.000	0.006	0.34	0.01
19	-12.000	1.046	1094.93	1446.3	554.2	96.0	1.015	0.976	1.002	0.051	0.975	0.002	0.006	0.35	0.11
20	-14.000	1.044	1094.38	1445.8	553.8	96.0	1.012	0.973	1.001	0.055	0.973	0.002	0.006	0.36	0.12

TEST	PART	REX10-6	ALPA	WING	XT.	YT	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	48	3.001	****	NONE	14.00	0.00	1- 1	2- 2-77	TRANSONIC 4T

POINT	ZT	M	VM	PT	Q	TT	ML	VMI/VM	PTL/PT	CPL	UT/VM	VT/VM	WT/VM	AATL	BWTL
5	14.000	1.100	1140.84	1434.7	569.3	96.0	1.094	0.995	1.000	0.009	0.995	0.001	0.005	0.27	0.08
6	12.000	1.101	1141.26	1435.6	569.9	95.9	1.094	0.995	1.000	0.010	0.995	0.003	0.003	0.16	0.15
7	10.000	1.098	1139.64	1432.1	567.8	95.9	1.090	0.993	1.001	0.014	0.993	0.005	0.000	0.01	0.27
8	8.000	1.099	1140.15	1437.9	568.2	96.3	1.092	0.995	1.001	0.012	0.995	0.006	-0.000	-0.01	0.32
9	6.000	1.098	1139.37	1434.7	568.4	96.1	1.092	0.995	1.001	0.010	0.995	0.005	-0.002	-0.14	0.28
10	4.000	1.098	1138.48	1433.4	567.8	96.0	1.091	0.995	1.001	0.011	0.995	0.005	-0.001	-0.07	0.29
11	2.000	1.097	1138.51	1434.7	568.7	96.3	1.093	0.997	1.000	0.007	0.997	0.003	0.003	0.16	0.15
12	0.000	1.099	1139.91	1436.0	569.3	95.2	1.093	0.996	0.999	0.008	0.996	0.002	0.006	0.34	0.09
13	-2.000	1.097	1138.08	1435.3	568.5	95.7	1.094	0.997	1.000	0.005	0.997	0.002	0.006	0.34	0.12
14	-4.000	1.097	1138.08	1433.4	567.7	95.9	1.089	0.994	1.000	0.011	0.994	0.002	0.006	0.33	0.12
15	-6.000	1.102	1141.86	1434.9	569.7	96.0	1.093	0.994	0.999	0.011	0.994	0.000	0.005	0.28	0.00
16	-8.000	1.100	1140.67	1436.5	569.2	95.9	1.092	0.994	1.000	0.013	0.994	-0.001	0.006	0.33	-0.04
17	-10.000	1.100	1140.08	1432.2	568.1	95.8	1.095	0.996	1.000	0.007	0.996	-0.000	0.005	0.30	-0.00
18	-12.000	1.098	1139.37	1433.3	568.1	96.1	1.096	0.998	1.000	0.005	0.998	0.001	0.007	0.39	0.05
19	-14.000	1.099	1139.71	1437.1	569.8	96.0	1.097	0.999	0.999	0.001	0.999	0.001	0.007	0.42	0.07

TEST	PAPT	RFX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-444	85	2.995	-0.15	48	3.00	-2.00	3- 1	2- 2-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	VL	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
9	10.333	0.798	854.19	1527.7	447.6	77.9	0.799	1.001	1.002	0.001	1.001	0.009	-0.007	-0.39	0.56
10	10.466	0.803	858.56	1526.1	450.2	78.0	0.802	0.999	1.002	0.005	0.999	0.010	-0.009	-0.51	0.59
12	11.000	0.798	854.01	1528.5	447.7	78.0	0.801	1.004	1.001	-0.006	1.004	0.011	-0.011	-0.62	0.65
13	11.333	0.802	858.17	1530.0	451.1	77.9	0.803	1.001	1.001	-0.001	1.001	0.011	-0.013	-0.74	0.66
14	11.666	0.800	855.79	1529.3	449.2	77.9	0.802	1.003	1.002	-0.002	1.003	0.011	-0.015	-0.87	0.65
15	12.000	0.803	859.01	1528.6	451.3	77.9	0.805	1.002	1.001	-0.001	1.002	0.011	-0.017	-0.97	0.60
16	12.333	0.801	856.55	1528.5	449.6	77.8	0.807	1.007	1.001	-0.012	1.007	0.009	-0.018	-1.04	0.52
17	12.666	0.801	856.70	1527.7	449.4	77.9	0.811	1.011	1.001	-0.020	1.011	0.008	-0.019	-1.09	0.47
18	13.000	0.801	857.36	1529.4	450.4	78.0	0.815	1.015	1.000	-0.028	1.014	0.007	-0.019	-1.05	0.40
19	13.333	0.801	857.44	1528.3	450.0	78.1	0.817	1.018	1.001	-0.033	1.018	0.005	-0.018	-1.01	0.29
20	13.666	0.800	856.57	1529.1	449.6	78.1	0.822	1.024	1.001	-0.046	1.024	0.004	-0.016	-0.90	0.26
21	14.000	0.800	856.17	1528.5	449.3	77.8	0.824	1.027	1.002	-0.050	1.027	0.004	-0.014	-0.78	0.21
22	14.333	0.800	855.99	1527.6	449.0	77.9	0.828	1.031	1.002	-0.059	1.031	0.003	-0.011	-0.63	0.16
23	14.666	0.801	856.40	1527.7	449.4	77.9	0.832	1.034	1.001	-0.067	1.034	0.002	-0.008	-0.46	0.12
24	15.000	0.801	856.85	1527.5	449.4	78.0	0.832	1.035	1.001	-0.069	1.035	0.002	-0.005	-0.25	0.09
25	15.333	0.801	857.50	1528.7	450.2	78.1	0.836	1.038	1.001	-0.075	1.038	0.002	-0.001	-0.04	0.09
26	15.466	0.802	857.80	1531.6	451.2	78.1	0.836	1.038	1.001	-0.075	1.038	0.001	0.004	0.21	0.08
27	16.000	0.798	854.51	1531.9	448.7	78.2	0.834	1.039	1.002	-0.076	1.039	0.002	0.009	0.47	0.11
28	16.333	0.799	854.45	1529.7	448.9	78.2	0.830	1.036	1.002	-0.069	1.036	0.002	0.013	0.70	0.12
29	16.466	0.802	858.19	1526.3	450.0	78.1	0.833	1.034	1.001	-0.066	1.034	0.002	0.016	0.90	0.13
30	17.000	0.800	856.01	1527.3	448.6	78.2	0.827	1.030	1.001	-0.058	1.079	0.003	0.020	1.09	0.16
31	17.333	0.799	854.90	1529.2	448.4	78.3	0.818	1.021	1.001	-0.040	1.021	0.003	0.020	1.14	0.17
32	17.666	0.800	856.25	1530.4	449.7	78.2	0.812	1.014	1.001	-0.025	1.013	0.004	0.019	1.10	0.20
33	18.000	0.800	856.36	1528.5	449.1	78.4	0.805	1.006	1.001	-0.008	1.005	0.004	0.017	0.97	0.21
34	18.333	0.798	853.11	1529.6	448.5	78.5	0.802	1.004	1.001	-0.006	1.004	0.004	0.015	0.86	0.23
35	18.466	0.802	858.74	1530.6	451.4	78.5	0.799	0.996	1.001	0.009	0.996	0.004	0.012	0.67	0.23
36	19.000	0.795	851.71	1531.9	446.9	78.2	0.798	1.003	1.001	-0.004	1.003	0.004	0.009	0.54	0.23

TEST	PART	RNX10-6	ALPI	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	86	2.999	-0.15	45	3.00	-1.00	3- 1	2- 2-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
2	10.333	0.801	857.74	1528.6	450.1	78.4	0.797	0.995	1.001	0.012	0.995	0.012	-0.005	-0.30	0.70
3	10.666	0.801	857.88	1527.9	450.0	78.5	0.795	0.993	1.001	0.016	0.993	0.015	-0.007	-0.39	0.84
5	11.000	0.799	855.73	1531.6	449.6	78.4	0.793	0.993	1.001	0.015	0.993	0.018	-0.011	-0.66	1.03
6	11.333	0.800	856.80	1530.4	450.0	78.4	0.789	0.988	1.001	0.028	0.987	0.019	-0.017	-0.97	1.13
8	11.666	0.800	856.38	1529.8	449.4	78.6	0.794	0.994	1.001	0.016	0.993	0.021	-0.023	-1.31	1.10
9	12.000	0.802	858.64	1530.3	451.2	78.5	0.796	0.993	1.001	0.017	0.992	0.018	-0.028	-1.64	1.06
10	12.333	0.801	858.00	1529.5	450.4	78.7	0.802	1.001	1.001	-0.000	1.000	0.015	-0.032	-1.84	0.85
11	12.666	0.800	856.56	1532.1	450.7	78.8	0.809	1.010	1.000	-0.020	1.010	0.011	-0.033	-1.89	0.60
12	13.000	0.802	859.41	1530.9	451.1	78.7	0.816	1.015	1.001	-0.029	1.015	0.007	-0.032	-1.83	0.40
13	13.333	0.802	858.57	1531.0	451.3	78.7	0.823	1.023	1.001	-0.045	1.023	0.004	-0.030	-1.67	0.21
14	13.666	0.799	856.17	1532.4	449.9	78.8	0.829	1.033	1.001	-0.064	1.032	0.002	-0.026	-1.42	0.10
15	14.000	0.800	856.79	1532.0	450.2	78.8	0.834	1.038	1.002	-0.072	1.037	0.000	-0.021	-1.18	0.02
16	14.333	0.798	855.05	1532.3	449.1	78.8	0.837	1.043	1.001	-0.084	1.043	-0.000	-0.017	-0.92	-0.03
17	14.666	0.804	860.67	1531.6	452.9	78.8	0.846	1.045	1.001	-0.090	1.045	-0.000	-0.012	-0.67	-0.03
18	15.000	0.798	854.88	1533.5	449.2	78.8	0.838	1.044	1.001	-0.087	1.044	-0.001	-0.006	-0.32	-0.08
19	15.333	0.800	856.44	1529.8	449.3	78.8	0.848	1.053	1.002	-0.103	1.053	-0.001	-0.000	-0.00	-0.06
20	15.666	0.800	857.04	1532.9	450.6	79.0	0.848	1.053	1.000	-0.106	1.053	-0.000	0.006	0.34	-0.03
21	16.000	0.800	857.42	1532.6	450.8	79.0	0.848	1.052	1.001	-0.102	1.052	-0.001	0.013	0.70	-0.04
22	16.333	0.798	854.74	1533.1	448.9	79.1	0.848	1.056	1.000	-0.111	1.055	-0.001	0.020	1.06	-0.03
23	16.666	0.793	855.30	1533.1	450.1	79.0	0.843	1.048	1.001	-0.096	1.048	-0.000	0.027	1.45	-0.01
24	17.000	0.800	857.16	1531.4	450.3	78.9	0.836	1.039	1.001	-0.077	1.039	0.001	0.033	1.82	0.06
25	17.333	0.798	855.31	1532.1	449.2	78.8	0.823	1.027	1.001	-0.052	1.026	0.002	0.038	2.03	0.13
26	17.666	0.801	857.90	1530.8	450.5	79.1	0.809	1.009	1.001	-0.015	1.008	0.003	0.033	1.89	0.18
27	18.000	0.799	856.32	1532.7	449.9	79.1	0.795	0.995	1.001	0.013	0.994	0.004	0.026	1.51	0.21
28	18.333	0.799	855.64	1534.2	449.9	79.1	0.786	0.986	1.000	0.029	0.986	0.004	0.018	1.06	0.22
29	18.666	0.800	857.50	1532.6	450.8	79.1	0.785	0.983	1.001	0.036	0.983	0.004	0.013	0.77	0.26
30	19.000	0.807	858.62	1533.3	451.8	79.1	0.791	0.988	1.001	0.076	0.988	0.005	0.010	0.59	0.30

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	93	2.997	0.08	48	3.00	-2.00	3- 1	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
6	10.333	0.850	909.86	1514.3	477.4	86.3	0.851	1.002	1.001	-0.001	1.001	0.013	-0.009	-0.51	0.75
7	10.666	0.850	910.25	1514.9	478.0	86.1	0.850	1.000	1.001	0.002	1.000	0.014	-0.011	-0.62	0.79
8	11.000	0.851	911.00	1515.9	478.6	86.5	0.850	0.999	1.001	0.004	0.999	0.015	-0.013	-0.74	0.84
9	11.333	0.851	911.38	1516.2	479.0	86.4	0.849	0.998	1.001	0.006	0.998	0.015	-0.016	-0.91	0.84
10	11.666	0.852	911.73	1516.8	479.3	86.5	0.851	0.999	1.001	0.003	0.999	0.015	-0.018	-1.04	0.85
11	12.000	0.852	912.15	1515.6	479.2	86.6	0.853	1.001	1.001	-0.000	1.001	0.014	-0.020	-1.17	0.82
12	12.333	0.852	912.52	1516.4	479.6	86.7	0.858	1.005	1.001	-0.008	1.005	0.013	-0.022	-1.26	0.75
13	12.666	0.853	913.48	1516.4	480.3	86.7	0.861	1.008	1.001	-0.015	1.008	0.012	-0.023	-1.31	0.67
14	13.000	0.854	913.99	1516.6	480.7	86.7	0.867	1.013	1.001	-0.024	1.013	0.010	-0.023	-1.28	0.58
15	13.333	0.855	914.76	1516.6	481.2	86.7	0.871	1.017	1.001	-0.032	1.017	0.009	-0.022	-1.22	0.51
16	13.666	0.851	911.40	1516.7	479.1	86.4	0.876	1.025	1.001	-0.048	1.025	0.008	-0.020	-1.14	0.44
18	14.000	0.845	905.91	1515.3	475.0	86.5	0.872	1.027	1.002	-0.051	1.027	0.007	-0.018	-0.98	0.37
19	14.333	0.848	907.94	1515.4	476.5	86.2	0.877	1.029	1.001	-0.057	1.029	0.006	-0.015	-0.84	0.34
20	14.666	0.848	908.73	1514.7	476.7	86.5	0.882	1.034	1.001	-0.066	1.034	0.006	-0.012	-0.68	0.32
21	15.000	0.850	909.96	1515.4	477.9	86.2	0.886	1.037	1.001	-0.072	1.037	0.006	-0.009	-0.50	0.31
22	15.333	0.851	911.43	1516.3	479.0	86.5	0.890	1.039	1.001	-0.077	1.039	0.006	-0.005	-0.27	0.32
23	15.666	0.854	913.51	1517.2	480.7	86.5	0.893	1.040	1.000	-0.080	1.040	0.006	-0.001	-0.05	0.34
24	16.000	0.854	913.26	1515.6	480.1	86.2	0.894	1.041	1.001	-0.081	1.041	0.007	0.003	0.19	0.39
25	16.333	0.855	914.54	1515.2	480.8	86.3	0.894	1.040	1.001	-0.078	1.040	0.008	0.008	0.45	0.42
32	16.666	0.854	914.55	1516.1	480.9	86.7	0.891	1.037	1.002	-0.072	1.037	0.008	0.013	0.72	0.46
45	17.000	0.853	912.16	1508.8	477.4	86.0	0.886	1.034	1.002	-0.065	1.034	0.009	0.017	0.92	0.52
46	17.333	0.848	908.59	1522.2	479.0	86.3	0.872	1.024	1.001	-0.047	1.024	0.010	0.017	0.97	0.57
47	17.666	0.849	908.99	1522.0	479.4	86.0	0.863	1.014	1.001	-0.027	1.014	0.010	0.017	0.94	0.56
48	18.000	0.850	909.35	1515.6	477.7	85.9	0.854	1.005	1.002	-0.005	1.005	0.010	0.014	0.82	0.57
49	18.333	0.853	912.87	1514.0	479.4	86.2	0.851	0.998	1.001	0.006	0.998	0.010	0.011	0.65	0.55
53	18.666	0.849	909.42	1517.4	478.0	86.4	0.844	0.995	1.000	0.011	0.995	0.010	0.008	0.48	0.56
54	19.000	0.851	910.95	1517.9	479.2	86.5	0.846	0.995	1.001	0.011	0.995	0.010	0.006	0.35	0.55

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TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN	SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	94	2.986	0.00	49	3.00	-1.00	3-	2	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
6	10.333	0.850	909.98	1510.3	476.0	86.7	0.845	0.995	1.001	0.012	0.995	0.016	-0.007	-0.38	0.92
7	10.666	0.850	910.65	1514.4	477.8	86.6	0.844	0.994	1.000	0.013	0.994	0.018	-0.009	-0.50	1.05
8	11.000	0.850	910.56	1515.8	478.3	86.4	0.839	0.988	1.001	0.025	0.988	0.021	-0.013	-0.77	1.23
9	11.333	0.851	910.67	1515.0	478.2	86.3	0.837	0.986	1.002	0.031	0.986	0.024	-0.018	-1.07	1.38
10	11.666	0.850	910.31	1513.3	477.4	86.3	0.837	0.986	1.002	0.031	0.986	0.024	-0.026	-1.52	1.40
11	12.000	0.852	912.53	1515.6	479.4	86.6	0.841	0.989	1.001	0.024	0.988	0.022	-0.032	-1.87	1.29
12	12.333	0.853	912.70	1515.4	479.5	86.6	0.849	0.997	1.001	0.009	0.996	0.018	-0.037	-2.13	1.02
13	12.666	0.854	913.57	1516.5	480.5	86.5	0.857	1.004	1.001	-0.006	1.003	0.013	-0.038	-2.17	0.77
14	13.000	0.854	914.39	1516.9	481.1	86.5	0.867	1.013	1.000	-0.024	1.012	0.010	-0.037	-2.08	0.54
15	13.333	0.854	913.89	1515.3	480.2	86.6	0.875	1.021	1.001	-0.041	1.021	0.006	-0.034	-1.89	0.35
16	13.666	0.852	911.81	1516.3	479.2	86.5	0.883	1.032	1.001	-0.063	1.031	0.004	-0.030	-1.67	0.23
17	14.000	0.848	908.41	1515.6	476.5	86.6	0.882	1.035	1.002	-0.067	1.035	0.003	-0.026	-1.44	0.18
18	14.333	0.849	909.55	1515.6	477.4	86.6	0.888	1.039	1.001	-0.077	1.039	0.003	-0.022	-1.19	0.16
19	14.666	0.850	910.61	1515.9	478.3	86.5	0.893	1.043	1.001	-0.084	1.043	0.002	-0.017	-0.93	0.13
20	15.000	0.851	911.36	1514.3	478.2	86.6	0.899	1.048	1.002	-0.094	1.048	0.003	-0.011	-0.62	0.16
21	15.333	0.853	912.76	1515.2	479.4	86.7	0.902	1.050	1.001	-0.098	1.050	0.003	-0.006	-0.35	0.17
23	15.666	0.851	911.30	1516.2	479.0	86.3	0.904	1.054	1.002	-0.105	1.054	0.004	0.000	0.01	0.24
24	16.000	0.852	912.12	1515.9	479.2	86.7	0.904	1.053	1.001	-0.104	1.053	0.005	0.005	0.30	0.26
25	16.333	0.854	913.70	1515.0	480.0	86.7	0.906	1.053	1.001	-0.106	1.053	0.006	0.013	0.70	0.31
45	16.666	0.854	913.55	1521.1	481.9	86.5	0.906	1.053	1.001	-0.104	1.052	0.007	0.022	1.21	0.40
46	17.000	0.853	913.24	1515.2	479.7	86.7	0.897	1.044	1.001	-0.086	1.044	0.009	0.028	1.56	0.49
48	17.333	0.848	908.05	1515.2	476.3	86.6	0.878	1.031	1.001	-0.060	1.030	0.011	0.032	1.79	0.61
49	17.666	0.848	907.94	1514.2	476.0	86.5	0.855	1.008	1.001	-0.013	1.007	0.011	0.030	1.69	0.64
50	18.000	0.849	908.99	1515.6	477.1	86.5	0.842	0.993	1.001	0.016	0.993	0.011	0.024	1.36	0.65
51	18.333	0.851	911.07	1516.0	478.7	86.4	0.835	0.983	1.001	0.036	0.983	0.011	0.017	0.97	0.63
52	18.666	0.853	913.36	1516.0	480.2	86.5	0.835	0.981	1.001	0.041	0.981	0.010	0.011	0.62	0.59
59	19.000	0.849	909.13	1520.0	478.5	86.7	0.831	0.982	1.001	0.038	0.982	0.010	0.006	0.36	0.57

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	96	2.999	0.08	48	3.00	-2.00	3- 1	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PIL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
5	10.333	0.899	957.08	1485.6	497.7	87.9	0.895	0.996	1.000	0.009	0.996	0.009	-0.005	-0.27	0.49
6	10.666	0.900	957.86	1486.9	498.7	87.6	0.895	0.994	1.001	0.013	0.994	0.010	-0.007	-0.38	0.56
7	11.000	0.900	957.44	1483.2	497.1	87.8	0.892	0.992	1.002	0.018	0.992	0.010	-0.009	-0.50	0.60
8	11.333	0.901	958.46	1483.6	498.0	87.7	0.892	0.992	1.001	0.018	0.992	0.011	-0.012	-0.67	0.65
9	11.666	0.902	959.93	1487.2	499.9	88.0	0.894	0.992	1.001	0.017	0.992	0.012	-0.014	-0.83	0.68
10	12.000	0.902	959.74	1486.2	499.5	88.0	0.895	0.994	1.001	0.015	0.993	0.011	-0.017	-0.98	0.61
11	12.333	0.903	960.34	1485.9	499.7	88.0	0.899	0.996	1.001	0.009	0.996	0.010	-0.019	-1.09	0.56
12	12.666	0.903	960.60	1485.0	499.6	87.9	0.903	1.000	1.001	0.002	0.999	0.008	-0.020	-1.15	0.48
14	13.000	0.904	961.57	1486.5	500.7	87.9	0.909	1.005	1.001	-0.008	1.004	0.007	-0.020	-1.16	0.37
17	13.333	0.904	961.47	1486.0	500.5	87.9	0.913	1.009	1.001	-0.016	1.008	0.005	-0.020	-1.12	0.27
18	13.666	0.905	962.07	1486.2	500.9	87.9	0.917	1.012	1.001	-0.022	1.012	0.004	-0.019	-1.07	0.20
19	14.000	0.903	960.95	1486.3	500.2	88.0	0.922	1.018	1.001	-0.034	1.018	0.003	-0.017	-0.94	0.15
20	14.333	0.902	960.01	1487.0	499.9	88.0	0.925	1.021	1.001	-0.041	1.021	0.001	-0.014	-0.78	0.08
21	14.666	0.904	961.36	1487.5	500.9	88.0	0.928	1.023	1.001	-0.045	1.023	0.001	-0.011	-0.63	0.05
22	15.000	0.905	962.33	1485.5	500.8	88.1	0.932	1.026	1.001	-0.050	1.025	0.001	-0.008	-0.47	0.03
24	15.333	0.902	959.39	1486.6	499.4	88.0	0.934	1.031	1.001	-0.061	1.031	0.001	-0.004	-0.21	0.05
25	15.666	0.902	959.48	1487.1	499.6	88.0	0.936	1.032	1.001	-0.063	1.032	0.001	-0.000	-0.01	0.04
26	16.000	0.904	961.13	1485.1	499.9	88.1	0.938	1.032	1.001	-0.063	1.032	0.001	0.004	0.24	0.08
28	16.333	0.903	960.17	1485.1	499.4	88.0	0.941	1.036	1.001	-0.070	1.036	0.002	0.010	0.57	0.13
29	16.666	0.901	958.80	1485.9	498.8	87.9	0.938	1.035	1.001	-0.067	1.035	0.004	0.016	0.89	0.20
30	17.000	0.903	960.34	1485.4	499.6	88.0	0.935	1.030	1.002	-0.058	1.030	0.005	0.021	1.18	0.25
33	17.333	0.903	960.09	1486.3	499.7	88.0	0.926	1.022	1.001	-0.043	1.022	0.006	0.024	1.33	0.33
34	17.666	0.903	960.40	1485.5	499.6	88.0	0.913	1.010	1.001	-0.017	1.009	0.006	0.022	1.27	0.36
36	18.000	0.902	959.59	1485.1	499.0	88.0	0.903	1.001	1.001	0.001	1.000	0.007	0.020	1.12	0.38
37	18.333	0.903	960.51	1486.2	499.9	88.0	0.896	0.993	1.001	0.016	0.993	0.006	0.016	0.93	0.35
38	18.666	0.903	961.01	1485.9	500.1	88.1	0.892	0.990	1.001	0.023	0.989	0.006	0.012	0.72	0.34
39	19.000	0.904	961.12	1486.6	500.4	88.1	0.891	0.988	1.001	0.026	0.988	0.006	0.010	0.56	0.32

AD-A062 270 NIELSEN ENGINEERING AND RESEARCH INC MOUNTAIN VIEW CALIF F/G 1/3
DATA REPORT FOR A TEST PROGRAM TO STUDY TRANSONIC FLOW FIELDS A--ETC(U)
JUL 77 S C PERKINS, S S STAHARA, M J HEMSCH F44620-75-C-0047
UNCLASSIFIED NEAR-TR-138-VOL-1 AFOSR-TR-78-1485 NL

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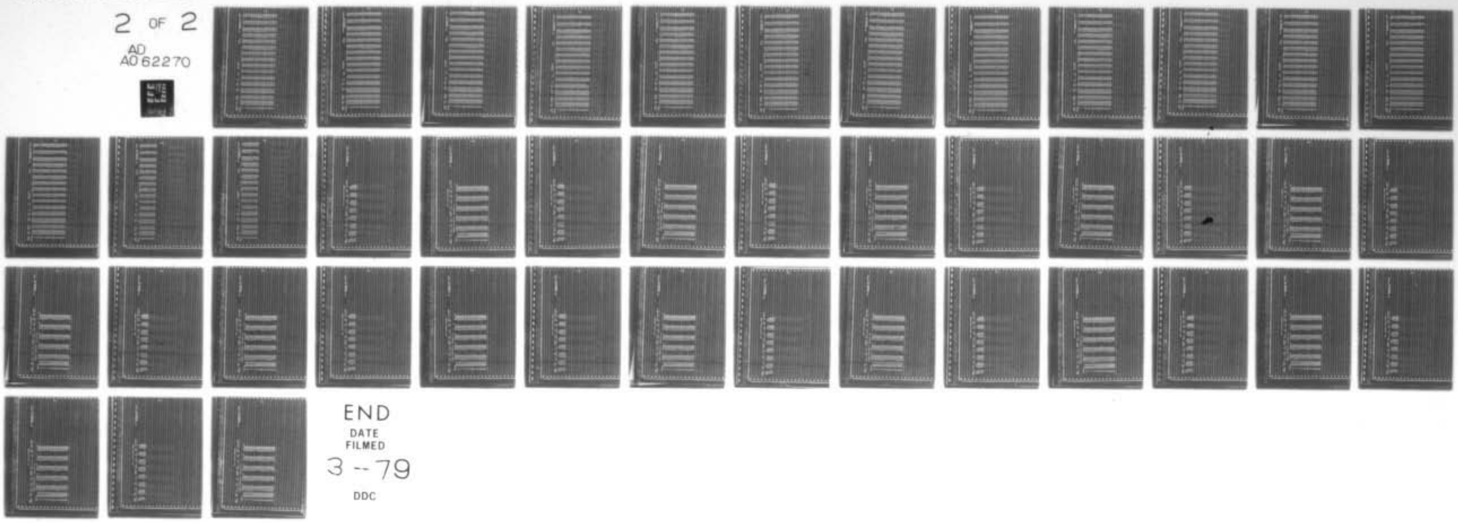

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DATE
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TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	97	2.996	0.08	48	3.00	-1.00	3- 2	2- 3-77	TRANSONIC 4T

POINT	X	P	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
5	10.333	0.897	955.74	1486.7	497.0	88.3	0.888	0.991	1.001	0.021	0.990	0.011	-0.003	-0.15	0.63
6	10.666	0.901	958.57	1486.2	498.7	88.0	0.889	0.988	1.001	0.025	0.988	0.014	-0.004	-0.26	0.78
7	11.000	0.902	959.49	1485.6	499.1	88.0	0.884	0.983	1.001	0.035	0.983	0.016	-0.008	-0.49	0.96
8	11.333	0.903	960.60	1486.7	500.1	88.1	0.882	0.980	1.001	0.041	0.980	0.020	-0.014	-0.84	1.18
9	11.666	0.903	961.11	1486.6	500.3	88.2	0.882	0.979	1.001	0.043	0.979	0.022	-0.022	-1.27	1.26
10	12.000	0.904	961.05	1487.3	500.6	88.0	0.884	0.981	1.001	0.039	0.981	0.020	-0.029	-1.69	1.15
11	12.333	0.902	959.98	1484.3	498.9	88.2	0.889	0.987	1.002	0.028	0.987	0.015	-0.034	-1.99	0.89
12	12.666	0.904	961.38	1486.7	500.6	88.0	0.898	0.994	1.001	0.012	0.994	0.011	-0.036	-2.06	0.64
15	13.000	0.901	958.58	1487.4	499.1	88.1	0.906	1.005	1.001	-0.008	1.004	0.006	-0.036	-2.03	0.33
16	13.333	0.899	957.05	1483.4	496.9	88.0	0.910	1.011	1.002	-0.017	1.010	0.002	-0.033	-1.88	0.12
17	13.666	0.902	959.92	1486.7	499.7	88.1	0.918	1.015	1.001	-0.030	1.015	0.000	-0.030	-1.68	0.00
18	14.000	0.902	959.63	1486.2	499.3	88.1	0.924	1.021	1.002	-0.039	1.020	-0.001	-0.026	-1.45	-0.07
19	14.333	0.903	961.06	1486.2	500.1	88.3	0.929	1.024	1.001	-0.047	1.024	-0.002	-0.021	-1.18	-0.12
20	14.666	0.901	958.58	1486.9	498.9	88.1	0.931	1.028	1.001	-0.056	1.028	-0.003	-0.017	-0.95	-0.17
21	15.000	0.901	959.29	1485.3	498.8	88.2	0.935	1.032	1.001	-0.062	1.032	-0.003	-0.012	-0.66	-0.15
22	15.333	0.903	960.64	1486.3	499.9	88.3	0.938	1.033	1.001	-0.066	1.033	-0.002	-0.007	-0.39	-0.14
23	15.666	0.905	962.13	1486.7	501.0	88.2	0.942	1.035	1.001	-0.069	1.035	-0.002	-0.001	-0.08	-0.10
25	16.000	0.900	957.82	1495.4	498.0	88.0	0.943	1.040	1.001	-0.080	1.040	-0.001	0.006	0.30	-0.05
26	16.333	0.900	958.11	1485.6	498.1	88.2	0.960	1.057	1.001	-0.111	1.057	-0.000	0.013	0.68	-0.01
27	16.666	0.903	960.32	1487.1	500.0	88.3	0.976	1.068	1.001	-0.136	1.068	0.002	0.023	1.24	0.09
28	17.000	0.904	962.06	1487.1	501.0	88.4	0.967	1.059	1.001	-0.116	1.058	0.004	0.032	1.74	0.23
30	17.333	0.902	959.78	1487.5	499.8	88.3	0.931	1.027	1.001	-0.053	1.027	0.007	0.037	2.09	0.40
31	17.666	0.902	960.27	1485.5	499.4	88.3	0.910	1.008	1.001	-0.013	1.007	0.008	0.036	2.03	0.47
33	18.000	0.904	961.58	1486.6	500.6	88.3	0.889	0.986	1.001	0.029	0.986	0.008	0.028	1.66	0.45
34	18.333	0.903	960.80	1487.6	500.4	88.4	0.877	0.975	1.001	0.050	0.975	0.007	0.021	1.23	0.40
35	18.666	0.904	961.89	1487.8	501.2	88.3	0.876	0.973	1.001	0.056	0.973	0.006	0.014	0.83	0.37
38	19.000	0.903	961.17	1485.9	500.0	88.4	0.880	0.978	1.001	0.046	0.978	0.006	0.010	0.56	0.36

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-404	99	2.994	0.00	48	3.00	-2.00	3- 1	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
7	10.333	0.949	1002.25	1457.4	514.4	88.6	0.927	0.980	1.001	0.040	0.980	0.010	-0.003	-0.19	0.56
8	10.666	0.949	1002.61	1459.1	515.2	88.6	0.926	0.979	1.001	0.043	0.979	0.010	-0.005	-0.29	0.59
9	11.000	0.949	1002.65	1459.0	515.1	88.6	0.923	0.977	1.001	0.048	0.977	0.011	-0.008	-0.45	0.66
10	11.333	0.949	1003.10	1458.4	515.2	88.6	0.971	0.975	1.001	0.052	0.975	0.012	-0.011	-0.66	0.72
11	11.666	0.950	1003.70	1459.1	515.9	88.4	0.922	0.975	1.001	0.052	0.975	0.013	-0.014	-0.81	0.76
12	12.000	0.950	1003.59	1458.9	515.8	88.3	0.924	0.976	1.001	0.049	0.976	0.013	-0.017	-1.01	0.74
13	12.333	0.951	1004.45	1457.8	515.9	88.3	0.926	0.977	1.001	0.046	0.977	0.012	-0.019	-1.13	0.69
14	12.666	0.952	1005.14	1458.0	516.2	88.5	0.930	0.981	1.001	0.040	0.980	0.010	-0.021	-1.22	0.60
15	13.000	0.953	1005.99	1459.2	517.1	88.6	0.933	0.983	1.001	0.036	0.983	0.009	-0.022	-1.27	0.50
16	13.333	0.953	1006.63	1459.3	517.5	88.6	0.937	0.985	1.001	0.032	0.985	0.007	-0.021	-1.23	0.42
17	13.666	0.954	1007.48	1458.0	517.5	88.6	0.940	0.987	1.002	0.029	0.987	0.005	-0.020	-1.18	0.31
20	14.000	0.954	1006.97	1459.2	517.7	88.5	0.944	0.991	1.001	0.020	0.991	0.004	-0.018	-1.07	0.24
22	14.333	0.952	1005.34	1458.9	516.7	88.5	0.964	1.010	1.001	-0.019	1.010	0.003	-0.016	-0.92	0.19
23	14.666	0.952	1004.94	1458.5	516.4	88.3	0.969	1.015	1.001	-0.028	1.015	0.003	-0.014	-0.79	0.14
24	15.000	0.953	1006.25	1457.7	516.8	88.4	0.975	1.019	1.001	-0.037	1.019	0.002	-0.011	-0.62	0.13
25	15.333	0.955	1007.56	1458.5	517.8	88.5	0.986	1.028	1.001	-0.054	1.028	0.002	-0.008	-0.42	0.09
26	15.666	0.952	1005.14	1460.5	517.1	88.6	0.991	1.035	1.001	-0.068	1.035	0.002	-0.004	-0.21	0.09
27	16.000	0.951	1004.83	1458.5	516.2	88.5	0.995	1.038	1.002	-0.075	1.038	0.002	0.001	0.05	0.10
28	16.333	0.953	1006.61	1458.2	517.1	88.6	1.007	1.047	1.001	-0.091	1.047	0.002	0.007	0.38	0.12
29	16.666	0.955	1007.64	1459.0	517.9	88.6	1.018	1.055	1.001	-0.109	1.055	0.003	0.014	0.77	0.18
30	17.000	0.953	1006.20	1459.4	517.3	88.6	1.044	1.079	1.000	-0.158	1.079	0.004	0.025	1.32	0.19
31	17.333	0.952	1005.59	1457.6	516.3	88.6	1.020	1.060	1.002	-0.117	1.059	0.009	0.030	1.61	0.40
32	17.666	0.954	1007.18	1458.0	517.3	88.6	0.964	1.008	1.001	-0.016	1.008	0.011	0.029	1.63	0.63
33	18.000	0.954	1006.73	1460.2	517.9	88.6	0.930	0.979	1.000	0.042	0.979	0.010	0.024	1.42	0.59
34	18.333	0.952	1005.38	1458.6	516.6	88.4	0.922	0.973	1.001	0.056	0.973	0.009	0.020	1.16	0.50
35	18.666	0.954	1007.30	1458.0	517.6	88.3	0.918	0.968	1.000	0.065	0.967	0.008	0.016	0.93	0.48
37	19.000	0.953	1006.43	1458.2	517.1	88.3	0.919	0.969	1.000	0.062	0.969	0.007	0.012	0.70	0.40

TEST	PART	RFX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	100	3.003	0.08	48	3.00	-1.00	3- 2	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	PL	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
5	10.333	0.948	1001.18	1461.6	515.3	88.4	0.924	0.978	1.000	0.043	0.978	0.011	-0.001	-0.07	0.66
6	10.666	0.947	1001.04	1460.3	514.8	88.4	0.919	0.974	1.001	0.053	0.974	0.014	-0.003	-0.19	0.84
7	11.000	0.947	1000.76	1455.4	513.0	88.2	0.913	0.969	1.002	0.065	0.969	0.018	-0.007	-0.44	1.07
8	11.333	0.949	1002.02	1456.4	514.1	88.2	0.911	0.966	1.001	0.070	0.965	0.022	-0.013	-0.79	1.29
9	11.666	0.950	1002.98	1459.9	515.8	88.3	0.910	0.964	1.001	0.073	0.964	0.024	-0.021	-1.27	1.41
10	12.000	0.950	1003.05	1457.7	514.9	88.3	0.911	0.965	1.001	0.072	0.964	0.022	-0.029	-1.74	1.33
11	12.333	0.951	1003.86	1457.8	515.5	88.4	0.916	0.969	1.001	0.064	0.968	0.019	-0.034	-2.01	1.11
12	12.666	0.952	1004.81	1460.4	517.0	88.4	0.923	0.974	1.001	0.054	0.973	0.013	-0.037	-2.18	0.78
13	13.000	0.952	1005.09	1458.8	516.5	88.5	0.928	0.979	1.001	0.045	0.978	0.008	-0.037	-2.17	0.48
14	13.333	0.952	1005.66	1457.7	516.4	88.5	0.934	0.983	1.001	0.035	0.983	0.005	-0.035	-2.06	0.27
15	13.666	0.954	1006.89	1459.6	517.7	88.6	0.939	0.987	1.001	0.027	0.987	0.002	-0.032	-1.86	0.12
17	14.000	0.953	1005.84	1459.1	517.0	88.5	0.943	0.991	1.001	0.019	0.991	0.001	-0.027	-1.57	0.05
18	14.333	0.954	1006.94	1460.0	518.1	88.2	0.964	1.009	1.000	-0.018	1.008	0.000	-0.024	-1.34	0.00
20	14.666	0.952	1005.27	1457.7	516.3	88.4	0.974	1.019	1.001	-0.036	1.019	-0.001	-0.019	-1.05	-0.08
21	15.000	0.951	1004.46	1458.0	515.9	88.4	0.981	1.026	1.001	-0.050	1.026	-0.001	-0.014	-0.80	-0.06
22	15.333	0.952	1005.53	1460.3	517.3	88.4	0.987	1.033	1.001	-0.054	1.033	-0.001	-0.009	-0.51	-0.05
23	15.666	0.952	1005.50	1457.4	516.3	88.3	0.995	1.037	1.002	-0.071	1.037	-0.001	-0.004	-0.24	-0.04
24	16.000	0.954	1007.07	1456.0	516.6	88.5	1.005	1.045	1.001	-0.088	1.045	-0.001	0.001	0.07	-0.04
27	16.333	0.951	1004.79	1461.1	517.1	88.6	1.021	1.061	1.001	-0.120	1.061	-0.000	0.011	0.57	-0.01
28	16.666	0.952	1004.82	1457.4	516.0	88.3	1.030	1.068	1.001	-0.135	1.068	0.001	0.019	1.03	0.05
29	17.000	0.955	1007.70	1458.9	518.1	88.3	1.047	1.080	1.000	-0.160	1.079	0.005	0.031	1.63	0.27
30	17.333	0.953	1006.35	1459.5	517.4	88.6	1.016	1.055	1.000	-0.110	1.054	0.013	0.037	2.04	0.73
31	17.666	0.953	1006.54	1459.3	517.5	88.5	0.930	0.987	1.000	0.027	0.986	0.014	0.037	2.16	0.81
37	18.000	0.951	1004.46	1457.1	515.6	88.3	0.916	0.968	1.000	0.064	0.967	0.012	0.031	1.83	0.74
38	18.333	0.953	1005.66	1456.8	516.2	88.3	0.905	0.957	1.000	0.085	0.957	0.010	0.024	1.43	0.60
39	18.666	0.954	1007.33	1458.8	517.7	88.6	0.903	0.954	0.999	0.090	0.954	0.009	0.018	1.05	0.52
41	19.000	0.952	1005.56	1459.3	516.9	88.6	0.907	0.959	1.000	0.082	0.959	0.008	0.013	0.75	0.49

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	102	3.028	0.08	48	3.00	-2.00	3- 1	2- 3-72	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
5	10.333	0.999	1046.82	1451.8	536.3	88.7	0.944	0.954	1.000	0.092	0.954	0.008	-0.005	-0.28	0.51
8	10.666	0.996	1044.69	1440.3	531.0	88.6	0.931	0.944	1.001	0.112	0.944	0.009	-0.006	-0.39	0.53
9	11.000	0.999	1046.75	1441.4	532.5	88.6	0.930	0.941	1.000	0.118	0.941	0.010	-0.009	-0.52	0.61
10	11.333	0.998	1046.31	1441.5	532.3	88.7	0.928	0.940	1.001	0.121	0.940	0.011	-0.011	-0.69	0.68
11	11.666	0.998	1046.10	1438.3	531.0	88.6	0.928	0.940	1.001	0.122	0.940	0.011	-0.015	-0.89	0.69
12	12.000	1.000	1047.61	1439.3	537.1	88.8	0.929	0.940	1.001	0.120	0.940	0.011	-0.017	-1.04	0.68
13	12.333	1.000	1048.06	1442.4	533.4	88.8	0.932	0.942	1.000	0.116	0.942	0.011	-0.020	-1.19	0.66
14	12.666	1.000	1048.23	1443.3	533.9	89.0	0.935	0.945	1.001	0.111	0.944	0.009	-0.021	-1.27	0.57
15	13.000	0.998	1046.22	1440.6	531.8	88.9	0.937	0.948	1.001	0.106	0.948	0.007	-0.022	-1.31	0.45
16	13.333	0.999	1046.78	1432.5	529.2	88.7	0.939	0.949	1.001	0.103	0.949	0.006	-0.023	-1.36	0.37
17	13.666	1.002	1050.02	1441.1	533.9	88.9	0.960	0.965	1.000	0.071	0.965	0.004	-0.021	-1.25	0.26
18	14.000	1.000	1047.46	1440.7	532.6	88.7	0.964	0.970	1.001	0.062	0.970	0.004	-0.020	-1.18	0.21
19	14.333	0.999	1046.66	1436.0	530.5	88.6	0.971	0.977	1.002	0.049	0.977	0.003	-0.018	-1.04	0.16
20	14.666	1.001	1048.39	1439.1	532.5	88.6	0.983	0.985	1.000	0.030	0.985	0.002	-0.015	-0.89	0.11
21	15.000	1.001	1048.59	1439.1	532.6	88.5	0.987	0.988	1.001	0.024	0.988	0.002	-0.013	-0.73	0.09
22	15.333	1.001	1048.99	1436.5	531.8	88.6	0.996	0.995	1.001	0.011	0.995	0.001	-0.009	-0.53	0.08
23	15.666	1.001	1048.72	1435.8	531.4	88.6	1.002	1.001	1.001	-0.001	1.001	0.001	-0.006	-0.33	0.07
24	16.000	1.002	1049.70	1438.6	533.0	88.6	1.008	1.005	1.000	-0.010	1.005	0.002	-0.001	-0.06	0.12
25	16.333	1.002	1049.83	1439.6	533.4	88.6	1.024	1.018	1.001	-0.034	1.018	0.002	0.004	0.25	0.13
26	16.666	1.002	1049.30	1434.1	531.1	88.6	1.059	1.047	1.001	-0.092	1.047	0.001	0.013	0.72	0.03
27	17.000	1.002	1050.03	1436.8	532.4	88.7	1.084	1.067	1.000	-0.133	1.066	0.001	0.020	1.06	0.03
28	17.333	1.003	1050.34	1438.9	533.3	88.8	1.111	1.087	1.000	-0.174	1.087	0.001	0.029	1.51	0.04
29	17.666	1.002	1049.48	1436.7	532.1	88.8	1.138	1.110	1.000	-0.219	1.109	0.001	0.039	2.02	0.07
30	18.000	1.003	1050.70	1437.7	533.1	88.6	1.157	1.123	0.998	-0.246	1.122	0.004	0.047	2.41	0.19
31	18.333	1.003	1050.90	1436.6	532.7	88.8	1.131	1.102	0.997	-0.207	1.101	0.011	0.038	1.96	0.59
32	18.666	1.004	1051.46	1441.8	534.8	89.1	1.006	1.002	0.998	-0.007	1.002	0.013	0.014	0.79	0.76
33	19.000	1.002	1049.75	1436.1	532.0	88.8	0.971	0.974	0.999	0.049	0.974	0.011	0.007	0.40	0.62

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	103	3.003	0.08	48	3.00	-1.00	3- 2	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
1	10.333	0.996	1044.70	1440.9	531.2	88.8	0.936	0.949	1.000	0.103	0.949	0.011	-0.004	-0.22	0.65
7	10.666	0.998	1046.52	1439.7	531.6	88.9	0.930	0.942	1.000	0.115	0.942	0.013	-0.005	-0.27	0.79
8	11.000	0.998	1046.51	1443.6	533.1	88.8	0.924	0.937	1.000	0.126	0.937	0.016	-0.008	-0.49	1.00
9	11.333	0.996	1044.91	1441.5	531.5	88.9	0.917	0.932	1.001	0.137	0.932	0.020	-0.013	-0.80	1.23
10	11.666	0.996	1044.70	1431.8	527.7	89.1	0.913	0.929	1.002	0.144	0.929	0.022	-0.021	-1.28	1.34
11	12.000	1.001	1048.99	1443.1	534.1	89.0	0.919	0.930	1.000	0.139	0.930	0.021	-0.030	-1.84	1.31
12	12.333	0.999	1046.47	1442.6	533.0	88.8	0.920	0.933	1.001	0.135	0.932	0.017	-0.035	-2.14	1.05
13	12.666	0.998	1046.07	1435.3	529.9	88.7	0.925	0.938	1.002	0.126	0.937	0.012	-0.037	-2.27	0.76
14	13.000	1.000	1047.89	1438.0	531.7	88.9	0.932	0.943	0.999	0.113	0.942	0.008	-0.037	-2.28	0.47
15	13.333	1.002	1049.84	1444.7	535.1	88.9	0.939	0.947	1.001	0.108	0.946	0.004	-0.035	-2.14	0.24
16	13.666	0.997	1045.69	1432.7	528.7	88.7	0.941	0.952	1.002	0.099	0.951	0.001	-0.032	-1.92	0.04
17	14.000	0.999	1047.53	1433.4	529.7	89.0	0.965	0.971	1.001	0.059	0.971	0.000	-0.028	-1.67	0.00
19	14.333	1.001	1049.16	1445.1	534.9	89.1	0.975	0.979	1.001	0.045	0.978	-0.001	-0.024	-1.39	-0.03
20	14.666	1.000	1048.49	1436.9	531.5	89.1	0.979	0.982	1.001	0.037	0.982	-0.001	-0.020	-1.16	-0.06
21	15.000	1.002	1049.99	1439.2	533.1	89.0	0.994	0.994	1.000	0.013	0.994	-0.002	-0.016	-0.92	-0.09
22	15.333	1.003	1051.18	1442.4	534.9	89.0	1.003	1.000	1.001	0.002	0.999	-0.001	-0.011	-0.62	-0.08
23	15.666	1.001	1048.84	1438.5	532.3	89.0	1.006	1.005	1.002	-0.007	1.005	-0.001	-0.006	-0.34	-0.05
24	16.000	1.001	1048.78	1436.9	531.6	89.1	1.014	1.011	1.001	-0.022	1.011	-0.000	-0.001	-0.05	-0.02
25	16.333	1.003	1050.72	1440.2	534.0	88.8	1.032	1.024	1.000	-0.048	1.024	0.000	0.006	0.34	0.02
26	16.666	1.003	1050.38	1440.5	533.8	89.0	1.055	1.043	1.000	-0.087	1.043	0.000	0.016	0.85	0.02
27	17.000	1.003	1050.52	1438.7	533.2	89.0	1.102	1.080	0.999	-0.161	1.080	0.000	0.028	1.50	0.01
28	17.333	1.002	1049.71	1439.4	533.1	89.0	1.148	1.117	0.999	-0.234	1.116	0.001	0.044	2.25	0.08
29	17.666	1.002	1049.98	1440.3	533.5	89.1	1.155	1.122	0.997	-0.246	1.121	0.006	0.049	2.51	0.32
30	18.000	1.001	1049.34	1437.4	532.0	89.2	1.102	1.082	0.997	-0.167	1.081	0.015	0.036	1.93	0.80
31	18.333	1.003	1050.59	1439.2	533.5	88.8	0.998	0.996	0.997	0.003	0.995	0.017	0.021	1.20	0.99
32	18.666	1.003	1050.59	1439.4	533.6	88.9	0.987	0.986	0.999	0.025	0.986	0.014	0.012	0.70	0.84
33	19.000	1.004	1052.05	1437.5	533.5	89.2	0.994	0.991	1.000	0.017	0.991	0.011	0.008	0.47	0.63

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL TRANSONIC 4T
TC-484	105	3.003	0.08	48	3.00	-2.00	3- 1	2- 3-77	

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
5	10.333	1.029	1075.33	1436.7	543.7	91.0	1.092	1.049	1.000	-0.099	1.049	-0.001	0.002	0.12	-0.05
6	10.666	1.030	1075.93	1437.5	544.3	91.0	1.073	1.034	0.999	-0.070	1.034	0.003	-0.001	-0.08	0.16
7	11.000	1.026	1072.73	1437.6	542.8	91.0	1.005	0.983	1.000	0.034	0.983	0.008	-0.008	-0.45	0.47
8	11.333	1.025	1071.96	1435.9	541.8	90.9	0.987	0.969	1.002	0.064	0.969	0.011	-0.012	-0.70	0.64
9	11.666	1.027	1073.66	1436.7	543.0	90.7	0.984	0.965	1.001	0.072	0.964	0.012	-0.017	-0.98	0.71
10	12.000	1.028	1074.22	1435.1	542.6	90.8	0.982	0.962	1.001	0.077	0.962	0.012	-0.019	-1.16	0.72
11	12.333	1.030	1075.71	1436.1	543.6	91.0	0.980	0.960	1.001	0.082	0.959	0.011	-0.022	-1.29	0.65
12	12.666	1.024	1071.01	1437.7	542.0	90.9	0.978	0.962	1.001	0.078	0.962	0.010	-0.023	-1.36	0.57
13	13.000	1.024	1070.79	1436.7	541.5	91.0	0.975	0.960	1.002	0.083	0.960	0.008	-0.023	-1.39	0.47
14	13.333	1.028	1074.19	1439.1	544.1	91.0	0.980	0.962	1.000	0.076	0.961	0.006	-0.023	-1.36	0.36
15	13.666	1.028	1074.14	1436.1	542.9	91.0	0.985	0.965	1.002	0.073	0.965	0.004	-0.022	-1.28	0.26
16	14.000	1.028	1074.08	1435.6	542.7	91.0	0.989	0.969	1.001	0.064	0.969	0.003	-0.019	-1.15	0.16
17	14.333	1.024	1070.56	1437.2	541.6	91.0	0.984	0.968	1.002	0.067	0.968	0.001	-0.016	-0.97	0.07
18	14.666	1.026	1073.21	1436.8	542.6	91.2	0.991	0.971	1.002	0.060	0.971	0.000	-0.014	-0.83	0.01
19	15.000	1.029	1075.68	1437.0	543.9	91.2	0.998	0.975	1.001	0.051	0.975	-0.000	-0.011	-0.65	-0.02
21	15.333	1.023	1070.17	1435.3	540.6	91.3	0.999	0.981	1.002	0.041	0.981	-0.001	-0.007	-0.43	-0.03
22	15.666	1.025	1071.88	1439.4	542.9	91.3	1.004	0.984	1.001	0.034	0.984	-0.001	-0.004	-0.21	-0.05
23	16.000	1.027	1073.61	1437.0	542.8	91.3	1.006	0.983	1.001	0.035	0.983	-0.000	0.001	0.04	-0.01
24	16.333	1.025	1071.93	1436.8	542.0	91.3	1.030	1.004	1.001	-0.006	1.004	-0.000	0.008	0.45	-0.00
25	16.666	1.025	1071.78	1441.7	543.7	91.3	1.057	1.026	1.000	-0.053	1.026	-0.002	0.015	0.86	-0.09
26	17.000	1.025	1072.62	1436.8	542.2	91.4	1.080	1.043	1.000	-0.086	1.043	-0.002	0.022	1.22	-0.13
27	17.333	1.029	1075.39	1437.6	543.9	91.4	1.106	1.063	1.000	-0.124	1.062	-0.002	0.031	1.68	-0.12
28	17.666	1.021	1069.19	1438.0	541.0	91.6	1.133	1.088	1.000	-0.175	1.087	-0.002	0.041	2.17	-0.09
29	18.000	1.027	1074.25	1439.8	544.2	91.4	1.151	1.096	0.998	-0.193	1.095	0.001	0.049	2.54	0.05
30	18.333	1.029	1075.81	1438.8	544.6	91.3	1.086	1.045	0.999	-0.092	1.045	0.010	0.031	1.68	0.54
34	18.666	1.022	1069.95	1441.6	542.8	91.4	0.982	0.967	0.997	0.061	0.967	0.010	0.014	0.85	0.59
35	19.000	1.027	1074.20	1437.4	543.1	91.7	0.973	0.956	0.999	0.086	0.956	0.007	0.008	0.47	0.41

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	106	2.996	0.08	48	3.00	-1.09	3- 2	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	BWL
2	10.333	1.027	1074.42	1435.9	542.7	91.6	1.081	1.042	1.000	-0.085	1.042	0.003	0.002	0.09	0.15
3	10.666	1.025	1072.82	1437.4	542.5	91.6	1.074	0.999	1.000	0.002	0.999	0.009	-0.003	-0.15	0.52
4	11.000	1.027	1073.86	1437.7	543.1	91.5	0.991	0.971	1.001	0.059	0.971	0.015	-0.008	-0.45	0.89
5	11.333	1.028	1075.16	1438.7	544.1	91.6	0.975	0.957	1.001	0.087	0.957	0.020	-0.014	-0.85	1.19
6	11.666	1.022	1070.12	1437.9	541.4	91.5	0.963	0.951	1.001	0.100	0.950	0.024	-0.021	-1.29	1.44
7	12.000	1.025	1072.38	1437.5	542.4	91.4	0.941	0.931	1.000	0.139	0.930	0.023	-0.030	-1.86	1.44
8	12.333	1.027	1074.10	1439.9	544.1	91.4	0.944	0.931	1.001	0.139	0.931	0.019	-0.036	-2.20	1.16
9	12.666	1.028	1074.66	1435.0	542.6	91.3	0.961	0.945	1.001	0.112	0.944	0.013	-0.038	-2.31	0.79
10	13.000	1.029	1075.63	1440.0	544.8	91.6	0.972	0.954	1.000	0.093	0.953	0.008	-0.038	-2.29	0.46
11	13.333	1.027	1073.61	1437.2	542.9	91.3	0.975	0.958	1.001	0.086	0.957	0.004	-0.035	-2.12	0.21
13	13.666	1.022	1069.82	1440.0	542.1	91.5	0.970	0.957	1.000	0.085	0.957	-0.000	-0.031	-1.85	-0.02
14	14.000	1.022	1069.97	1437.8	541.3	91.5	0.978	0.964	1.002	0.075	0.964	-0.001	-0.027	-1.60	-0.07
15	14.333	1.026	1073.09	1438.7	543.1	91.6	0.991	0.971	1.001	0.059	0.971	-0.002	-0.023	-1.34	-0.09
16	14.666	1.028	1075.06	1438.5	544.0	91.6	0.996	0.974	1.001	0.054	0.973	-0.003	-0.019	-1.10	-0.15
17	15.000	1.029	1075.51	1435.7	543.2	91.4	1.004	0.980	1.002	0.043	0.980	-0.003	-0.014	-0.82	-0.17
18	15.333	1.023	1071.04	1439.3	542.3	91.6	1.009	0.988	1.001	0.026	0.988	-0.003	-0.009	-0.50	-0.17
19	15.666	1.022	1069.66	1439.3	541.7	91.5	1.009	0.989	1.000	0.022	0.989	-0.003	-0.004	-0.23	-0.19
20	16.000	1.024	1071.85	1438.2	542.3	91.7	1.017	0.994	1.001	0.013	0.994	-0.003	0.002	0.10	-0.17
21	16.333	1.027	1074.32	1438.6	543.5	91.9	1.031	1.003	1.001	-0.005	1.003	-0.002	0.009	0.49	-0.10
24	16.666	1.022	1070.14	1436.8	540.8	91.9	1.052	1.024	1.000	-0.047	1.024	-0.002	0.018	0.98	-0.12
26	17.000	1.025	1072.40	1438.3	542.7	91.5	1.092	1.053	1.001	-0.105	1.052	-0.002	0.029	1.60	-0.11
27	17.333	1.029	1075.38	1437.2	543.7	91.5	1.135	1.083	0.999	-0.167	1.082	-0.001	0.043	2.27	-0.04
29	17.666	1.024	1071.39	1440.0	542.7	91.7	1.136	1.088	0.997	-0.179	1.087	0.006	0.046	2.42	0.38
30	18.000	1.025	1072.59	1438.3	542.7	91.7	1.076	1.041	0.997	-0.086	1.040	0.013	0.035	1.94	0.71
31	18.333	1.030	1076.93	1441.6	545.8	92.0	0.986	0.964	0.996	0.066	0.964	0.014	0.022	1.30	0.88
38	18.666	1.021	1069.30	1439.0	541.2	92.0	0.978	0.965	1.000	0.070	0.965	0.010	0.013	0.78	0.62
39	19.000	1.023	1070.89	1439.6	542.2	91.9	0.993	0.976	1.000	0.049	0.976	0.008	0.010	0.59	0.49

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	108	3.002	0.00	42	3.00	-2.00	3- 1	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
6	10.333	1.049	1093.25	1433.5	550.3	92.0	1.110	1.047	0.999	-0.096	1.047	0.001	0.002	0.09	0.07
7	10.666	1.052	1095.81	1431.3	550.8	91.7	1.068	1.013	1.000	-0.025	1.013	0.006	-0.002	-0.12	0.36
8	11.000	1.050	1094.33	1431.3	550.1	91.8	1.046	0.997	1.000	0.007	0.997	0.010	-0.008	-0.45	0.56
9	11.333	1.052	1095.62	1433.1	551.3	91.8	1.015	0.971	1.000	0.058	0.971	0.013	-0.014	-0.83	0.77
10	11.666	1.053	1096.45	1431.6	551.1	91.8	1.015	0.970	1.000	0.060	0.970	0.014	-0.019	-1.15	0.83
11	12.000	1.054	1097.84	1432.8	552.2	91.8	1.015	0.969	1.000	0.062	0.969	0.014	-0.022	-1.33	0.85
13	12.333	1.046	1090.73	1432.6	548.9	91.9	1.010	0.971	1.001	0.059	0.971	0.013	-0.026	-1.55	0.76
14	12.666	1.049	1093.25	1432.9	550.2	91.9	1.008	0.968	1.000	0.064	0.967	0.011	-0.028	-1.63	0.66
15	13.000	1.051	1095.42	1431.9	550.7	91.9	1.015	0.971	1.001	0.060	0.971	0.009	-0.028	-1.63	0.54
16	13.333	1.053	1097.05	1432.8	551.8	91.9	1.023	0.976	1.001	0.049	0.976	0.006	-0.027	-1.60	0.37
17	13.666	1.054	1097.54	1429.8	550.9	91.9	1.035	0.986	1.001	0.031	0.985	0.005	-0.025	-1.47	0.31
20	14.000	1.051	1094.83	1432.1	550.6	91.9	1.051	1.000	1.001	0.001	1.000	0.004	-0.021	-1.22	0.25
21	14.333	1.050	1094.68	1431.8	550.3	92.0	1.056	1.004	1.001	-0.007	1.004	0.004	-0.019	-1.06	0.21
22	14.665	1.053	1096.51	1431.9	551.3	91.8	1.072	1.015	1.000	-0.029	1.014	0.003	-0.015	-0.83	0.16
24	15.000	1.052	1096.06	1431.8	551.0	91.9	1.089	1.028	1.000	-0.056	1.028	0.001	-0.009	-0.50	0.08
25	15.333	1.049	1093.15	1432.7	550.0	92.0	1.095	1.036	1.000	-0.073	1.036	0.000	-0.006	-0.35	0.03
26	15.666	1.050	1094.59	1433.8	551.1	92.0	1.109	1.045	1.000	-0.090	1.045	0.000	-0.003	-0.16	0.00
27	16.000	1.053	1096.36	1433.4	551.7	91.9	1.119	1.051	1.000	-0.102	1.051	0.001	0.000	0.02	0.05
28	16.333	1.054	1097.84	1431.6	551.7	91.9	1.131	1.058	1.000	-0.116	1.058	0.002	0.004	0.20	0.08
30	16.666	1.052	1095.57	1432.0	550.8	92.0	1.139	1.066	0.999	-0.133	1.066	0.000	0.011	0.58	0.03
31	17.000	1.050	1094.39	1433.2	550.7	92.0	1.131	1.062	0.999	-0.123	1.061	-0.005	0.021	1.14	-0.30
32	17.333	1.052	1095.56	1431.2	550.5	92.0	1.152	1.076	0.999	-0.152	1.076	-0.005	0.029	1.56	-0.27
34	17.666	1.046	1090.67	1432.1	548.6	92.1	1.156	1.084	1.000	-0.167	1.083	0.001	0.040	7.09	0.03
35	18.000	1.049	1093.55	1433.3	550.3	92.1	1.180	1.099	0.998	-0.198	1.097	-0.001	0.051	2.64	-0.06
36	18.333	1.051	1095.67	1432.4	550.9	92.2	1.191	1.105	0.996	-0.213	1.104	0.001	0.055	2.85	0.05
38	18.666	1.053	1097.05	1431.4	551.2	92.1	1.144	1.069	0.994	-0.144	1.069	0.009	0.026	1.39	0.50
40	19.000	1.048	1092.50	1431.8	549.3	92.0	1.001	0.963	0.997	0.069	0.963	0.013	0.008	0.47	0.75

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	109	2.994	0.08	48	3.00	-1.00	3- 2	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
2	10.333	1.046	1090.81	1430.0	548.0	91.8	1.099	1.041	0.999	-0.082	1.041	0.004	0.001	0.07	0.20
3	10.666	1.051	1095.22	1432.7	550.9	92.0	1.058	1.006	0.999	-0.012	1.005	0.011	-0.002	-0.10	0.64
4	11.000	1.052	1096.21	1432.5	551.3	92.0	1.013	0.969	1.000	0.063	0.969	0.018	-0.007	-0.44	1.08
5	11.333	1.053	1096.75	1430.6	550.8	92.0	1.000	0.958	1.001	0.085	0.958	0.024	-0.014	-0.81	1.42
7	11.666	1.052	1096.21	1432.5	551.3	92.0	0.994	0.954	1.001	0.094	0.953	0.027	-0.024	-1.44	1.64
8	12.000	1.053	1096.21	1431.7	551.1	91.8	0.992	0.952	1.001	0.097	0.952	0.026	-0.032	-1.95	1.58
9	12.333	1.054	1097.64	1432.0	551.7	92.0	0.996	0.954	1.001	0.093	0.953	0.022	-0.038	-2.30	1.31
11	12.666	1.051	1095.27	1433.2	551.1	92.0	1.003	0.962	1.000	0.077	0.961	0.015	-0.042	-2.51	0.88
12	13.000	1.051	1095.22	1432.5	550.8	92.0	1.006	0.964	1.001	0.073	0.963	0.009	-0.042	-2.50	0.56
13	13.333	1.053	1097.05	1431.8	551.4	91.9	1.012	0.967	1.001	0.067	0.966	0.005	-0.040	-2.36	0.28
14	13.666	1.055	1098.38	1432.1	552.2	91.9	1.016	0.969	1.000	0.062	0.969	0.003	-0.037	-2.18	0.18
16	14.000	1.046	1090.53	1432.6	548.8	91.9	1.021	0.980	1.000	0.040	0.980	0.C01	-0.031	-1.84	0.07
20	14.333	1.051	1094.83	1433.1	550.9	92.0	1.027	0.981	1.000	0.038	0.981	0.001	-0.027	-1.56	0.07
21	14.666	1.054	1097.45	1433.6	552.3	92.0	1.037	0.987	0.999	0.025	0.987	0.001	-0.023	-1.33	0.07
22	15.000	1.054	1097.40	1432.3	551.7	92.1	1.049	0.996	1.000	0.007	0.996	0.00U	-0.019	-1.07	0.01
24	15.333	1.049	1093.25	1432.5	550.0	91.8	1.061	1.010	1.000	-0.019	1.009	0.000	-0.014	-0.80	0.01
25	15.666	1.049	1093.79	1432.0	550.0	92.0	1.069	1.015	0.999	-0.031	1.015	-0.001	-0.010	-0.54	-0.07
26	16.000	1.051	1095.32	1432.5	550.9	92.0	1.100	1.037	0.999	-0.075	1.037	-0.001	-0.005	-0.27	-0.08
27	16.333	1.054	1097.79	1432.7	552.1	92.0	1.132	1.059	0.999	-0.119	1.059	-0.002	0.004	0.22	-0.10
28	16.666	1.054	1097.59	1432.3	551.8	92.0	1.173	1.090	0.998	-0.181	1.090	-0.003	0.021	1.13	-0.15
30	17.000	1.051	1095.32	1432.1	550.7	92.0	1.199	1.111	0.997	-0.223	1.110	-0.003	0.039	2.04	-0.16
31	17.333	1.050	1094.14	1433.7	550.8	92.1	1.195	1.109	0.996	-0.220	1.108	-0.007	0.054	2.79	-0.38
32	17.666	1.052	1095.77	1433.4	551.4	92.0	1.214	1.121	0.994	-0.245	1.119	-0.005	0.067	3.44	-0.26
33	18.000	1.054	1097.64	1431.6	551.5	92.2	1.202	1.111	0.991	-0.228	1.109	0.002	0.058	3.00	0.10
35	18.333	1.047	1091.62	1433.4	549.6	92.0	1.031	0.987	0.995	0.019	0.987	0.017	0.021	1.25	0.97
36	18.666	1.047	1091.82	1433.2	549.6	92.0	1.018	0.977	0.999	0.044	0.977	0.016	0.014	0.84	0.96
37	19.000	1.050	1094.58	1433.2	550.8	92.0	1.029	0.983	0.999	0.033	0.983	0.013	0.010	0.60	0.74

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-404	111	3.000	0.08	48	3.00	-2.00	3- 1	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
11	10.333	1.099	1135.68	1420.3	563.2	91.9	1.161	1.044	0.997	-0.092	1.044	0.000	0.003	0.18	0.02
12	10.666	1.099	1135.89	1422.2	564.0	92.0	1.152	1.038	0.997	-0.079	1.038	0.002	0.003	0.18	0.10
13	11.000	1.099	1135.88	1420.5	563.3	92.0	1.127	1.020	0.998	-0.042	1.020	0.006	-0.001	-0.07	0.33
14	11.333	1.101	1137.26	1421.1	564.1	92.0	1.064	0.973	0.998	0.052	0.973	0.012	-0.012	-0.72	0.72
15	11.666	1.103	1138.54	1421.7	564.8	92.0	1.026	0.943	0.998	0.113	0.942	0.014	-0.019	-1.14	0.83
16	12.000	1.105	1140.21	1422.5	565.8	92.0	1.034	0.948	0.999	0.104	0.947	0.014	-0.023	-1.41	0.82
17	12.333	1.104	1139.89	1421.4	565.2	92.0	1.030	0.945	1.001	0.112	0.944	0.012	-0.025	-1.54	0.75
18	12.666	1.105	1140.27	1419.6	564.7	92.0	1.027	0.942	1.000	0.117	0.942	0.011	-0.026	-1.61	0.66
21	13.000	1.099	1135.29	1419.7	562.7	92.0	1.026	0.946	1.001	0.110	0.945	0.009	-0.026	-1.60	0.54
22	13.333	1.099	1135.71	1421.0	563.4	92.0	1.029	0.948	1.001	0.105	0.947	0.007	-0.026	-1.60	0.40
23	13.666	1.100	1136.27	1420.5	563.5	92.0	1.040	0.955	1.001	0.091	0.955	0.005	-0.025	-1.49	0.29
24	14.000	1.101	1136.98	1419.9	563.5	92.0	1.056	0.967	1.001	0.067	0.967	0.004	-0.022	-1.30	0.22
25	14.333	1.103	1138.71	1420.6	564.4	92.0	1.073	0.978	1.001	0.045	0.978	0.002	-0.019	-1.09	0.11
28	14.666	1.098	1134.74	1419.8	562.6	92.0	1.083	0.989	1.001	0.023	0.989	0.001	-0.014	-0.81	0.04
29	15.000	1.098	1135.30	1421.4	563.3	92.2	1.089	0.993	1.000	0.013	0.993	0.001	-0.010	-0.60	0.04
30	15.333	1.100	1136.26	1421.1	563.6	92.2	1.105	1.004	1.000	-0.007	1.004	0.000	-0.007	-0.40	0.01
31	15.666	1.102	1138.25	1422.8	565.1	92.0	1.117	1.011	0.998	-0.023	1.011	0.000	-0.003	-0.16	0.01
33	16.000	1.104	1139.78	1422.7	565.6	92.2	1.133	1.021	0.999	-0.042	1.021	0.001	0.000	0.01	0.06
34	16.333	1.098	1134.56	1421.1	563.0	92.1	1.131	1.024	1.000	-0.048	1.024	0.003	0.006	0.32	0.18
35	16.666	1.099	1135.73	1420.0	563.0	92.1	1.144	1.033	1.000	-0.066	1.033	0.005	0.010	0.54	0.26
36	17.000	1.101	1137.74	1421.5	564.4	92.1	1.163	1.045	0.998	-0.091	1.044	0.004	0.016	0.88	0.23
37	17.333	1.103	1139.31	1421.7	565.1	92.0	1.183	1.057	0.998	-0.115	1.056	0.004	0.024	1.29	0.22
39	17.666	1.101	1137.85	1420.7	564.1	92.2	1.198	1.069	0.996	-0.141	1.068	0.006	0.032	1.73	0.30
40	18.000	1.100	1136.88	1421.6	564.0	92.2	1.215	1.082	0.995	-0.166	1.081	0.008	0.040	2.14	0.40
41	18.333	1.102	1138.22	1420.0	563.9	92.3	1.223	1.086	0.992	-0.177	1.085	0.010	0.047	2.48	0.53
42	18.666	1.104	1140.07	1421.4	565.2	92.3	1.228	1.087	0.991	-0.181	1.086	0.014	0.051	2.68	0.73
43	19.000	1.100	1136.81	1422.1	564.2	92.2	1.187	1.062	0.990	-0.134	1.062	0.017	0.013	0.68	0.89

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	112	2.997	0.08	48	3.00	-1.00	3- 2	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
3	10.333	1.103	1138.99	1418.7	563.8	92.0	1.163	1.043	0.998	-0.088	1.043	0.001	0.003	0.14	0.08
7	10.666	1.101	1137.76	1422.0	564.6	92.2	1.146	1.032	0.997	-0.067	1.032	0.006	0.000	0.00	0.36
8	11.000	1.098	1135.14	1421.9	563.5	92.1	1.055	0.968	0.998	0.061	0.968	0.019	-0.008	-0.48	1.14
9	11.333	1.098	1135.15	1421.3	563.3	92.1	1.013	0.936	0.999	0.128	0.935	0.026	-0.016	-0.96	1.57
10	11.666	1.099	1135.76	1420.0	563.0	92.1	1.013	0.935	1.001	0.131	0.935	0.028	-0.024	-1.46	1.72
11	12.000	1.102	1138.19	1423.5	565.4	92.1	1.014	0.934	1.000	0.132	0.933	0.027	-0.033	-2.02	1.64
12	12.333	1.101	1137.49	1421.6	564.4	92.0	1.010	0.932	1.002	0.140	0.930	0.022	-0.039	-2.40	1.35
13	12.666	1.103	1138.76	1420.7	564.5	92.0	1.008	0.929	1.001	0.143	0.928	0.015	-0.041	-2.54	0.95
14	13.000	1.104	1139.83	1419.8	564.5	92.1	1.010	0.930	1.001	0.143	0.929	0.010	-0.041	-2.54	0.64
15	13.333	1.098	1134.77	1421.1	563.0	92.2	1.013	0.936	1.000	0.128	0.935	0.006	-0.039	-2.38	0.35
16	13.666	1.098	1135.11	1421.4	563.3	92.1	1.017	0.939	1.001	0.123	0.939	0.002	-0.035	-2.15	0.13
17	14.000	1.099	1136.02	1420.8	563.4	92.1	1.021	0.941	1.000	0.118	0.941	0.000	-0.032	-1.93	0.02
18	14.333	1.102	1137.84	1421.5	564.5	92.0	1.034	0.950	1.000	0.100	0.949	0.000	-0.028	-1.68	0.01
19	14.666	1.102	1138.16	1420.9	564.3	92.0	1.048	0.960	1.000	0.081	0.959	-0.001	-0.024	-1.44	-0.03
20	15.000	1.104	1139.68	1421.0	565.0	91.9	1.062	0.969	1.000	0.062	0.969	-0.001	-0.020	-1.18	-0.06
22	15.333	1.102	1138.10	1420.4	564.1	92.0	1.081	0.984	0.999	0.030	0.984	-0.002	-0.015	-0.87	-0.08
23	15.666	1.099	1135.92	1420.6	563.4	92.0	1.082	0.987	0.999	0.025	0.967	0.001	-0.010	-0.60	0.04
24	16.000	1.101	1137.56	1420.7	564.0	92.0	1.112	1.008	0.998	-0.017	1.008	0.001	-0.005	-0.28	0.05
25	16.333	1.103	1139.19	1421.0	564.7	92.2	1.151	1.034	0.997	-0.071	1.034	-0.001	0.005	0.29	-0.05
28	16.666	1.097	1133.97	1419.0	562.0	92.0	1.169	1.052	0.998	-0.105	1.052	0.001	0.022	1.20	0.06
29	17.000	1.099	1135.55	1421.5	563.5	92.1	1.200	1.072	0.997	-0.146	1.072	0.002	0.039	2.09	0.10
30	17.333	1.100	1136.85	1419.6	563.2	92.2	1.216	1.082	0.994	-0.168	1.080	0.006	0.052	2.75	0.31
31	17.666	1.103	1139.15	1422.6	565.3	92.3	1.232	1.091	0.992	-0.187	1.089	0.005	0.064	3.35	0.29
32	18.000	1.103	1139.48	1421.2	564.9	92.2	1.229	1.088	0.989	-0.185	1.087	0.012	0.061	3.24	0.65
33	18.333	1.100	1136.29	1421.4	563.7	92.2	1.174	1.054	0.989	-0.118	1.053	0.018	0.015	0.84	0.97
34	18.666	1.101	1137.08	1421.4	564.1	92.1	1.054	0.966	0.990	0.056	0.965	0.019	0.000	0.03	1.13
35	19.000	1.102	1138.01	1420.4	564.0	92.3	1.090	0.991	1.000	0.017	0.991	0.018	-0.002	-0.12	1.07

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	114	2.997	0.08	49	3.00	-2.00	3- 1	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SML
6	10.333	1.146	1174.69	1412.5	574.2	92.1	1.199	1.036	0.994	-0.078	1.036	-0.000	0.004	0.21	-0.02
7	10.666	1.148	1175.93	1414.2	575.3	92.1	1.199	1.035	0.994	-0.075	1.035	0.001	0.003	0.17	0.03
8	11.000	1.149	1176.52	1413.5	575.2	92.1	1.194	1.031	0.994	-0.067	1.031	0.002	0.002	0.12	0.10
9	11.333	1.151	1178.13	1413.5	575.7	92.1	1.186	1.024	0.994	-0.054	1.024	0.004	-0.001	-0.07	0.23
10	11.666	1.151	1177.89	1413.2	575.6	92.0	1.127	0.984	0.996	0.027	0.984	0.015	-0.018	-1.05	0.85
11	12.000	1.151	1178.62	1413.7	576.0	92.1	1.116	0.977	0.999	0.045	0.976	0.014	-0.022	-1.31	0.84
12	12.333	1.152	1179.00	1413.9	576.2	92.0	1.120	0.978	0.999	0.044	0.977	0.012	-0.024	-1.44	0.69
13	12.666	1.153	1179.81	1413.7	576.4	92.0	1.124	0.980	0.999	0.039	0.979	0.009	-0.025	-1.46	0.50
14	13.000	1.154	1180.68	1414.3	577.0	91.9	1.131	0.984	0.999	0.030	0.984	0.007	-0.025	-1.45	0.39
20	13.333	1.154	1180.43	1414.0	576.7	92.0	1.138	0.989	0.999	0.020	0.989	0.005	-0.024	-1.39	0.30
21	13.666	1.154	1180.33	1414.5	577.0	91.7	1.142	0.992	0.998	0.015	0.991	0.003	-0.023	-1.35	0.16
22	14.000	1.154	1181.01	1412.6	576.3	92.0	1.146	0.994	0.998	0.009	0.994	0.001	-0.021	-1.19	0.07
24	14.333	1.152	1179.54	1413.8	576.3	92.1	1.153	1.001	0.999	-0.003	1.000	0.001	-0.018	-1.03	0.05
25	14.666	1.153	1180.04	1415.6	577.2	92.1	1.151	0.999	0.998	0.000	0.999	0.000	-0.016	-0.93	0.02
26	15.000	1.153	1179.58	1413.0	576.0	92.0	1.158	1.003	0.999	-0.008	1.003	-0.001	-0.013	-0.73	-0.05
27	15.333	1.155	1181.42	1412.6	576.4	92.1	1.166	1.008	0.997	-0.019	1.008	-0.001	-0.009	-0.49	-0.07
29	15.666	1.151	1178.27	1415.2	576.4	92.2	1.174	1.016	0.996	-0.035	1.016	-0.000	-0.004	-0.23	-0.01
30	16.000	1.151	1178.27	1413.4	575.7	92.3	1.177	1.018	0.997	-0.039	1.018	-0.001	-0.002	-0.13	-0.04
31	16.333	1.152	1178.77	1413.2	575.9	92.0	1.185	1.022	0.996	-0.049	1.022	-0.001	0.001	0.07	-0.05
32	16.666	1.154	1180.47	1414.1	576.8	92.0	1.197	1.029	0.995	-0.063	1.029	-0.000	0.006	0.36	-0.02
33	17.000	1.155	1181.25	1414.7	577.2	92.1	1.204	1.033	0.994	-0.072	1.033	0.000	0.011	0.63	0.00
34	17.333	1.153	1179.71	1414.4	576.6	92.2	1.213	1.040	0.994	-0.086	1.040	0.000	0.018	1.01	0.01
35	17.666	1.150	1177.65	1412.7	575.2	92.2	1.221	1.048	0.993	-0.102	1.048	0.001	0.027	1.50	0.07
36	18.000	1.152	1178.77	1414.1	576.2	92.0	1.234	1.055	0.991	-0.118	1.054	-0.001	0.038	2.06	-0.05
37	18.333	1.153	1179.80	1414.4	576.7	92.0	1.247	1.063	0.988	-0.135	1.062	0.002	0.048	2.60	0.08
38	18.666	1.153	1179.88	1412.1	575.8	92.0	1.255	1.068	0.987	-0.146	1.066	0.006	0.054	2.91	0.32
41	19.000	1.150	1177.05	1414.1	575.7	91.9	1.254	1.070	0.985	-0.152	1.068	0.011	0.056	3.00	0.59

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	115	3.002	0.08	48	3.00	-1.00	3- 2	2- 3-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
8	10.333	1.151	1178.33	1414.2	576.1	92.0	1.201	1.034	0.993	-0.074	1.034	-0.000	0.003	0.18	-0.01
9	10.666	1.153	1179.58	1413.4	576.2	92.0	1.199	1.032	0.994	-0.069	1.032	0.001	0.003	0.16	0.05
10	11.000	1.155	1181.22	1413.7	576.8	92.1	1.175	1.014	0.993	-0.035	1.014	0.011	-0.004	-0.23	0.60
11	11.333	1.152	1179.14	1413.6	576.1	92.0	1.092	0.958	0.995	0.079	0.958	0.026	-0.019	-1.12	1.57
12	11.666	1.151	1178.75	1413.1	575.8	92.1	1.084	0.953	1.000	0.095	0.952	0.028	-0.029	-1.73	1.68
13	12.000	1.153	1180.24	1413.8	576.6	92.0	1.090	0.956	1.000	0.089	0.955	0.024	-0.037	-2.23	1.41
14	12.333	1.154	1180.28	1414.8	577.0	92.0	1.093	0.958	0.999	0.084	0.957	0.016	-0.042	-2.50	0.96
16	12.666	1.154	1180.72	1412.2	576.1	91.9	1.102	0.964	0.999	0.071	0.963	0.007	-0.043	-2.57	0.45
17	13.000	1.152	1179.34	1414.5	576.6	92.0	1.109	0.970	0.999	0.060	0.969	0.003	-0.042	-2.47	0.20
18	13.333	1.152	1179.46	1413.4	576.2	92.0	1.117	0.976	0.998	0.047	0.975	0.001	-0.039	-2.28	0.03
19	13.666	1.154	1180.21	1413.1	576.4	91.8	1.129	0.983	0.997	0.031	0.982	-0.002	-0.034	-2.00	-0.09
21	14.000	1.154	1181.03	1414.0	576.9	92.0	1.140	0.990	0.997	0.017	0.990	-0.003	-0.029	-1.71	-0.15
22	14.333	1.151	1178.35	1413.4	575.8	92.0	1.132	0.987	0.997	0.023	0.987	-0.002	-0.027	-1.54	-0.10
23	14.666	1.152	1179.00	1414.4	576.4	92.0	1.145	0.995	0.997	0.006	0.995	-0.002	-0.022	-1.28	-0.14
24	15.000	1.152	1179.36	1413.4	576.1	92.0	1.160	1.005	0.996	-0.014	1.005	-0.004	-0.016	-0.90	-0.20
25	15.333	1.154	1180.32	1413.3	576.4	92.0	1.176	1.016	0.996	-0.035	1.016	-0.004	-0.009	-0.51	-0.25
27	15.666	1.149	1176.91	1412.6	575.0	92.0	1.179	1.020	0.996	-0.044	1.020	-0.003	-0.004	-0.24	-0.15
28	16.000	1.150	1177.63	1413.8	575.7	92.0	1.187	1.025	0.995	-0.055	1.025	-0.002	0.003	0.19	-0.13
29	16.333	1.151	1178.21	1413.3	575.7	92.0	1.205	1.037	0.995	-0.076	1.037	-0.003	0.014	0.77	-0.17
30	16.666	1.152	1179.23	1412.9	575.9	92.0	1.217	1.043	0.994	-0.092	1.043	-0.004	0.022	1.20	-0.23
31	17.000	1.154	1180.53	1414.0	576.8	92.0	1.230	1.051	0.991	-0.110	1.051	-0.004	0.034	1.84	-0.22
33	17.333	1.149	1177.10	1413.5	575.4	92.1	1.246	1.065	0.990	-0.138	1.064	-0.002	0.049	2.65	-0.13
34	17.666	1.152	1179.40	1412.8	575.9	92.1	1.259	1.071	0.986	-0.153	1.069	-0.000	0.061	3.26	-0.01
35	18.000	1.153	1180.34	1412.9	576.2	92.1	1.266	1.075	0.982	-0.163	1.073	0.003	0.065	3.46	0.15
36	18.333	1.153	1180.31	1414.4	576.8	92.1	1.260	1.071	0.982	-0.157	1.070	0.012	0.058	3.11	0.63
38	18.666	1.151	1178.20	1413.5	575.8	92.1	1.094	0.960	0.982	0.059	0.960	0.017	-0.010	-0.60	1.03
39	19.000	1.151	1178.48	1413.2	575.7	92.2	1.121	0.980	0.998	0.039	0.979	0.017	-0.006	-0.35	0.99

TEST TC-484 PART 924 REX10-6 3.000 ALFA 5.01 RING 48 Y 4.00 Z -1.00 RUN SURVEY 4-901 DATE 2- 7-77

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	BWL
20	11.000	0.851	906.34	1494.3	471.9	80.7	0.826	0.974	1.000	0.053	0.968	0.032	0.105	6.19	1.90
21	11.666	0.852	907.15	1495.0	472.6	80.7	0.808	0.954	1.000	0.090	0.948	0.048	0.103	6.23	2.90
22	12.333	0.853	907.93	1495.3	473.3	80.6	0.787	0.932	1.000	0.136	0.925	0.070	0.088	5.46	4.31
25	13.000	0.853	908.04	1497.1	473.9	80.7	0.772	0.916	1.000	0.166	0.911	0.081	0.053	3.31	5.10
26	13.666	0.850	904.90	1493.7	470.7	80.7	0.778	0.925	1.001	0.151	0.922	0.069	0.023	1.45	4.31
27	14.333	0.852	907.07	1494.7	472.5	80.7	0.802	0.949	1.001	0.103	0.947	0.055	0.014	0.82	3.35
28	15.000	0.854	906.91	1493.6	473.3	80.8	0.829	0.974	1.001	0.054	0.973	0.044	0.014	0.82	2.57
29	15.666	0.851	906.24	1496.4	472.3	80.9	0.848	0.997	1.001	0.009	0.996	0.038	0.021	1.19	2.18
30	16.333	0.854	908.99	1493.7	473.4	80.5	0.867	1.013	1.001	-0.024	1.012	0.034	0.031	1.74	1.94
58	17.000	0.853	907.70	1490.4	471.6	80.3	0.875	1.072	1.002	-0.041	1.021	0.035	0.047	2.65	1.98
61	17.666	0.853	907.79	1489.5	471.4	80.6	0.855	1.003	1.002	-0.001	1.000	0.040	0.053	3.04	2.27
45	18.333	0.849	903.63	1490.8	469.2	80.5	0.832	0.983	1.001	0.035	0.981	0.042	0.042	2.44	2.46
70	19.000	0.851	905.54	1491.0	470.9	80.8	0.839	0.987	1.002	0.029	0.986	0.044	0.035	2.03	2.56

TEST	PART	REX10-6	ALFA	WING	Y	Z	RUN SURVEY	DATE	AEDC PROPULSION WIND TUNNEL
TC-484	926	3.000	-5.00	48	4.00	-1.00	4-901	2- 7-77	TRANSONIC 4T

POINT	X	M	VM	PT	Q	TT	ML	VML/VM	PTL/PT	CPL	UL/VM	VL/VM	WL/VM	AAL	SWL
8	11.000	0.850	905.21	1496.8	471.7	81.0	0.889	1.040	1.000	-0.081	1.038	0.005	-0.072	-3.98	0.27
9	11.666	0.852	906.79	1495.2	472.5	80.7	0.866	1.015	1.001	-0.029	1.004	-0.002	-0.148	-8.39	-0.11
10	12.333	0.853	907.79	1491.4	472.0	80.6	0.892	1.040	1.000	-0.080	1.025	-0.027	-0.175	-9.68	-1.51
11	13.000	0.855	910.14	1493.1	473.9	80.9	0.905	1.139	0.998	-0.294	1.123	-0.092	-0.167	-8.49	-4.70
13	13.666	0.849	904.57	1500.7	472.5	81.1	1.082	1.227	1.000	-0.463	1.217	-0.127	-0.092	-4.31	-5.97
14	14.333	0.850	904.99	1498.6	472.3	80.7	1.066	1.212	1.002	-0.427	1.207	-0.101	-0.027	-1.29	-4.89
15	15.000	0.853	908.25	1492.6	472.3	81.2	1.018	1.163	1.001	-0.328	1.161	-0.067	-0.007	-0.35	-3.31
18	15.666	0.852	907.44	1494.1	472.4	81.0	0.979	1.127	1.001	-0.255	1.126	-0.050	0.001	0.05	-2.54
19	16.333	0.852	907.50	1497.5	473.6	80.8	0.937	1.085	1.001	-0.170	1.084	-0.044	0.010	0.52	-2.33
20	17.000	0.852	907.60	1495.8	473.0	81.0	0.923	1.072	1.001	-0.142	1.071	-0.042	0.024	1.27	-2.24
21	17.666	0.854	908.95	1495.9	474.1	80.7	0.886	1.033	1.001	-0.065	1.032	-0.037	0.028	1.58	-2.07
22	18.333	0.857	907.49	1494.8	472.7	80.9	0.842	0.990	1.000	0.021	0.999	-0.033	0.012	0.71	-1.90
23	19.000	0.854	906.54	1496.3	474.4	81.1	0.841	0.966	1.000	0.029	0.986	-0.031	-0.000	-0.01	-1.82

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-486	73	0.799	1530.9	1005.2	2.997	855.9	449.3	78.8	48	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.01	-0.4635	-0.0014	0.0107	-0.0353	-0.0002	0.0030	0.0101
-1.99	-0.1723	-0.0010	0.0162	-0.0137	-0.0002	0.0029	0.0101
-0.01	-0.0012	-0.0008	0.0178	0.0016	-0.0002	0.0027	0.0099
2.02	0.1765	-0.0006	0.0148	0.0177	-0.0002	0.0030	0.0099
5.02	0.4740	-0.0007	0.0080	0.0398	-0.0003	0.0029	0.0100

PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-404	73	0.799	1530.9	1005.2	2.997	855.9	449.3	78.9	43	3	1

PRESSURE COEFFICIENTS CPS=(PS-P.)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.99	ALFA=-0.01	ALFA= 2.02	ALFA= 5.02
1	CPS 1=	0.1101	0.0616	0.0314	0.0053	-0.0349
2	CPS 2=	0.0707	-0.0144	-0.0394	-0.0629	-0.0944
3	CPS 3=	-0.0322	-0.0701	-0.0903	-0.1076	-0.1324
4	CPS 4=	-0.0780	-0.1195	-0.1240	-0.1379	-0.1525
5	CPS 5=	-0.0079	-0.1083	-0.1184	-0.1257	-0.1320
6	CPS 6=	-0.0315	-0.0526	-0.0613	-0.0451	-0.0701
7	CPS 7=	-0.0348	-0.0577	-0.0640	-0.0677	-0.0708
8	CPS 8=	0.0291	-0.0031	-0.0175	-0.0288	-0.0473
9	CPS 9=	0.0595	0.0150	-0.0066	-0.0299	-0.0696
10	CPS10=	0.0942	0.0414	0.0082	-0.0279	-0.1015
11	CPS11=	0.0972	0.0301	-0.0117	-0.0578	-0.1519
12	CPS12=	0.0877	0.0217	-0.0228	-0.0719	-0.1609
13	CPS13=	0.0693	0.0720	-0.0414	-0.0883	-0.1786
14	CPS14=	0.0624	-0.0216	-0.0416	-0.0852	-0.1612
15	CPS15=	0.0346	-0.0222	-0.0571	-0.0934	-0.1545
16	CPS16=	0.0244	-0.0249	-0.0540	-0.0826	-0.1295
17	CPS17=	0.0141	-0.0751	-0.0467	-0.0675	-0.1001
18	CPS18=	0.0157	-0.0130	-0.0283	-0.0429	-0.0641
19	CPS19=	0.0250	0.0048	-0.0044	-0.0137	-0.0263
20	CPS20=	0.0293	0.0111	0.0053	0.0000	-0.0070
21	CPS21=	0.0217	0.0108	0.0075	0.0033	-0.0008
22	CPS22=	0.0139	0.0057	0.0040	0.0004	-0.0017
23	CPS23=	0.0126	0.0048	0.0033	0.0013	-0.0004
24	CPS24=	0.0048	-0.0023	-0.0033	-0.0060	-0.0070
25	CPS25=	-0.0355	-0.0404	-0.0403	-0.0414	-0.0440

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

PAGE 1 SUMMARY FORCE, MOMENT

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-404	74	0.850	1501.6	936.5	3.001	906.5	473.4	82.3	48	3	1

ALPHA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.01	-0.4792	-0.0021	0.0109	-0.0334	-0.0002	0.0029	0.0101
-1.99	-0.1793	-0.0014	0.0159	-0.0135	-0.0002	0.0029	0.0102
-0.01	-0.0034	-0.0014	0.0170	0.0019	-0.0002	0.0027	0.0100
2.02	0.1804	-0.0013	0.0153	0.0176	-0.0002	0.0030	0.0097
5.02	0.4842	-0.0008	0.0083	0.0379	-0.0003	0.0029	0.0101

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PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-404	74	0.850	1501.6	936.5	3.001	906.5	473.4	82.3	49	3	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.99	ALFA=-0.01	ALFA= 2.02	ALFA= 5.02
1	CPS 1=	0.1183	0.0667	0.0405	0.0114	-0.0295
2	CPS 2=	0.0333	-0.0139	-0.0367	-0.0598	-0.0920
3	CPS 3=	-0.0377	-0.0742	-0.0910	-0.1101	-0.1343
4	CPS 4=	-0.0833	-0.1191	-0.1314	-0.1429	-0.1571
5	CPS 5=	-0.0846	-0.1170	-0.1249	-0.1299	-0.1343
6	CPS 6=	-0.0315	-0.0557	-0.0595	-0.0619	-0.0638
7	CPS 7=	-0.0330	-0.0578	-0.0618	-0.0694	-0.0604
8	CPS 8=	0.0346	-0.0704	-0.0116	-0.0202	-0.0328
9	CPS 9=	0.0495	0.0217	0.0017	-0.0183	-0.0539
10	CPS10=	0.1123	0.0498	0.0179	-0.0193	-0.0903
11	CPS11=	0.1088	0.0373	-0.0036	-0.0532	-0.1512
12	CPS12=	0.1037	0.0274	-0.0175	-0.0705	-0.1785
13	CPS13=	0.0797	0.0051	-0.0382	-0.0916	-0.1931
14	CPS14=	0.0697	-0.0013	-0.0431	-0.0909	-0.1798
15	CPS15=	0.0388	-0.0745	-0.0610	-0.1014	-0.1726
16	CPS16=	0.0283	-0.0302	-0.0593	-0.0918	-0.1436
17	CPS17=	0.0137	-0.0317	-0.0528	-0.0739	-0.1084
18	CPS18=	0.0147	-0.0171	-0.0314	-0.0423	-0.0648
19	CPS19=	0.0287	0.0067	-0.0023	-0.0065	-0.0204
20	CPS20=	0.0317	0.0162	0.0120	0.0103	0.0027
21	CPS21=	0.0285	0.0160	0.0135	0.0137	0.0086
22	CPS22=	0.0196	0.0101	0.0091	0.0097	0.0066
23	CPS23=	0.0173	0.0078	0.0069	0.0087	0.0051
24	CPS24=	0.0090	0.0002	-0.0013	0.0000	-0.0029
25	CPS25=	-0.0313	-0.0384	-0.0386	-0.0366	-0.0400

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	75	0.899	1473.9	872.1	2.999	954.3	493.8	94.7	49	1	1

ALPA	CNF	CY	CAF	CLMF	CLN	CLL	CAR
-5.00	-0.5014	-0.0021	0.0111	-0.0294	-0.0002	0.0034	0.0099
-2.00	-0.1900	-0.0013	0.0160	-0.0126	-0.0003	0.0030	0.0105
0.01	-0.0029	-0.0012	0.0182	0.0016	-0.0002	0.0028	0.0101
2.02	0.1983	-0.0010	0.0157	0.0160	-0.0002	0.0030	0.0103
5.01	0.5050	-0.0012	0.0087	0.0329	-0.0003	0.0029	0.0104

PAGE 2 SUMMARY PRESSURES

TEST	PART	M	PT	P	RE*10-6	VN	Q	TT	WING	RUN	SURVEY
TC-484	75	0.899	1473.9	972.1	2.998	954.3	493.8	84.7	48	3	1

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.00	ALFA=-2.00	ALFA= 0.01	ALFA= 2.02	ALFA= 5.01
1	CPS 1=	0.1295	0.0718	0.0490	0.0133	-0.0258
2	CPS 2=	0.0396	-0.0142	-0.0342	-0.0639	-0.0948
3	CPS 3=	-0.0373	-0.0826	-0.0987	-0.1227	-0.1452
4	CPS 4=	-0.0923	-0.1361	-0.1455	-0.1641	-0.1736
5	CPS 5=	-0.0989	-0.1357	-0.1391	-0.1486	-0.1444
6	CPS 6=	-0.0315	-0.0606	-0.0603	-0.0657	-0.0594
7	CPS 7=	-0.0285	-0.0586	-0.0582	-0.0591	-0.0467
8	CPS 8=	0.0436	0.0032	-0.0037	-0.0146	-0.0147
9	CPS 9=	0.0877	0.0292	0.0127	-0.0102	-0.0316
10	CPS10=	0.1295	0.0401	0.0302	-0.0114	-0.0719
11	CPS11=	0.1277	0.0472	0.0064	-0.0500	-0.1442
12	CPS12=	0.1184	0.0353	-0.0108	-0.0740	-0.1836
13	CPS13=	0.0916	0.0090	-0.0364	-0.1029	-0.2203
14	CPS14=	0.0795	-0.0007	-0.0455	-0.1102	-0.2305
15	CPS15=	0.0445	-0.0297	-0.0702	-0.1290	-0.2519
16	CPS16=	0.0247	-0.0416	-0.0750	-0.1225	-0.1989
17	CPS17=	0.0096	-0.0455	-0.0696	-0.0975	-0.1238
18	CPS18=	0.0120	-0.0265	-0.0374	-0.0514	-0.0626
19	CPS19=	0.0319	0.0068	0.0032	-0.0043	-0.0101
20	CPS20=	0.0400	0.0270	0.0218	0.0169	0.0147
21	CPS21=	0.0380	0.0226	0.0240	0.0204	0.0195
22	CPS22=	0.0205	0.0153	0.0179	0.0153	0.0149
23	CPS23=	0.0254	0.0115	0.0137	0.0107	0.0113
24	CPS24=	0.0175	0.0026	0.0044	0.0010	0.0013
25	CPS25=	-0.0242	-0.0368	-0.0334	-0.0362	-0.0378

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	930	0.977	1443.3	832.0	3.000	970.6	495.9	79.6	48	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-4.98	-0.5242	-0.0006	0.0122	-0.0710	-0.0006	0.0043	0.0104
-1.99	-0.1966	-0.0012	0.0179	-0.0124	-0.0002	0.0033	0.0096
0.00	-0.0056	-0.0007	0.0187	0.0002	-0.0001	0.0029	0.0104
2.01	0.1921	-0.0008	0.0165	0.0127	-0.0001	0.0031	0.0099
5.01	0.5400	-0.0007	0.0090	0.0164	-0.0000	0.0061	0.0100

PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	930	0.922	1443.3	832.8	3.000	970.6	495.9	79.6	4B	3	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-4.99	ALFA=-1.99	ALFA= 0.00	ALFA= 2.01	ALFA= 5.01
1	CPS 1=	0.1372	0.0890	0.0495	0.0266	-0.0150
2	CPS 2=	0.0338	-0.0030	-0.0399	-0.0585	-0.0924
3	CPS 3=	-0.0417	-0.0738	-0.1075	-0.1218	-0.1473
4	CPS 4=	-0.1094	-0.1354	-0.1661	-0.1724	-0.1871
5	CPS 5=	-0.1193	-0.1356	-0.1585	-0.1548	-0.1495
6	CPS 6=	-0.0409	-0.0511	-0.0656	-0.0593	-0.0502
7	CPS 7=	-0.0302	-0.0429	-0.0576	-0.0487	-0.0314
8	CPS 8=	0.0440	0.0186	-0.0026	-0.0016	0.0032
9	CPS 9=	0.0840	0.0457	0.0150	0.0036	-0.0102
10	CPS10=	0.1334	0.0742	0.0335	0.0032	-0.0482
11	CPS11=	0.1307	0.0643	0.0086	-0.0374	-0.1250
12	CPS12=	0.1057	0.0375	-0.0229	-0.0755	-0.1789
13	CPS13=	0.0940	0.0246	-0.0387	-0.0949	-0.2067
14	CPS14=	0.0797	0.0120	-0.0520	-0.1093	-0.2219
15	CPS15=	0.0416	-0.0201	-0.0821	-0.1384	-0.2531
16	CPS16=	0.0142	-0.0379	-0.0967	-0.1498	-0.2729
17	CPS17=	-0.0046	-0.0461	-0.0943	-0.1262	-0.2857
18	CPS18=	-0.0034	-0.0239	-0.0502	-0.0529	-0.0990
19	CPS19=	0.0733	0.0184	0.0020	0.0060	0.0114
20	CPS20=	0.0370	0.0360	0.0248	0.0298	0.0408
21	CPS21=	0.0396	0.0399	0.0289	0.0337	0.0442
22	CPS22=	0.0383	0.0320	0.0206	0.0262	0.0348
23	CPS23=	0.0743	0.0250	0.0134	0.0184	0.0246
24	CPS24=	0.0134	0.0152	0.0032	0.0076	0.0102
25	CPS25=	-0.0292	-0.0247	-0.0369	-0.0318	-0.0310

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	PT*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	76	0.950	1429.3	799.9	2.968	999.3	505.0	84.2	40	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.01	-0.5483	-0.0027	0.0154	0.0014	-0.0002	0.0045	0.0102
-1.98	-0.2088	-0.0013	0.0194	-0.0023	-0.0003	0.0030	0.0101
0.01	-0.0064	-0.0010	0.0194	0.0032	-0.0002	0.0026	0.0107
2.02	0.1996	-0.0011	0.0190	0.0087	-0.0002	0.0029	0.0103
5.02	0.5438	-0.0005	0.0142	0.0055	-0.0004	0.0029	0.0100

PAGE 7 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-404	76	0.950	1429.3	799.9	2.968	999.3	505.0	84.2	48.	3	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.98	ALFA= 0.01	ALFA= 2.02	ALFA= 5.02
1	CPS 1=	0.1432	0.0923	0.0586	0.0296	-0.0083
2	CPS 2=	0.0462	-0.0005	-0.0316	-0.0550	-0.0853
3	CPS 3=	-0.0383	-0.0888	-0.1094	-0.1284	-0.1518
4	CPS 4=	-0.1136	-0.1480	-0.1696	-0.1828	-0.1901
5	CPS 5=	-0.1441	-0.1860	-0.2095	-0.2177	-0.2018
6	CPS 6=	-0.0265	-0.0421	-0.0453	-0.0425	-0.0337
7	CPS 7=	-0.0191	-0.0404	-0.0410	-0.0372	-0.0153
8	CPS 8=	0.0573	0.0224	0.0107	0.0087	0.0214
9	CPS 9=	0.1016	0.0533	0.0306	0.0183	0.0128
10	CPS10=	0.1517	0.0870	0.0494	0.0171	-0.0245
11	CPS11=	0.1478	0.0733	0.0244	-0.0237	-0.1023
12	CPS12=	0.1383	0.0586	0.0042	-0.0511	-0.1461
13	CPS13=	0.1080	0.0293	-0.0275	-0.0859	-0.1856
14	CPS14=	0.0916	0.0138	-0.0427	-0.1019	-0.2020
15	CPS15=	0.0500	-0.0225	-0.0777	-0.1370	-0.2337
16	CPS16=	0.0201	-0.0490	-0.1020	-0.1601	-0.2550
17	CPS17=	-0.0173	-0.0888	-0.1375	-0.1876	-0.2822
18	CPS18=	-0.0261	-0.0598	-0.1350	-0.1738	-0.3037
19	CPS19=	0.0145	0.0111	-0.0028	0.0007	-0.0748
20	CPS20=	0.0415	0.0409	0.0373	0.0441	0.0509
21	CPS21=	0.0502	0.0456	0.0418	0.0476	0.0668
22	CPS22=	0.0464	0.0374	0.0337	0.0386	0.0590
23	CPS23=	0.0403	0.0297	0.0242	0.0277	0.0429
24	CPS24=	0.0298	0.0177	0.0113	0.0138	0.0247
25	CPS25=	-0.0144	-0.0233	-0.0294	-0.0272	-0.0198

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-404	930	0.976	1425.6	774.0	3.005	1020.5	516.5	81.6	48	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.01	-0.5559	-0.0017	0.0234	0.0139	-0.0001	0.0032	0.0098
-1.99	-0.2210	-0.0015	0.0312	0.0049	-0.0002	0.0028	0.0106
0.01	-0.0072	-0.0010	0.0278	0.0007	-0.0001	0.0024	0.0106
2.02	0.2022	-0.0007	0.0266	0.0013	-0.0001	0.0028	0.0108
5.02	0.5524	-0.0007	0.0208	-0.0116	-0.0002	0.0035	0.0100

PAGE 2 SUMMARY PRESSURES

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	939	0.976	1425.6	774.0	3.005	1020.5	518.5	81.6	48	3	1

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.99	ALFA= 0.01	ALFA= 2.02	ALFA= 5.02
1	CPS 1=	0.1563	0.1086	0.0783	0.0468	0.0074
2	CPS 2=	0.0527	0.0094	-0.0169	-0.0435	-0.0770
3	CPS 3=	-0.0311	-0.0684	-0.0857	-0.1174	-0.1396
4	CPS 4=	-0.1176	-0.1460	-0.1659	-0.1846	-0.2021
5	CPS 5=	-0.1655	-0.1923	-0.2042	-0.2196	-0.2284
6	CPS 6=	-0.2305	-0.2474	-0.2482	-0.2478	-0.1097
7	CPS 7=	0.0190	-0.0160	-0.0226	-0.0237	0.0308
8	CPS 8=	0.0911	0.0616	0.0541	0.0515	0.0569
9	CPS 9=	0.1265	0.0819	0.0648	0.0540	0.0429
10	CPS10=	0.1737	0.1121	0.0787	0.0486	0.0062
11	CPS11=	0.1685	0.0950	0.0498	0.0045	-0.0727
12	CPS12=	0.1424	0.0655	0.0166	-0.0367	-0.1290
13	CPS13=	0.1249	0.0487	-0.0037	-0.0630	-0.1602
14	CPS14=	0.1091	0.0324	-0.0210	-0.0783	-0.1793
15	CPS15=	0.0674	-0.0045	-0.0551	-0.1133	-0.2109
16	CPS16=	0.0347	-0.0333	-0.0785	-0.1401	-0.2323
17	CPS17=	-0.0090	-0.0710	-0.1156	-0.1700	-0.2590
18	CPS18=	-0.0609	-0.1166	-0.1585	-0.2050	-0.2896
19	CPS19=	-0.1002	-0.1304	-0.1604	-0.2210	-0.3151
20	CPS20=	-0.0195	0.0236	0.0241	-0.0048	-0.0869
21	CPS21=	0.0380	0.0579	0.0621	0.0620	0.0575
22	CPS22=	0.0477	0.0552	0.0586	0.0645	0.0777
23	CPS23=	0.0512	0.0462	0.0465	0.0515	0.0686
24	CPS24=	0.0473	0.0310	0.0292	0.0313	0.0466
25	CPS25=	-0.0001	-0.0121	-0.0144	-0.0142	-0.0031

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-404	77	1.002	1476.0	751.6	3.003	1046.1	320.2	85.0	48	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.01	-0.5538	-0.0021	0.0263	0.0238	-0.0003	0.0028	0.0005
-1.99	-0.2157	-0.0016	0.0314	0.0076	-0.0003	0.0025	0.0105
0.01	-0.0088	-0.0014	0.0325	0.0033	-0.0003	0.0023	0.0108
2.02	0.2918	-0.0011	0.0306	-0.0013	-0.0003	0.0024	0.0106
5.00	0.5440	-0.0010	0.0247	-0.0159	-0.0005	0.0034	0.0086

PAGE 2 SUMMARY PRESSURES

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	77	1.002	1426.0	751.6	3.003	1046.1	520.2	85.0	48	3	1

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.99	ALFA= 0.01	ALFA= 2.02	ALFA= 5.00
1	CPS 1=	0.1742	0.1253	0.0952	0.0653	0.0259
2	CPS 2=	0.0744	0.0297	0.0044	-0.0206	-0.0509
3	CPS 3=	-0.0111	-0.0489	-0.0715	-0.0921	-0.1199
4	CPS 4=	-0.0929	-0.1275	-0.1444	-0.1619	-0.1838
5	CPS 5=	-0.1402	-0.1772	-0.1897	-0.2027	-0.2151
6	CPS 6=	-0.2186	-0.2365	-0.2405	-0.2442	-0.2399
7	CPS 7=	-0.2166	-0.2228	-0.2193	-0.2146	-0.1911
8	CPS 8=	-0.0975	-0.1402	-0.1301	-0.1096	0.0013
9	CPS 9=	0.1597	0.0760	0.0541	0.0486	0.0676
10	CPS10=	0.2115	0.1441	0.1095	0.0811	0.0518
11	CPS11=	0.2092	0.1362	0.0927	0.0496	-0.0171
12	CPS12=	0.1964	0.1212	0.0735	0.0223	-0.0613
13	CPS13=	0.1645	0.0996	0.0401	-0.0143	-0.1041
14	CPS14=	0.1450	0.0708	0.0205	-0.0339	-0.1254
15	CPS15=	0.1025	0.0319	-0.0176	-0.0715	-0.1612
16	CPS16=	0.0695	0.0031	-0.0420	-0.0990	-0.1859
17	CPS17=	0.0245	-0.0368	-0.0800	-0.1319	-0.2163
18	CPS18=	-0.0242	-0.0792	-0.1212	-0.1682	-0.2464
19	CPS19=	-0.0706	-0.1239	-0.1617	-0.2073	-0.2797
20	CPS20=	-0.1016	-0.1411	-0.1613	-0.2087	-0.3090
21	CPS21=	-0.0640	-0.0330	-0.0132	-0.0721	-0.1241
22	CPS22=	-0.0407	0.0240	0.0382	0.0432	0.0150
23	CPS23=	0.0125	0.0493	0.0566	0.0617	0.0566
24	CPS24=	0.0430	0.0518	0.0535	0.0563	0.0599
25	CPS25=	0.0209	0.0165	0.0159	0.0175	0.0257

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	79	1.025	1394.0	715.2	2.995	1060.7	526.3	79.4	48	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.02	-0.5614	-0.0023	0.0278	0.0201	-0.0003	0.0028	0.0078
-1.99	-0.2222	-0.0012	0.0308	0.0054	-0.0004	0.0024	0.0104
0.01	-0.0105	-0.0011	0.0320	0.0037	-0.0004	0.0023	0.0120
2.02	0.2067	-0.0013	0.0306	0.0030	-0.0004	0.0026	0.0103
5.02	0.5499	-0.0009	0.0244	-0.0121	-0.0005	0.0032	0.0096

PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	78	1.025	1394.8	715.2	2.995	1040.7	526.3	79.4	48	3	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.02	ALFA=-1.98	ALFA= 0.01	ALFA= 2.02	ALFA= 3.02
1	CPS 1=	0.2073	0.1484	0.1224	0.0863	0.0484
2	CPS 2=	0.1012	0.0578	0.0305	0.0004	-0.0336
3	CPS 3=	0.0155	-0.0768	-0.0460	-0.0732	-0.0998
4	CPS 4=	-0.0677	-0.1069	-0.1231	-0.1455	-0.1648
5	CPS 5=	-0.1254	-0.1583	-0.1710	-0.1874	-0.2001
6	CPS 6=	-0.1978	-0.2709	-0.2276	-0.2332	-0.2314
7	CPS 7=	-0.1974	-0.2116	-0.2137	-0.2045	-0.1988
8	CPS 8=	-0.1371	-0.1444	-0.1469	-0.1356	-0.1198
9	CPS 9=	-0.0527	-0.1128	-0.1189	-0.1031	-0.0478
10	CPS10=	0.1934	0.0535	-0.0099	-0.0274	0.0018
11	CPS11=	0.2156	0.0981	0.0290	-0.0196	-0.0475
12	CPS12=	0.2135	0.1065	0.0317	-0.0196	-0.0794
13	CPS13=	0.1897	0.0917	0.0222	-0.0251	-0.1076
14	CPS14=	0.1738	0.0858	0.0226	-0.0221	-0.1139
15	CPS15=	0.1392	0.0569	-0.0016	-0.0443	-0.1359
16	CPS16=	0.1048	0.0349	-0.0132	-0.0611	-0.1478
17	CPS17=	0.0673	-0.0000	-0.0373	-0.0854	-0.1673
18	CPS18=	0.0146	-0.0391	-0.0734	-0.1179	-0.1931
19	CPS19=	-0.0304	-0.0801	-0.1073	-0.1524	-0.2233
20	CPS20=	-0.0595	-0.0885	-0.0899	-0.1377	-0.2421
21	CPS21=	-0.0257	-0.0008	0.0220	-0.0063	-0.0898
22	CPS22=	-0.0151	0.0239	0.0434	0.0285	-0.0023
23	CPS23=	-0.0095	0.0213	0.0364	0.0234	0.0105
24	CPS24=	-0.0086	0.0116	0.0205	0.0089	0.0002
25	CPS25=	-0.0297	-0.0192	-0.0162	-0.0219	-0.0271

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	79	1.049	1388.0	681.7	2.998	1080.8	533.0	79.1	40	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.01	-0.5762	-0.0024	0.0304	0.0373	-0.0004	0.0025	0.0154
-1.98	-0.2211	-0.0015	0.0343	0.0155	-0.0004	0.0026	0.0140
0.01	-0.0042	-0.0017	0.0345	0.0037	-0.0004	0.0025	0.0142
2.01	0.2171	-0.0012	0.0327	-0.0083	-0.0004	0.0027	0.0148
5.00	0.5694	-0.0009	0.0276	-0.0276	-0.0005	0.0029	0.0168

PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	PT*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	79	1.049	1388.0	691.7	2.998	1080.8	533.0	79.1	40	3	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.00	ALFA= 0.01	ALFA= 2.01	ALFA= 5.00
1	CPS 1=	0.2271	0.1793	0.1479	0.1185	0.0802
2	CPS 2=	0.1295	0.0969	0.0601	0.0353	0.0008
3	CPS 3=	0.0481	0.0120	-0.0112	-0.0325	-0.0564
4	CPS 4=	-0.0307	-0.0632	-0.0817	-0.0998	-0.1155
5	CPS 5=	-0.0838	-0.1068	-0.1212	-0.1314	-0.1473
6	CPS 6=	-0.1530	-0.1654	-0.1714	-0.1732	-0.1801
7	CPS 7=	-0.1571	-0.1515	-0.1571	-0.1532	-0.1536
8	CPS 8=	-0.0944	-0.0894	-0.0993	-0.0956	-0.0996
9	CPS 9=	-0.0907	-0.0789	-0.0868	-0.0814	-0.0708
10	CPS10=	0.0766	-0.0093	-0.0361	-0.0413	-0.0428
11	CPS11=	0.1598	0.0549	-0.0050	-0.0408	-0.0789
12	CPS12=	0.1665	0.0552	-0.0065	-0.0578	-0.1137
13	CPS13=	0.1576	0.0350	-0.0254	-0.0797	-0.1490
14	CPS14=	0.1403	0.0324	-0.0303	-0.0868	-0.1591
15	CPS15=	0.1220	0.0049	-0.0565	-0.1095	-0.1856
16	CPS16=	0.1042	-0.0160	-0.0770	-0.1299	-0.2032
17	CPS17=	0.0712	-0.0413	-0.0997	-0.1524	-0.2208
18	CPS18=	0.0318	-0.0707	-0.1257	-0.1771	-0.2287
19	CPS19=	-0.0086	-0.0501	-0.1135	-0.1966	-0.2019
20	CPS20=	-0.0425	-0.0817	-0.1087	-0.1593	-0.2312
21	CPS21=	-0.0326	-0.0428	-0.0376	-0.0473	-0.1131
22	CPS22=	-0.0170	-0.0108	0.0160	0.0278	0.0192
23	CPS23=	0.0179	0.0605	0.0899	0.0902	0.0808
24	CPS24=	0.0301	0.0919	0.1006	0.0988	0.0783
25	CPS25=	0.0188	0.0998	0.0605	0.0692	0.0297

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VN	Q	TT	WING	RUN	SURVEY
TC-404	80	1.181	1379.3	644.0	3.003	1123.0	547.1	79.1	48	3	1

ALPA	CNF	CY	CAF	CLNF	CLF	CLL	CAB
-4.90	-0.5448	-0.0027	0.0374	0.0396	-0.0002	0.0024	0.0161
-1.94	-0.2157	-0.0018	0.0352	0.0169	-0.0003	0.0027	0.0156
0.01	-0.0058	-0.0019	0.0360	0.0044	-0.0003	0.0026	0.0150
2.02	0.2054	-0.0013	0.0346	-0.0066	-0.0004	0.0026	0.0152
5.01	0.5380	-0.0010	0.0297	-0.0289	-0.0005	0.0028	0.0170

PAGE 3 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	PR*10-6	VP	Q	TT	WING	RUN	SURVEY
TC-486	80	1.101	1370.3	644.0	3.003	1173.0	547.1	79.1	48	3	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-4.98	ALFA=-1.98	ALFA= 0.01	ALFA= 2.02	ALFA= 5.01
1	CPS 1=	0.2406	0.1940	0.1661	0.1305	0.1008
2	CPS 2=	0.1477	0.1050	0.0790	0.0519	0.0187
3	CPS 3=	0.0742	0.0372	0.0143	-0.0071	-0.0363
4	CPS 4=	-0.0060	-0.0401	-0.0589	-0.0766	-0.0953
5	CPS 5=	-0.0622	-0.0859	-0.0998	-0.1146	-0.1259
6	CPS 6=	-0.1776	-0.1470	-0.1570	-0.1619	-0.1656
7	CPS 7=	-0.1370	-0.1409	-0.1433	-0.1476	-0.1305
8	CPS 8=	-0.0814	-0.0894	-0.0913	-0.0891	-0.0803
9	CPS 9=	-0.0735	-0.0772	-0.0747	-0.0662	-0.0517
10	CPS10=	-0.0185	-0.0280	-0.0233	-0.0150	-0.0126
11	CPS11=	0.1932	0.0849	0.0355	-0.0024	-0.0348
12	CPS12=	0.2108	0.1070	0.0470	-0.0005	-0.0597
13	CPS13=	0.1959	0.0944	0.0322	-0.0240	-0.0911
14	CPS14=	0.1901	0.0873	0.0278	-0.0312	-0.0993
15	CPS15=	0.1610	0.0703	0.0073	-0.0400	-0.1187
16	CPS16=	0.1406	0.0404	-0.0006	-0.0378	-0.1169
17	CPS17=	0.1053	0.0429	-0.0073	-0.0521	-0.1449
18	CPS18=	0.0647	0.0036	-0.0353	-0.0972	-0.1629
19	CPS19=	0.0217	-0.0432	-0.0833	-0.1394	-0.1946
20	CPS20=	-0.0260	-0.0896	-0.1321	-0.1881	-0.2411
21	CPS21=	-0.0422	-0.0903	-0.1228	-0.1600	-0.2177
22	CPS22=	-0.0415	-0.0943	-0.0670	-0.0775	-0.0966
23	CPS23=	-0.0503	-0.0435	-0.0333	-0.0066	-0.0031
24	CPS24=	-0.0280	-0.0059	0.0086	0.0170	0.0233
25	CPS25=	-0.0386	-0.0152	-0.0086	-0.0073	-0.0077

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-404	01	1.152	1371.7	401.9	3.001	1165.5	559.0	79.5	48	3	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.00	-0.5289	-0.0026	0.0320	0.0504	-0.0001	0.0019	0.0170
-2.00	-0.2122	-0.0019	0.0349	0.0722	-0.0002	0.0021	0.0163
0.01	-0.0102	-0.0018	0.0357	0.0061	-0.0003	0.0022	0.0180
2.02	0.1962	-0.0016	0.0338	-0.0111	-0.0003	0.0024	0.0171
5.02	0.5210	-0.0009	0.0295	-0.0388	-0.0004	0.0024	0.0181

PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	81	1.197	1371.7	601.9	3.001	1165.5	559.0	79.5	48	3	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.00	ALFA=-2.00	ALFA= 0.01	ALFA= 2.02	ALFA= 5.02
1	CPS 1=	0.2470	0.1990	0.1693	0.1449	0.1061
2	CPS 2=	0.1509	0.1106	0.0813	0.0515	0.0194
3	CPS 3=	0.0806	0.0430	0.0201	-0.0015	-0.0208
4	CPS 4=	0.0042	-0.0263	-0.0466	-0.0684	-0.0772
5	CPS 5=	-0.0419	-0.0725	-0.0875	-0.0896	-0.1103
6	CPS 6=	-0.1179	-0.1215	-0.1334	-0.1442	-0.1407
7	CPS 7=	-0.1080	-0.1172	-0.1268	-0.1317	-0.1211
8	CPS 8=	-0.0602	-0.0732	-0.0805	-0.0758	-0.0761
9	CPS 9=	-0.0620	-0.0626	-0.0606	-0.0636	-0.0514
10	CPS10=	-0.0317	-0.0296	-0.0333	-0.0275	-0.0220
11	CPS11=	0.1044	0.0414	0.0104	-0.0150	-0.0390
12	CPS12=	0.1839	0.0833	0.0330	-0.0094	-0.0683
13	CPS13=	0.1013	0.0808	0.0215	-0.0265	-0.0935
14	CPS14=	0.1797	0.0885	0.0273	-0.0304	-0.0988
15	CPS15=	0.1613	0.0620	0.0016	-0.0491	-0.1157
16	CPS16=	0.1459	0.0622	0.0038	-0.0530	-0.1250
17	CPS17=	0.1146	0.0379	-0.0203	-0.0725	-0.1301
18	CPS18=	0.0871	0.0150	-0.0385	-0.0805	-0.1540
19	CPS19=	0.0493	-0.0156	-0.0635	-0.1125	-0.1763
20	CPS20=	0.0184	-0.0580	-0.1038	-0.1495	-0.2092
21	CPS21=	-0.0106	-0.0494	-0.0895	-0.1240	-0.1797
22	CPS22=	-0.0045	-0.0237	-0.0371	-0.0534	-0.0730
23	CPS23=	-0.0149	-0.0090	-0.0102	-0.0070	0.0131
24	CPS24=	-0.0138	-0.0015	0.0226	0.0519	0.0423
25	CPS25=	0.0271	0.0398	0.0346	0.0205	0.0142

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VW	Q	TT	WING	RUN	SURVEY
TC-484	945	0.925	1442.5	839.2	2.993	873.6	496.8	80.7	65	6	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAB
-5.01	-0.5555	-0.0007	0.0148	-0.0036	-0.0003	-0.0001	0.0095
-1.98	-0.2110	-0.0009	0.0192	-0.0073	-0.0002	0.0003	0.0098
0.02	-0.0047	-0.0006	0.0200	-0.0011	-0.0002	0.0002	0.0101
2.01	0.2048	-0.0007	0.0181	0.0056	-0.0001	0.0007	0.0096
5.01	0.5511	-0.0009	0.0118	0.0021	0.0001	0.0010	0.0100

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PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	945	0.925	1442.5	930.2	2.993	973.6	496.8	80.7	68	6	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.98	ALFA= 0.02	ALFA= 2.01	ALFA= 5.01
1	CPS 1=	0.1394	0.0874	0.0547	0.0279	-0.0139
2	CPS 2=	0.0424	-0.0050	-0.0343	-0.0559	-0.0904
3	CPS 3=	-0.0321	-0.0756	-0.1015	-0.1173	-0.1441
4	CPS 4=	-0.0969	-0.1358	-0.1565	-0.1665	-0.1794
5	CPS 5=	-0.1035	-0.1338	-0.1465	-0.1449	-0.1393
6	CPS 6=	-0.0738	-0.0443	-0.0531	-0.0477	-0.0390
7	CPS 7=	-0.0095	-0.0313	-0.0391	-0.0305	-0.0163
8	CPS 8=	0.0678	0.0361	0.0215	0.0215	0.0234
9	CPS 9=	0.1094	0.0646	0.0415	0.0298	0.0129
10	CPS10=	0.1597	0.0878	0.0501	0.0197	-0.0299
11	CPS11=	0.1336	0.0580	0.0085	-0.0363	-0.1145
12	CPS12=	0.0951	0.0132	-0.0403	-0.0925	-0.1862
13	CPS13=	0.0722	-0.0142	-0.0703	-0.1271	-0.2273
14	CPS14=	0.0481	-0.0391	-0.0971	-0.1553	-0.2571
15	CPS15=	0.0068	-0.0802	-0.1433	-0.2007	-0.3002
16	CPS16=	-0.0093	-0.0872	-0.1569	-0.2313	-0.3299
17	CPS17=	-0.0067	-0.0592	-0.0915	-0.1759	-0.3609
18	CPS18=	0.0140	-0.0132	-0.0249	-0.0275	-0.1355
19	CPS19=	0.0448	0.0309	0.0237	0.0300	0.0230
20	CPS20=	0.0567	0.0467	0.0407	0.0484	0.0562
21	CPS21=	0.0591	0.0483	0.0431	0.0500	0.0600
22	CPS22=	0.0483	0.0377	0.0325	0.0400	0.0478
23	CPS23=	0.0400	0.0783	0.0231	0.0291	0.0338
24	CPS24=	0.0283	0.0168	0.0111	0.0157	0.0179
25	CPS25=	-0.0174	-0.0757	-0.0313	-0.0253	-0.0275

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	946	0.973	1432.2	780.2	2.907	1020.4	517.5	84.3	68	6	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAR
-5.01	-0.5530	-0.0014	0.0294	0.0191	-0.0002	0.0006	0.0093
-1.99	-0.7090	-0.0009	0.0346	0.0013	-0.0001	0.0007	0.0099
0.02	-0.0057	-0.0012	0.0346	-0.0008	-0.0001	0.0006	0.0101
2.01	0.2021	-0.0010	0.0328	-0.0025	-0.0001	0.0006	0.0100
5.01	0.5521	-0.0012	0.0279	-0.0261	0.0000	0.0006	0.0094

PAGE 2 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	946	0.973	1432.2	780.2	2.997	1020.4	517.5	84.3	68	6	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.99	ALFA= 0.02	ALFA= 2.01	ALFA= 5.01
1	CPS 1=	0.1573	0.1086	0.0795	0.0464	0.0095
2	CPS 2=	0.0533	0.0100	-0.0160	-0.0436	-0.0742
3	CPS 3=	-0.0309	-0.0671	-0.0855	-0.1169	-0.1354
4	CPS 4=	-0.1125	-0.1446	-0.1645	-0.1832	-0.2017
5	CPS 5=	-0.1647	-0.1919	-0.2037	-0.2147	-0.2274
6	CPS 6=	-0.2076	-0.2411	-0.2234	-0.1767	-0.0663
7	CPS 7=	0.0402	0.0249	0.0230	0.0293	0.0517
8	CPS 8=	0.1059	0.0854	0.0728	0.0704	0.0820
9	CPS 9=	0.1461	0.1103	0.0900	0.0768	0.0720
10	CPS10=	0.1870	0.1317	0.0977	0.0652	0.0315
11	CPS11=	0.1603	0.0989	0.0539	0.0074	-0.0544
12	CPS12=	0.1242	0.0571	0.0037	-0.0498	-0.1238
13	CPS13=	0.0983	0.0212	-0.0297	-0.0899	-0.1693
14	CPS14=	0.0690	-0.0079	-0.0587	-0.1174	-0.2021
15	CPS15=	0.0170	-0.0584	-0.1075	-0.1644	-0.2462
16	CPS16=	-0.0197	-0.0962	-0.1442	-0.1994	-0.2769
17	CPS17=	-0.0547	-0.1344	-0.1824	-0.2361	-0.3139
18	CPS18=	-0.0835	-0.1624	-0.2137	-0.2676	-0.3427
19	CPS19=	-0.0873	-0.1617	-0.1639	-0.2378	-0.3596
20	CPS20=	0.0023	0.0737	0.0351	0.0162	-0.1309
21	CPS21=	0.0433	0.0634	0.0734	0.0756	0.0501
22	CPS22=	0.0531	0.0661	0.0718	0.0791	0.0829
23	CPS23=	0.0574	0.0604	0.0600	0.0460	0.0816
24	CPS24=	0.0500	0.0451	0.0419	0.0454	0.0622
25	CPS25=	0.0062	-0.0012	-0.0056	-0.0034	0.0108

PAGE 1 SUMMARY FORCE, MOMENT

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VW	Q	TT	WING	RUN	SURVEY
TC-484	947	1.024	1421.7	730.1	2.990	1066.9	536.0	86.8	6B	6	1

ALFA	CNF	CY	CAF	CLMF	CLN	CLL	CAR
-5.01	-0.5569	-0.0012	0.0345	0.0298	-0.0002	0.0005	0.0075
-1.99	-0.2106	-0.0010	0.0381	0.0072	-0.0002	0.0004	0.0085
0.00	-0.0077	-0.0007	0.0397	-0.0002	-0.0001	0.0007	0.0080
2.02	0.2082	-0.0007	0.0379	-0.0070	-0.0002	0.0006	0.0075
5.01	0.5459	-0.0009	0.0314	-0.0290	-0.0001	0.0006	0.0085

PAGE 7 SUMMARY PRESSURES

AEDC PROPULSION WIND TUNNEL
TRANSONIC 4T

TEST	PART	M	PT	P	RE*10-6	VM	Q	TT	WING	RUN	SURVEY
TC-484	947	1.024	1421.7	730.1	2.990	1066.9	536.0	86.0	60	5	1

PRESSURE COEFFICIENTS CPS=(PS-P)/Q

ORIFICE		ALFA=-5.01	ALFA=-1.99	ALFA= 0.00	ALFA= 2.02	ALFA= 5.01
1	CPS 1=	0.2071	0.1560	0.1277	0.0959	0.0537
2	CPS 2=	0.0974	0.0554	0.0259	0.0037	-0.0355
3	CPS 3=	0.0144	-0.0232	-0.0485	-0.0688	-0.0996
4	CPS 4=	-0.0714	-0.1046	-0.1253	-0.1407	-0.1653
5	CPS 5=	-0.1790	-0.1595	-0.1717	-0.1824	-0.2001
6	CPS 6=	-0.2030	-0.2217	-0.2204	-0.2282	-0.2316
7	CPS 7=	-0.2030	-0.2115	-0.2105	-0.1991	-0.1957
8	CPS 8=	-0.1412	-0.1461	-0.1402	-0.1261	-0.0911
9	CPS 9=	0.0937	-0.0553	-0.0339	-0.0127	0.0419
10	CPS10=	0.2711	0.1340	0.0944	0.0713	0.0521
11	CPS11=	0.2181	0.1292	0.0793	0.0412	-0.0143
12	CPS12=	0.1866	0.0993	0.0485	0.0063	-0.0695
13	CPS13=	0.1607	0.0757	0.0243	-0.0249	-0.1033
14	CPS14=	0.1330	0.0534	0.0038	-0.0460	-0.1270
15	CPS15=	0.0845	0.0101	-0.0356	-0.0825	-0.1647
16	CPS16=	0.0493	-0.0226	-0.0680	-0.1129	-0.1910
17	CPS17=	0.0142	-0.0572	-0.1031	-0.1464	-0.2250
18	CPS18=	-0.0147	-0.0834	-0.1324	-0.1750	-0.2530
19	CPS19=	-0.0323	-0.0962	-0.1462	-0.1931	-0.2703
20	CPS20=	-0.0455	-0.0812	-0.1212	-0.1713	-0.2744
21	CPS21=	-0.0276	-0.0005	-0.0056	-0.0252	-0.1027
22	CPS22=	-0.0190	0.0151	0.0196	0.0200	-0.0099
23	CPS23=	-0.0177	0.0140	0.0162	0.0190	0.0073
24	CPS24=	-0.0183	0.0062	0.0077	0.0121	0.0021
25	CPS25=	-0.0382	-0.0224	-0.0192	-0.0119	-0.0228